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Flutter and Vibration Analysis by a Modal Method: Analytical Development and Computational Procedure

31 JULY 1963

Prepared by WILLIAM P. RODDEN and EDITH F. FARKAS

Aeromechanics Department

Aerodynamics and Propulsion Research Laboratory

Laboratories Division

and

HEATHER A. MALCOM

Computation and Data Processing Center

Prepared for COMMANDER SPACE SYSTEMS DIVISION

UNITED STATES AIR FORCE

Inglewood, California



LABORATORIES DIVISION • AEROSPACE CORPORATION
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William P. Rodden and Edith F. Farkas
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Prepared by: William P. Rodden
William P. Rodden, Head
Dynamics Section
Aerodynamics and Propulsion
Research Laboratory

Edith F. Farkas
Edith F. Farkas, MTS
Dynamics Section
Aerodynamics and Propulsion
Research Laboratory

Heather A. Malcom
Heather A. Malcom, MTS
Computation and Data
Processing Center

Approved by: J. G. Logan
J. G. Logan, Director
Aerodynamics and Propulsion
Research Laboratory

B. A. Troesch
B. A. Troesch
Department Head,
Computation and Data
Processing Center

AEROSPACE CORPORATION
El Segundo, California

ABSTRACT

A modal solution of the flutter and vibration problems for a multiple component system is presented. The formulation requires the aerodynamic characteristics in the form of aerodynamic influence coefficients, and the structural and inertial characteristics in the form of free vibration modes and frequencies and a mass matrix for each component. The use of a rigid-body modal matrix permits a general analysis for a system free in space with up to six rigid-body degrees of freedom. The solution also utilizes a newly developed subroutine for finding the eigenvalues and eigenvectors of a non-Hermitian matrix having close roots, and obtains the corresponding values of frequency, required structural damping, and velocity of the flutter system.

The Aerospace IBM 7090 Computer Program No. LD014A provides the flutter or vibration solution for a system composed of as many as 20 flexible components, each of which may have up to 50 control points. The solution is carried out by a modal (series) method choosing a series of up to 40 cantilever and/or free-free vibration modes to approximate the control point deflections. The program may be used to find the vibration characteristics of a composite system if the cantilever vibration characteristics of the components are known, and the generalized masses may be obtained for use in subsequent modal analyses of flying qualities or stability and control of flexible vehicles. An option is provided to vary the density as well as the reduced velocity in the flutter analysis.

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SYMBOLS

A	Element of aerodynamic force matrix, lb
a	Element of normalized flexibility matrix, in/lb
a_F	Amplitude coefficient of cantilever free vibration modal series, in.
a_G	Amplitude coefficient of generalized centroidal coordinates
a_R	Generalized amplitude coefficient of rigid-body modal series
a_{RF}	Element of rigid component modal matrix
b_r	Reference semichord, ft
C_h	Element of oscillatory aerodynamic influence coefficient matrix
F	Number of control points at which the modes are specified
g	Structural damping coefficient
h_o	Control point deflection due to rigid-body motion, in.
h_l	Control point deflection, in.
K	Flexibility matrix normalizing constant
K_F	Element of generalized spring matrix
k	Element of cantilever spring matrix, lb/in.
k_r	Reference reduced frequency
M	Element of mass matrix, lb
\bar{M}	Element of complex mass matrix including aerodynamic effects, lb
m	Element of generalized mass matrix
\bar{m}	Element of sum of generalized mass and aerodynamic matrices
N	Number of assumed flexible modes

SYMBOLS (Continued)

Q	Element of generalized aerodynamic force matrix
R	Number of rigid-body degrees of freedom
s	Reference semispan, ft (i. e., span measured from root to tip)
V	Velocity, knots
W	Element of aerodynamic weighting matrix
x, y	Cartesian coordinates
ϵ	Error of series solution
ρ	Air density, slugs/ft ³
Ω	Eigenvalue, (cps) ⁻²
ω	Frequency, cps
()'	Prime denotes use of free-free modes
[]	Square matrix
{ }	Column matrix
[]	Diagonal matrix
[] ^T	Transposed matrix
[] ⁻¹	Inverse matrix

SECTION I

FORMULATION OF PROBLEM

A. Introduction

The use of structural influence coefficients in a collocation method of vibration analysis and of both structural and aerodynamic influence coefficients in a collocation flutter analysis (both of which are reviewed in Ref. 1) provides a number of advantages in terms of accuracy and convenience. However, the computation time in handling very large matrices becomes prohibitive, while computer capacity also may limit the analysis of extremely large systems, and a series or modal solution must be used. A modal formulation of the flutter problem using aerodynamic influence coefficients has been presented in Ref. 2. The present report reviews the derivations in Ref. 2 and extends its computational aspects, particularly in regard to the simplifications that accompany an option for a modal vibration analysis and an option to vary the density as well as the reduced velocity in the flutter analysis. The present program also utilizes a recently developed subroutine for finding the eigenvalues and eigenvectors of a non-Hermitian matrix having close roots, and its requirement for modes and frequencies is consistent with those generated by the vibration option of Ref. 1.

B. Sign Convention

The NASA body axis system with the x, y, and z axes directed forward, starboard, and downward, respectively, is recommended for consistency with the formulation of the static aeroelastic problems in Ref. 3. However, the usual flutter convention with the x, y, and z axes directed aft, starboard, and downward, respectively, may be used instead. In either case, the vertical normal force and deflection are positive downward.

C. Derivation of Equations

It is convenient to present two derivations of the modal formulation of the flutter and vibration problems according to whether cantilever or

free-free modes are to be utilized. The design analysis normally will make use of cantilever modes, but often the final analysis will be able to take advantage of the free-free modes available from ground vibration measurements. The two derivations are only slightly different; nevertheless it is useful to separate them.

The modal formulation may be derived equally well from an assumed series for the deflection modes and either Lagrange's equations for the dynamic system or the collocation (lumped parameter) equations of motion and Galerkin's method. Since the collocation equations of motion have been developed in Ref. 1, it is convenient to choose the latter approach. It is further convenient to summarize the significant features in the derivation of Ref. 1 to provide the basis for the present derivation. In Ref. 1, a single type of coordinate, viz., the deflection h , has been chosen as an adequate measure of both the deformation and the free-stream disturbance, not only for simplicity in the resulting equations but also because deflections have a more general meaning on a cambered vehicle and deflection influence coefficients are more readily obtained from a structural analysis than slope (or twist) influence coefficients. The resulting deformation matrix equation becomes

$$\{h_1\} - \{h_0\} = K[a](\{F_i\} + \{F_a\}) \quad (1)$$

where $\{h_1\}$ is the set of components of the absolute deflections of the control points, $\{h_0\}$ is the set of components of the deflections of the control points due to the rigid-body motion of some reference point, $[a]$ is the set of structural influence coefficients (SICs, or flexibility matrix) for the system cantilevered from (or otherwise restrained at) the reference point, $\{F_i\}$ is the set of inertial force components integrated throughout the region adjacent to each control point, $\{F_a\}$ is the set of aerodynamic force components integrated over the vehicle surface adjacent to each control point, and the scalar K has been introduced as a factor to the SICs for convenience in investigating variations in stiffness levels.

The inertial forces may be written in terms of a mass matrix $[M]$ and the control point accelerations.

$$\{F_i\} = -(1/386)[M]\{\ddot{h}_1\} \quad (2)$$

where the diagonal elements of the mass matrix are found from integrating the structural mass density throughout the region adjacent to the control points. (N. B. The mass matrix need not be diagonal, and, in general, will not be so if the elements must be derived from a set of weight data previously lumped at a system of control points different from those required in the aeroelastic analysis. The use of a coupled mass matrix permits simulation of given inertial characteristics at a set of control points frequently dictated by more difficult aerodynamic considerations. A special technique for deriving the elements of a coupled mass matrix corresponding to a specified set of control points from a discrete set of given weight data is suggested in Appendix B of Ref. 1.)

The aerodynamic forces may be written in terms of a set of aerodynamic influence coefficients (AICs) for oscillatory motion and the control point deflections as

$$\{F_a\} = (4\pi^2/12) \rho \omega^2 b_r^2 s [W][C_h]\{h_1\} \quad (3)$$

where $[C_h]$ is the theoretically derived dimensionless (complex) matrix of oscillatory AICs, ω is the frequency of the assumed harmonic motion, ρ is the atmospheric density, b_r is the reference semichord, s is the reference span, and $[W]$ is an empirically derived weighting matrix for modification of the theoretical AICs. A method for obtaining the elements of the weighting matrix has been suggested in Ref. 4.

The sum of the force components may now be written from Eqs. (2) and (3) for the case of harmonic motion.

$$\{F_i\} + \{F_a\} = (4\pi^2\omega^2/386)([M] + 32.174\rho b_r^2s[W][C_h])\{h_1\} \quad (4a)$$

$$= (4\pi^2\omega^2/386)([M] + [A])\{h_1\} \quad (4b)$$

$$= (4\pi^2\omega^2/386)[\bar{M}]\{h_1\} \quad (4c)$$

We next discuss the manner of inclusion of the rigid-body degrees of freedom in Eq. (1). The matrix $\{h_o\}$ has been defined as the set of components of the deflections of the control points due to the rigid-body motion of the reference point. Each component of the control point deflections h_o is linearly related to the rigid-body translations and rotations provided the rotations are small. Therefore, we may define a rigid-body modal matrix $[h_R]$ as the transformation

$$\{h_o\} = [h_R]\{a_R\} \quad (5)$$

where $\{a_R\}$ is the set of amplitudes of rigid-body translations and rotations of the reference point. As an example, if we consider symmetrical vertical motion, $[h_R]$ is composed of two columns: The first is a unit column corresponding to the plunging mode; the second consists of the x-coordinate of each control point corresponding to the pitching mode; $\{a_R\}$ is composed of two elements: The first is the plunging displacement z_o ; the second is the pitching angular displacement θ .

Before proceeding in the derivation, we should review the format of the various matrices in the case of a multiple flexible component system. As an example, we consider a symmetrical flutter analysis of an aircraft whose wing, aft fuselage, and tail are flexible and whose forward fuselage may be assumed to be rigid. We assume that the reference point (the cantilever

point) can be located in the vehicle such that its various components are independent. If we choose a point at the intersection of the wing and fuselage, then the wing is independent of the aft fuselage-tail combination, but the tail and aft fuselage must be considered together. The motion of the rigid forward fuselage is determined by the motion of the reference point, and the forward fuselage will not enter into any flexible considerations but only into the free-free boundary conditions. From the foregoing, it is seen that the various matrices will appear in partitioned form. If we denote the wing and aft fuselage-tail system by the subscripts 1 and 2, respectively, then the flexibility matrix appears as

$$[a] = \begin{bmatrix} a_1 & 0 \\ 0 & a_2 \end{bmatrix} \quad (6)$$

the mass matrix as

$$[M] = \begin{bmatrix} M_1 & 0 \\ 0 & M_2 \end{bmatrix} \quad (7)$$

the weighting matrix as

$$[W] = \begin{bmatrix} W_1 & 0 \\ 0 & W_2 \end{bmatrix} \quad (8)$$

the AICs as

$$[C_h] = \begin{bmatrix} C_{h1} & 0 \\ 0 & C_{h2} \end{bmatrix} \quad (9)$$

and the rigid body modal matrix as

$$[h_R] = \begin{bmatrix} h_{R1} \\ \text{---} \\ h_{R2} \end{bmatrix} \quad (10)$$

Two requirements should be emphasized with regard to the AICs. The first concerns the proper inclusion of the reference geometry associated with the nondimensional AICs. The dimensional form of Eq. (9) may be written

$$b_r^2 s [C_h] = \left[\begin{array}{c|c} b_1^2 s_1 C_{h1} & 0 \\ \text{---} & \text{---} \\ 0 & b_2^2 s_2 C_{h2} \end{array} \right] \quad (11)$$

where b_r and s are the reference semichord and span of the composite system, b_1 and s_1 are the reference geometry for the first component, and b_2 and s_2 are the reference geometry for the second component. The second requirement is that the AICs for each component must be determined for the same "dimensional" reduced velocity V/ω . If the reference reduced velocity is

$$1/k_r = V/b_r \omega \quad (12)$$

then the reduced velocity for the first component must be

$$1/k_1 = (1/k_r)(b_r/b_1) \quad (13)$$

and for the second component

$$1/k_2 = (1/k_r)(b_r/b_2) \quad (14)$$

Both of these requirements can be met in formulating the composite AICs by choosing the same reference geometry in determining the AICs for each component.

The rigid-body modal matrix provides the basis for a general statement of the boundary conditions for the free-free flutter of the composite system. The boundary conditions for harmonic motion may be written as

$$[h_R]^T [\bar{M}] \{h_1\} + [\Delta \bar{m}] \{a_R\} = \{0\} \quad (15)$$

where $[\Delta \bar{m}]$ is an incremental generalized mass matrix, including aerodynamic effects, of any rigid component of the system attached to the reference point (e. g. , the forward fuselage that was assumed to be rigid in the foregoing example), and not considered in the formulation of the flexible component mass and aerodynamic matrices. The form of the matrix $[\Delta \bar{m}]$ may be illustrated by the previous example with the rigid forward fuselage again in symmetrical motion. We may write

$$[\Delta \bar{m}] = [\Delta m] + [\Delta Q] \quad (16)$$

where the generalized rigid component mass matrix of the forward fuselage is

$$[\Delta m] = \begin{bmatrix} M_o & S_o \\ S_o & I_{yo} \end{bmatrix} \quad (17)$$

in which M_o , S_o , and I_{yo} are the mass, static unbalance about the reference point, and pitching moment of inertia about the reference point, respectively, of the forward fuselage, and the generalized aerodynamic forces on the forward fuselage (if not negligible) are found from

$$[\Delta Q] = 32.174 \rho b_o^2 s_o [h_{Ro}]^T [C_{ho}] [h_{Ro}] \quad (18)$$

where $[h_{Ro}]$ is the rigid-body modal matrix, $[C_{ho}]$ is the set of AICs, and b_o and s_o are the reference geometry for the forward fuselage. Again the AICs must be found for the reduced velocity of the composite system.

The collocation equation for flutter follows by substituting Eq. (4c) into Eq. (1) and adding the structural damping factor $1/(1 + ig)$ to the flexibility matrix to provide the artificial structural damping necessary to sustain the assumed harmonic motion of the flutter system

$$\{h_1\} - \{h_o\} = (4\pi^2 K \omega^2 / 386 (1 + ig)) [a][\overline{M}]\{h_1\} \quad (19)$$

The rigid-body degrees of freedom may be eliminated by use of Eqs. (5) and (15). For the purpose of seeking series solutions to Eq. (19), it is convenient to introduce a cantilever spring matrix defined by

$$[k] = (386/K)[a]^{-1} \quad (20)$$

and to rewrite the flutter equation as

$$\Omega[k](\{h_1\} - \{h_o\}) = 4\pi^2 [\overline{M}]\{h_1\} \quad (21)$$

where the eigenvalue $\Omega = (1 + ig)/\omega^2$.

This review of the derivation of the collocation equation and of the formats of the various matrices places us in a position to consider the two modal formulations of the flutter and vibration problems. We consider first the choice of a series of cantilever modes, and then the choice of a series of free-free modes.

Case I: Cantilever Modes

Let us denote by N the number of flexible modes given, by F the number of control points at which the modes are specified, and by R the number of rigid-body degrees of freedom. If we select a series of cantilever modes (which satisfy the requirement of the Galerkin method that all boundary conditions be met) for the solution of Eq. (21) then the series may be written

$$\{h_1\} - \{h_0\} = [h_F]\{a_F\} \quad (22)$$

where $[h_F]$ is the matrix of given cantilever modes and the $\{a_F\}$ are the modal amplitudes. The matrix $[h_F]$ appears in a partitioned form (size $F \times N$) in which each partition corresponds to a component of the composite system

$$[h_F] = \left[\begin{array}{c|c} h_{F1} & 0 \\ \hline 0 & h_{F2} \end{array} \right] \quad (23)$$

Substituting Eqs. (5) and (22) into Eq. (15) yields

$$[\bar{m}_{RF}]\{a_F\} + [\bar{m}_{RR}]\{a_R\} = 0 \quad (24)$$

where

$$[\bar{m}_{RF}] = [m_{RF}] + [Q_{RF}] \quad (25a)$$

$$= [h_R]^T [M] [h_F] + [h_R]^T [A] [h_F] \quad (25b)$$

$$[\bar{m}_{RR}] = [m_{RR}] + [Q_{RR}] \quad (26a)$$

$$= ([h_R]^T [M] [h_R] + [\Delta m]) + ([h_R]^T [A] [h_R] + [\Delta Q]) \quad (26b)$$

and from which

$$\{a_R\} = -[\bar{m}_{RR}]^{-1}[\bar{m}_{RF}]\{a_F\} \quad (27)$$

The flutter equation, Eq. (21), now becomes

$$\Omega[k][h_F]\{a_F\} = 4\pi^2[\bar{M}](\{h_F\}\{a_F\} + \{h_R\}\{a_R\}) + \{\epsilon\} \quad (28)$$

where $\{\epsilon\}$ is the error in the series solution. Applying the Galerkin principle ($[h_F]^T\{\epsilon\} = 0$) yields

$$\Omega[K_F]\{a_F\} = 4\pi^2([\bar{m}_{FF}]\{a_F\} + [\bar{m}_{FR}]\{a_R\}) \quad (29)$$

where

$$[K_F] = [h_F]^T[k][h_F] \quad (30)$$

$$[\bar{m}_{FF}] = [m_{FF}] + [Q_{FF}] \quad (31a)$$

$$= [h_F]^T[M][h_F] + [h_F]^T[A][h_F] \quad (31b)$$

$$[\bar{m}_{FR}] = [m_{FR}] + [Q_{FR}] \quad (32a)$$

$$= [m_{RF}]^T + [h_F]^T[A][h_R] \quad (32b)$$

If we eliminate the rigid-body degrees of freedom by substituting Eq. (27) into Eq. (29), we obtain the final form of the cantilever modal formulation

$$\Omega[K_F]\{a_F\} = 4\pi^2([\bar{m}_{FF}] - [\bar{m}_{FR}][\bar{m}_{RR}]^{-1}[\bar{m}_{RF}])\{a_F\} \quad (33)$$

or in the canonical form

$$\Omega\{a_F\} = [K_F/4\pi^2]^{-1}([\bar{m}_{FF}] - [\bar{m}_{FR}][\bar{m}_{RR}]^{-1}[\bar{m}_{RF}])\{a_F\} \quad (34)$$

We note that the generalized spring matrix is diagonal because our formulation is not statically coupled (it is dynamically coupled instead). The generalized spring matrix may be found in terms of the generalized free vibration mass matrix and the natural (normal or uncoupled) frequencies

$$[K_F/4\pi^2] = [\text{diagonal elements of } [m_{FF}]] [\omega_F^2] \quad (35)$$

The flutter equation, Eq. (34), can be solved by the power method for the eigenvalues, $\Omega = \Omega_R + i\Omega_I$, from which we find the frequency

$$\omega = 1/\sqrt{\Omega_R} \quad (\text{cps}) \quad (36)$$

the required structural damping

$$g = \Omega_I/\Omega_R \quad (37)$$

and, since the aerodynamic matrix required the assumption of a reduced velocity, $1/k_r$, for its formulation, the velocity

$$V = 0.5921(2\pi\omega b_r)(1/k_r) \quad (\text{knots}) \quad (38)$$

The power method as developed for non-Hermitian matrices with close roots is discussed in Appendix A of Ref. 1. The flutter mode is found from the originally assumed series

$$\{h_1\} = [h_F]\{a_F\} + [h_R]\{a_R\} \quad (39a)$$

$$= ([h_F] - [h_R][\bar{m}_{RR}]^{-1}[\bar{m}_{RF}])\{a_F\} \quad (39b)$$

From a series of solutions of Eq. (34) for various reduced velocities the conventional required damping versus velocity stability curve can be constructed for a specific altitude and the flutter point is determined as the velocity for which the required damping and actual damping are equal. An alternative approach to the flutter analysis is based on a single representative reduced velocity and a series of solutions of Eq. (34), carried out for various densities. The density at which the required damping and actual damping are equal may be used to find a stiffness-altitude similarity parameter for flutter from which the flutter stability may be determined. However, at present, the validity of this latter approach requires further investigation, particularly the sensitivity of the results to the choice of representative reduced velocity.

The generalized mass corresponding to each free vibration mode is also of interest in various modal analyses of flying qualities, stability and control characteristics, or transient response of flexible vehicles. If Eq. (34) is solved for the free vibration modes (by deleting the aerodynamic terms), then the n'th generalized mass is given by*

$$m^{(n)} = \{h_1^{(n)}\}^T [M] \{h_1^{(n)}\} + \{a_R^{(n)}\}^T [\Delta m] \{a_R^{(n)}\} \quad (40a)$$

*N. B. When the vibration modes $\{h_1^{(n)}\}$ are normalized on their maximum elements, the modal amplitudes $\{a_F^{(n)}\}$ cannot be normalized but must be scaled accordingly for the calculation of the generalized masses.

$$\begin{aligned}
m^{(n)} &= \{a_F^{(n)}\}^T [m_{FF}] \{a_F^{(n)}\} + \{a_R^{(n)}\}^T [m_{RR}] \{a_R^{(n)}\} \\
&\quad + \{a_R^{(n)}\}^T [m_{RF}] \{a_F^{(n)}\} + (\{a_R^{(n)}\}^T [m_{RF}] \{a_F^{(n)}\})^T
\end{aligned} \tag{40b}$$

$$= \{a_F^{(n)}\}^T ([m_{FF}] \{a_F^{(n)}\} + [m_{RF}]^T \{a_R^{(n)}\}) \tag{40c}$$

where $\{a_F^{(n)}\}$ is the n'th set of cantilever modal amplitudes and the corresponding rigid component mode is found from Eq. (27) (again without the aerodynamic terms).

$$\{a_R^{(n)}\} = -[m_{RR}]^{-1} [m_{RF}] \{a_F^{(n)}\} \tag{41}$$

Case II: Free-Free Modes

When free-free vibration data are available for a flutter analysis, it is necessary to write two series expressions that provide the basis for the modal solution.

$$\{h_1\} = [h_F'] \{a_F'\} + [h_R] \{a_G\} \tag{42}$$

$$\{a_R\} = [a_{RF}] \{a_F'\} + \{a_G\} \tag{43}$$

where the prime denotes the free-free condition, $\{a_G\}$ is the matrix of centroidal generalized coordinates, and $[a_{RF}]$ is the rigid component modal matrix. The rigid component modal matrix may be found from the boundary conditions for the free vibration. However, it is desirable to permit the consideration of some cantilever modes in addition to the orthogonal free-free modes. Hence, if we write

$$[h'_F] = [h'_{Fo} \quad h'_{Fc}] \quad (44)$$

then

$$[a_{RF}] = [a_{RFo} \quad 0] \quad (45)$$

where the subscripts o and c denote orthogonal and cantilever, respectively. The orthogonal partition follows from the free vibration boundary condition and is given by

$$[a_{RFo}] = -[\Delta m]^{-1} [h_R]^T [M] [h'_{Fo}] \quad (46)$$

Making the series substitutions, Eqs. (42) and (43), into the boundary condition, Eq. (15), yields

$$[h_R]^T [\bar{M}] ([h'_F] \{a'_F\} + [h_R] \{a_G\}) + [\Delta \bar{m}] ([a_{RF}] \{a'_F\} + \{a_G\}) = 0 \quad (47)$$

from which

$$\{a_G\} = -[\bar{m}_{RR}]^{-1} [\bar{m}'_{RF}] \{a'_F\} \quad (48)$$

where

$$[\bar{m}'_{RF}] = [h_R]^T [\bar{M}] [h'_F] + [\Delta \bar{m}] [a_{RF}] \quad (49)$$

In order to apply the Galerkin principle when free-free modes are used in the series solution, it is necessary to obtain the equation of motion of the rigid

component. If the flexible system mass term is eliminated between Eqs. (15) and (21), then the rigid component equation of motion is obtained.

$$-\Omega[h_R]^T[k]\{h_1\} - \{h_0\} = 4\pi^2[\Delta\bar{m}]\{a_R\} \quad (50)$$

Substituting the series solutions, Eqs. (42) and (43), into Eq. (50) yields

$$\Omega[h_R]^T[k][h_F'']\{a_F'\} = 4\pi^2[\Delta\bar{m}](\{a_{RF}\}\{a_F'\} + \{a_G\}) + \{\epsilon_R\} \quad (51)$$

where $\{\epsilon_R\}$ is the error in Eq. (50) due to the series solution and

$$[h_F''] = [h_F'] - [h_R][a_{RF}] \quad (52)$$

Substituting the series solutions into the flexible system equation of motion, Eq. (21), yields

$$\Omega[k][h_F'']\{a_F'\} = 4\pi^2[\bar{M}](\{h_F'\}\{a_F'\} + [h_R]\{a_G\}) + \{\epsilon_F\} \quad (53)$$

where $\{\epsilon_F\}$ is the error in Eq. (21) due to the series solution. Applying the Galerkin principle (in this case, $[a_{RF}]^T\{\epsilon_R\} + [h_F']^T\{\epsilon_F\} = 0$) yields the generalized equation of motion

$$\Omega[K_F']\{a_F'\} = 4\pi^2([\bar{m}'_{FF}]\{a_F'\} + [\bar{m}'_{FR}]\{a_G\}) \quad (54)$$

where

$$[K_F'] = [h_F'']^T[k][h_F''] \quad (55)$$

$$[\bar{m}'_{FF}] = [m'_{FF}] + [Q'_{FF}] \quad (56a)$$

$$= ([h'_F]^T [M] [h'_F] + [a_{RF}]^T [\Delta m] [a_{RF}]) \\ + ([h'_F]^T [A] [h'_F] + [a_{RF}]^T [\Delta Q] [a_{RF}]) \quad (56b)$$

$$[\bar{m}'_{FR}] = [h'_F]^T [\bar{M}] [h_R] + [a_{RF}]^T [\Delta \bar{m}] \quad (57)$$

If we eliminate the centroidal degrees of freedom by substituting Eq. (48) into Eq. (54), we obtain the final form of the free-free modal formation

$$\Omega [K'_F] \{a'_F\} = 4\pi^2 ([\bar{m}'_{FF}] - [\bar{m}'_{FR}] [\bar{m}_{RR}]^{-1} [\bar{m}'_{RF}]) \{a'_F\} \quad (58)$$

or in the canonical form

$$\Omega \{a'_F\} = [K'_F / 4\pi^2]^{-1} ([\bar{m}'_{FF}] - [\bar{m}'_{FR}] [\bar{m}_{RR}]^{-1} [\bar{m}'_{RF}]) \{a'_F\} \quad (59)$$

We again note that the diagonal generalized spring matrix is found from the generalized mass matrix and natural frequencies

$$[K'_F / 4\pi^2] = [\text{diagonal elements of } [m'_{FF}]] [\omega_F'^2] \quad (60)$$

The solution of the flutter equation by iteration has been discussed in the previous case. The flutter mode follows from the assumed series. Substituting Eq. (48) into Eq. (42) yields

$$\{h_1\} = ([h'_F] - [h_R] [\bar{m}_{RR}]^{-1} [\bar{m}'_{RF}]) \{a'_F\} \quad (61)$$

The n'th generalized mass follows from the solution of Eq. (59) with the aerodynamic terms deleted, and Eqs. (40a), (42), and (43).*

$$m^{(n)} = \{a_F^{(n)}\}^T [m'_{FF}] \{a_F^{(n)}\} + \{a_G^{(n)}\}^T [m_{RR}] \{a_G^{(n)}\} + \{a_G^{(n)}\}^T [m'_{RF}] \{a_F^{(n)}\} + (\{a_G^{(n)}\}^T [m'_{RF}] \{a_F^{(n)}\})^T \quad (62)$$

$$= \{a_F^{(n)}\}^T ([m'_{FF}] \{a_F^{(n)}\} + [m'_{RF}]^T \{a_G^{(n)}\}) \quad (62a)$$

where

$$\{a_G^{(n)}\} = -[m_{RR}]^{-1} [m'_{RF}] \{a_F^{(n)}\} \quad (63)$$

D. References

1. W. P. Rodden, E. F. Farkas and H. A. Malcom. "Flutter and Vibration Analysis by a Collocation Method: Analytical Development and Computational Procedure." Aerospace Corporation Report No. TDR-169(3230-11)TN-14, 31 July 1963.
2. W. P. Rodden, E. F. Farkas, P. E. Williams, and F. C. Slack. "Flutter Analysis by a Modal Method Using Aerodynamic Influence Coefficients: Analytical Development and Procedure for the IBM 7090 Computer." Norair Division, Northrop Corporation Report NOR-61-54, 14 April 1961.

* N. B. When the vibration modes $\{h_1^{(n)}\}$ are normalized on their maximum elements, the modal amplitudes $\{a_F^{(n)}\}$ cannot be normalized but must be scaled accordingly for the calculation of the generalized masses.

3. W. P. Rodden, E. F. Farkas, and H. A. Malcom. "Quasi-Static Aero-Thermo-Elastic Analysis: Analytical Development and Computational Procedure." Aerospace Corporation Report No. TDR-169(3230-11)TN-8, 1 March 1963.

4. W. P. Rodden. "An Empirical Weighting Matrix for Use with Aerodynamic Influence Coefficients in Aeroelastic Analyses." Northrop Corporation Report NOR-59-320, 1 April 1959.

5. R. L. Bisplinghoff, H. Ashley, and R. L. Halfman. Aeroelasticity. Reading: Addison-Wesley Publishing Company, Inc., 1955.

SECTION II

GENERAL DESCRIPTION OF INPUT

A. Units

The dimensional data required for each component consist of the mass matrix in pounds, the free vibration frequencies in cycles per second, and the reference semichords and semispans in feet. The free vibration modes and the aerodynamic influence coefficients are dimensionless. Compatibility is required between the units of the rigid-body modal matrices and the generalized rigid-body mass matrix; for example, if S_o and I_{y_o} are in the pound-foot system, then the x-coordinates, which correspond to the pitching mode, must be measured in feet, whereas if the generalized masses are given in the pound-inch system, the x-coordinates must be measured in inches. The density is required in slugs per cubic foot.

B. Classes of Data and Example Problems

Five classes of data must be provided: mass, aerodynamic influence coefficients (AICs) and their weighting matrices, free vibration modes (cantilever and/or free-free) and corresponding frequencies, rigid-body modes, and the generalized masses for any rigid component in the system. (The cantilever case does not require the rigid-body modes or the generalized masses).

Two example problems will be considered. Each problem will be a symmetrical flutter analysis of the jet transport wing treated as an example throughout Ref. 5. In the first analysis, we will use cantilever vibration modes and frequencies, and in the second we use free-free symmetric modes and frequencies such as might be obtained from a ground vibration survey. In each problem, we assume that four modes will be sufficient to obtain convergence in the flutter solution.

The four flexible cantilever modes (calculated by the collocation method of Ref. 1) are

$$[h_F] =$$

0.045185	-0.163556	-0.219686	0.291415
0.042502	0.544990	-0.243887	0.279638
0.148256	-0.279631	-0.205463	0.306804
0.143727	1.0	-0.362884	0.353608
0.379411	-0.226740	-0.371824	-0.559460
0.376733	0.946563	-0.428613	-0.490161
0.692995	-0.138889	0.203824	-0.146405
0.692655	0.887880	0.163356	-0.020285
0.998969	-0.039737	1.0	0.837729
1.0	0.822370	0.981915	1.0

The corresponding frequencies are

$$\begin{aligned}\omega_1 &= 2.036802 \text{ cps} \\ \omega_2 &= 3.552589 \text{ cps} \\ \omega_3 &= 7.280460 \text{ cps} \\ \omega_4 &= 11.698543 \text{ cps}\end{aligned}$$

The four free-free symmetric modes (also found by the method of Ref. 1) are

$$[h_F] =$$

-0.129721	-0.091809	-0.035610	0.305979
-0.131332	0.568102	-0.047615	0.287803
-0.017537	-0.221627	-0.162942	0.442843
-0.015998	1.0	-0.189775	0.503783
0.247806	-0.215371	-0.449387	-0.561462
0.250893	0.905541	-0.477740	-0.443473
0.622665	-0.205131	0.106948	-0.225098
0.627940	0.776147	0.097215	-0.006219
0.993742	-0.186900	0.984408	0.741022
1.0	0.636687	1.0	1.0

and the corresponding frequencies are

$$\begin{aligned}\omega_1 &= 2.442962 \\ \omega_2 &= 3.603113 \\ \omega_3 &= 8.553045 \\ \omega_4 &= 12.707073\end{aligned}$$

The symmetrical rigid component (one-half of the fuselage) generalized mass matrix is given below in the pound-inch system.

$$[\Delta m] = \begin{bmatrix} M_o & S_o \\ S_o & I_{yo} \end{bmatrix} = \begin{bmatrix} 17,400 & 1,370,250 \\ 1,370,250 & 4,457,907,200 \end{bmatrix}$$

(In the antisymmetric case, the rigid component generalized mass matrix is $[\Delta m] = [I_{xo}]$; in the composite symmetric-antisymmetric case, the generalized mass matrix is

$$[\Delta m] = \begin{bmatrix} M_o & S_o & 0 \\ S_o & I_{yo} & 0 \\ 0 & 0 & I_{xo} \end{bmatrix} \quad .)$$

The mass and aerodynamic matrices for the wing and the aerodynamic matrix for the fuselage are shown in the example problem printed output (pages 47-73) and will not be repeated here. The aerodynamic weighting matrix is taken as unity, and no input is required.

In addition to the flexible modes, the symmetrical analysis requires the two rigid-body modes for the degrees of freedom of plunging and pitching. The rigid-body modal matrix therefore consists of two columns: a unit column and a column of the x-coordinate of each control point. The rigid-body modal matrix for the wing is

$$[h_R] = [1 \ x] = \begin{bmatrix} 1 & 20.25 \\ 1 & -81.00 \\ 1 & 17.85 \\ 1 & -71.40 \\ 1 & 15.80 \\ 1 & -63.20 \\ 1 & 13.30 \\ 1 & -53.20 \\ 1 & 11.05 \\ 1 & -44.20 \end{bmatrix}$$

The rigid-body modal matrix for the fuselage is

$$[h_{Ro}] = [1 \ x] \begin{bmatrix} 1 & -373.30 \\ 1 & -248.30 \\ 1 & -123.30 \\ 1 & +1.70 \\ 1 & +126.70 \end{bmatrix}$$

Note that the above matrix is used in computing the incremental complex generalized mass which results from the aerodynamic loads on one-half of the fuselage and, therefore, the x-coordinates must be given in the proper order for the control points used in computing the fuselage AICs. (In this problem the fuselage AICs are hypothetical but the x-coordinates are given in the order necessary for the slender-body theory AICs referred to later (page 39) in this report.)

The reference geometry for the wing is $b_{rw} = 5.468$ ft and $s_w = 37.917$ ft. The reference geometry for the fuselage is $b_{ro} = 5.468$ ft and $s_o = 18.9585$ ft. (We have assumed that the wing reference geometry was used to nondimensionalize the fuselage AIC matrix, and since we need only one-half of the fuselage aerodynamic forces they can be obtained by setting $s_o = (1/2)s_w$.) The flutter analyses are carried out for the reduced velocity $1/k_r = 16.67$ using the reference semichord for the system $b_r = 5.468$ ft. Both analyses are carried out for sea level; i.e., $\rho = 0.002378$ slugs per cubic foot.

The results for each of the two problems are shown in the example printed output, Section IV, Part A. The final results are shown also in the cases of 6-, 8-, and 10-mode solutions to demonstrate the convergence of the four-mode solution. We note that the four-mode convergence is very poor, emphasizing the importance of checking convergence in any modal solution. We also note that the 10-mode solution agrees exactly with the collocation solution shown in Ref. 1 as is to be expected for a system with 10 degrees

of freedom. A final point of interest is to note that the use of free-free input modes does not provide any better convergence than the cantilever modes.

C. Program Restrictions and Options

1. The maximum number of vibration modes that can be input for the system is 40.
2. The maximum number of cantilever vibration modes is 40 whether the system consists of one or more components.
3. The maximum number of free-free vibration modes for the system is 40 minus the total number of cantilever modes to be used for the system. Since free-free modes represent the entire flexible system, a portion of each free-free mode must be input corresponding to each flexible component.
4. The number of flexible components is ≤ 20 .
5. The number of values used in the reference reduced velocity ($1/k_r$) series is ≤ 20 .
6. The maximum number of values used in a density series is ≤ 20 .
7. The program provides for varying the densities with each $1/k_r$ or for using the same densities with all $1/k_r$'s.
8. The maximum number of output modes is 25; however, the number cannot exceed the number of input vibration modes.
9. The maximum number of control points for the rigid component or any flexible surface is 49.
10. The number of rigid-body modes is ≤ 6 .
11. It is possible to reserve a partition in the upper left-hand corner of any flexible component AIC matrix for control points whose aerodynamic forces may be neglected or found from some alternate theory to that used for the primary control points. This partition is termed the "external stores" region since external stores are an example of a source of additional control

points requiring such special consideration. The maximum number of control points that can be reserved on each surface for external stores is ≤ 48 .

12. A weighting matrix is optional input for each surface. The order of this matrix must agree with the order of the AIC matrices for the particular surface.

13. Any number of complete sets (decks) of input data may be stacked and run in one machine pass.

SECTION III

DATA DECK SETUP

A. Loading Order

Input data decks are loaded behind column binary deck No. LD014A. The data decks are assembled using cards punched from keypunch forms and cards that are punched-card output from appropriate IBM programs. The data items in each deck have the following order with this exception: some of the items may be omitted if indicators used in the control cards (Items 2 and 4) specify their absence.

1. Title card.
2. Data deck general control card.
3. Data card for changes in matrix iteration tests.
4. Control card(s) for external stores and weighting matrices.
5. Reference semichord (b_r) and reference reduced velocities ($1/k_r$'s).
6. Reference semichords (b_{ri}) and reference semispans (s_i) for rigid and flexible components (surfaces).
7. Density series (if same densities are used for all $1/k_r$'s).
8. Generalized mass matrix ($[\Delta m]$) for rigid component.
9. Mass matrix ($[M]_i$) for each flexible surface.
10. Rigid-body modal matrix ($[h_{Ro}]$) for rigid component.
11. Rigid-body modal matrix ($[h_R]_i$) for each flexible surface.
12. Frequencies (ω_F) and vibration modal matrix ($[h_F]_i$) for each flexible surface.
13. Weighting matrices ($[W]_i$).

14. Aerodynamic input repeated for each $1/k_r$
 - a. Density series cards (if densities vary with each $1/k_r$),
 - b. AIC matrix ($[C_{ho}]$) for rigid component (if present),
 - c. AIC matrix ($[C_h]_i$) for each flexible surface.

B. Input Data Description

1. The title card may contain any alphanumeric characters desired in Columns 2 through 80. Column 1 should be blank.
2. Data deck general control card (FORMAT 18I4): The first 11 fields in this card contain, in the order defined, the following indicators:

NSUR: Number of flexible components (surfaces), ≤ 20 .

NAERO: Number of reference reduced velocities, ≤ 20 ;
NAERO = 0 is used for the vibration analysis.

NRIGID: Number of rigid-body motion modes to be input (= number of columns in $[h_R]$); NRIGID = 0 for the cantilever case.

NFUS: NFUS must = 1 if AICs ($[C_{ho}]_j$, $j = 1, NAERO$) are input for the rigid component. NFUS = 0 if $[C_{ho}]$ is not input.

NDENS: If NDENS = 0, the densities are to vary with each $1/k_r$ and are input as part of Item 14; if NDENS > 0, this number of densities must be input as Item 7 and each density value will be used for all $1/k_r$'s.

MODES: Number of output modes, ≤ 25 .

NFREE: Number of free-free vibration modes to be input, $\leq (40 - NCANT)$, (see next indicator). Note that NFREE does not vary with the number of flexible components (surfaces) but each free-free mode will have NSUR partitions and each partition will consist of the control point deflections for one surface. NFREE = 0 if only cantilever modes are used.

NCANT: Total number of cantilever modes to be input for the system. $\leq (40 - \text{NFREE})$

NDELM: NDELM = 0 if no rigid component generalized mass matrix ($[\Delta m]$) is input; NDELM = 1 if $[\Delta m]$ is input.

NPUNCH: This indicator is used to obtain a printout of the dynamic matrix ($[U]$) and to obtain the frequencies and modes in punched-card format. NPUNCH = 0 if no printout of $[U]$ or punched output is desired; NPUNCH = 1 if only punched-card output is desired and NPUNCH = -1 will provide for the printout of $[U]$ and the punched output. (The minus must be in Column 37 and the 1 in Column 40.)

NCON: This indicator provides for changing five program test numbers used in the matrix iteration subroutine. NCON = 0 if no changes are desired and NCON \neq 0 if any of the tests are to be changed.

3. Data card for changes in matrix iteration tests (FORMAT 3E12.8 and 2I4): There are three test numbers and two control numbers associated with the iteration procedure (Ref. 1, Appendix A) that are present in the program but that may be changed by the program user. The numbers built into the program are given below. An explanation of its purpose follows each number.

EPSP = 0.5×10^{-6} ; test for eigenvector convergence when the iteration procedure is approaching a single root.

EPDP = 0.5×10^{-7} ; test for convergence when the iteration procedure is approaching a pair of close roots.

AITKEN = 0.9; if this test is met the program uses a procedure (known as Aitken's δ^2 method) to accelerate convergence.

NITRSP = 40; a maximum of 40 single-precision arithmetic iterations will be performed for each eigenvalue if its eigenvectors have not converged in a lesser number.

NITRDP = 100; if the eigenvectors for any one eigenvalue have not converged in NITRSP single-precision iterations the program will then use up to a maximum of 100 double-precision arithmetic iterations.

Altering any of the above numbers (NCON \neq 0, Item 2) requires entering all five numbers on one card even if some of the numbers are to be the same as shown above. Input EPSP, EPDP, and AIKEN in the first three E12.8 fields on the card, the next two I4 fields (Columns 37 through 40 and 41 through 44) contain NITRSP and NITRDP, respectively. The EPSP and EPDP tests present in the program represent six-significant-figure accuracy although some loss in significance will occur as the higher modes are found; increasing these numbers would relax the requirements for accuracy and decrease the computing time. The AITKEN test must be less than but not too close to unity; an optimum convergence rate has been observed for values around 0.9. The NITRSP and NITRDP can be increased to the limit of the I4 fields, and either control may be decreased to zero. Errors in input data often create convergence problems, therefore all input should be carefully checked before an increase is made in the maximum number of iterations.

ITEMS 4, 5, 6, 7, 10, 13, and 14 DESCRIBED BELOW SHOULD BE OMITTED
WHEN NAERO = 0 IS USED IN ITEM 2 (Vibration Analysis).

4. Control card(s) for external stores and weighting matrices (FORMAT 18I4): The consecutive fields in these cards contain ISXT₁, ISW₁, ISXT₂, ISW₂, ..., ISXT_{NSUR}, ISW_{NSUR}. ISXT_i = number of control points reserved for the external stores on surface i; if no external stores are present on surface i, enter zero for ISXT_i or leave its field blank.

$ISW_i = 0$ (or blank field) if no weighting matrix is to be input for surface i , $ISW_i = 1$ if a weighting matrix will be input for surface i . Include only the number of cards needed for $i = 1$, NSUR.

5. Reference semichord, b_r and $1/k_r$ (sometimes denoted by $V/b_r \omega$) series (FORMAT 6E12.8): These reference parameters are used in computing the flutter velocities [Eq. (38)]. The b_r (feet) may be any value, but the proper relationship must exist between the $1/k_r$'s used for reference reduced velocities and the $1/k_{ri}$ used when computing the AIC matrices for each surface; that is, $1/k_{ri} = (1/k_r)(b_r/b_{ri})$ where b_{ri} is the reference semichord for surface i . To input these parameters enter b_r in Field 1 of the first card; Field 2 and consecutive fields in the first and successive cards (if NAERO > 5) contain the $1/k_r$'s, ≤ 20 .

6. The b_{ri} and s_i series (FORMAT 6E12.8): A reference semichord (b_{ri}) and a reference semispan (s_i) must be input for the rigid component if NFUS = 1 and for each flexible component if NAERO > 0. When NFUS = 1 the consecutive fields in the first and successive (if NSUR > 2) card(s) contain in order b_{r0} , s_0 , b_{r1} , s_1 , ..., b_{rNSUR} , and s_{NSUR} . When NFUS = 0, the consecutive fields in the necessary card(s) contain in order b_{r1} , s_1 , b_{r2} , s_2 , ..., b_{rNSUR} , and s_{NSUR} .

7. Density series (FORMAT 6E12.8): Omit this input if NDEN = 0 in Item 2. If NDEN > 0, begin in Field 1 of the first card and enter NDEN densities, ≤ 20 .

8. Rigid component generalized mass matrix $[\Delta m]$ (FORMAT 6E12.8): (Omit this matrix when NDELM = 0 in Item 2.) The size of the $[\Delta m]$ matrix for the rigid component(s) of the system must be compatible with the corresponding matrix for the flexible components given by $[h_R]^T [M] [h_R]$; i. e., each element in $[\Delta m]$ is based upon the same rigid-body generalized coordinates as the respective element in the product matrix. If NDELM = 1, input the $[\Delta m]$ matrix (size NRGID \times NRGID) by column with each column beginning on a new card.

9. Mass matrix, $[M]$: Although the $[M]$ matrix defined in Section I, Part C, has $NSUR \times NSUR$ partitions, only the nonzero partitions are input; i. e., a separate mass matrix ($[M]_i$) is input for each surface. The sequence for considering the surfaces was established in Items 4 and 6. Repeat the following input for $i = 1, NSUR$.

a. Size control card $_i$ (FORMAT 18I4):

Field 1 contains $NSIZE_i$, the order of $[M]_i$ (external stores control points + surface control points, ≤ 49).

b. Control card(s) for omitting zeros in the coupled mass matrix $[M]_i$ (FORMAT 18I4):

Field 1 and consecutive fields in the first and successive cards ($NSIZE > 9$) contain in order $LOW_1, LHIGH_1, LOW_2, LHIGH_2, \dots, LOW_{NSIZE}, LHIGH_{NSIZE}$. LOW_j is the row number in which the first nonzero appears in Column j , and $LHIGH_j$ is the row number in which the last nonzero element appears in Column j . If only one nonzero element is in Column j , the row number in which it appears must be used for both LOW_j and $LHIGH_j$.

c. Elements in $[M]_i$ matrix (FORMAT 6E12.8): The elements are entered by column as shown below; the input for each column starts on a new card.

Column 1: elements LOW_1 through $LHIGH_1$.

Column 2: elements LOW_2 through $LHIGH_2$.

⋮

Column $NSIZE$: elements LOW_{NSIZE} through $LHIGH_{NSIZE}$.

Any zero elements appearing in rows between LOW_j and $LHIGH_j$ must be entered or their respective fields left blank. If all elements in a column are zero, one zero element must be entered for this column (a blank card

may be used) giving $LOW_j = LHIGH_j$ for this element in the above control cards. If external stores are present ($ISXT_i > 0$, Item 4) all store control points must be considered before the surface control points; i. e., the elements representing the external stores mass must occupy the upper left-hand corner of the mass matrix.

NOTE: One possible method for determining the $[M]$ matrices is suggested in Appendix B of Ref. 1.

10. Rigid component modal matrix, $[h_{Ro}]$: (Omit this input when $NFUS = 0$, Item 2.) The rigid-body modes for the rigid component are input in order to compute the incremental generalized force matrix $[\Delta Q]$ that results from aerodynamic loads on the component. The number of rows in $[h_{Ro}]$ must be the same as the number of control points considered when computing the $[C_{ho}]$ matrices, the number of columns must agree with $NRIGID$, Item 2. These data are input in the following manner.

a. Size card (FORMAT 18I4):

Field 1 contains $NROWS$, number of rows in $[h_{Ro}]$.

b. Elements in $[h_{Ro}]$ (FORMAT 6E12.8): Tabulate the matrix elements by column with each column beginning on a new card.

11. Rigid-body modal matrix, $[h_R]$: (Omit this input if $NRIGID = 0$, Item 2.) If $NRIGID > 0$, input the $[h_R]$ matrix by partitions; i. e., input $[h_R]_i$ size ($NSIZE_i \times NRIGID$) for each surface. The $[h_R]_i$ matrix requires the following input, repeating the input for $i = 1, NSUR$.

a. Size control card $_i$ (FORMAT 18I4):

Field 1 contains $NSIZE_i$, the number of rows (≤ 49) in $[h_R]_i$ (must agree with order of $[M_i]$).

b. $[h_R]_i$ matrix elements (FORMAT 6E12.8): Enter the matrix elements by column with each column beginning on a new card.

12. Vibration modes ($[h_F]$ matrix) and frequencies: The $[h_F]$ matrix is also partitioned and a separate $[h_F]_i$ matrix is considered for each surface. The columns in $[h_F]_i$ are the free vibration modes (orthogonal free-free and/or cantilever) for surface i . The total number of modes (NFREE + NCANT) used for the system cannot exceed 40. The frequency (cps) associated with each mode must also be input.

Input the frequencies and modal column matrices ($[h_F]_i$) in the following manner: if NFREE = 0 repeat Parts a, b, and c (below) for $i = 1, \text{NSUR}$; if NFREE > 0 omit Part b for $i = 2, \text{NSUR}$.

a. Control card $_i$ (FORMAT 18I4):

Field 1 contains NSIZE $_i$, the number of rows in $[h_F]_i$, ≤ 49 (must agree with order of $[M]_i$).

Field 2 contains NMODES, number of columns in $[h_F]_i$ (NFREE + NCANT $_i$). NCANT $_i$ is the number of cantilever modes to be input for surface i .

Field 3 contains NFORM; NFORM = 1 if the frequencies and the $[h_F]_i$ matrix elements are to be input using FORTRAN (FORMAT 6E12.8); NFORM = 0 if these data are input using column binary format.

b. Frequencies (ω_F series $_i$) (Format in above control card $_i$):

If only cantilever modes are used for the system, an ω_F series is input following each control card $_i$. The ω_F 's are listed in the order in which their respective modal columns will appear in $[h_F]_i$. (Number of ω_F 's = NMODES, above card.)

If NFREE > 0, $i = 1$ for ω_F series $_i$, that is, only one ω_F series will be input for the system and this series must follow control card $_1$ (control card for the first surface). The number (≤ 40) of ω_F 's in

the series must equal $N_{FREE} + N_{CANT}$ used in Item 2. The ω_F 's are listed in the order in which their respective modal columns appear in the unpartitioned $[h_F]$ matrix.

The ω_F series are input using the format specified in control card i ; $N_{FORM} = 1$ (FORMAT 6E12.8) permits entering six ω_F 's per card; use only the number of cards necessary to complete the series; $N_{FORM} = 0$ (column binary format) should be used only if the ω_F 's and modes are available as punched-card output from a suitable vibration program. When this format is used, the first ω_F starts in Origin 1; the number of ω_F 's per card ≤ 22 and the ω_F card(s) ≤ 2 are followed by a TRA card.*

c. $[h_F]_i$ matrix elements (Format given in control card i):
When only cantilever modes are used for the system, the modal column order used in any one $[h_F]_i$ is independent of the order used for any other $[h_F]_i$, but if any free-free modes are used ($N_{FREE} > 0$, Item 2), each free-free mode is partitioned for N_{SUR} surfaces and the appropriate part of the mode must have the same column location in each $[h_F]_i$ matrix. The free-free modes must appear as the first N_{FREE} columns in each $[h_F]_i$ matrix, the remaining columns (if any) are the cantilever modes peculiar to surface i . Use the format given in control card i and input $[h_F]_i$ by columns. If $N_{FORM} = 1$, begin each column on a new card and use the number of cards (≤ 9) necessary to complete the column. If $N_{FORM} = 0$, each column begins in Origin 1 of a new card, the number of elements per card ≤ 22 , and the card(s) ≤ 3 for each column are followed by a TRA card.

NOTE: Punched cards output from the IBM program based on Ref. 1 may be used for the above Parts a, b, and c with no alterations if the output data

*The TRA card has a 7 and a 9 punch in Column 1, Columns 2 through 72 are blank, and Columns 73 through 80 will contain the characters used for identification and sequencing in the punched-card output deck.

(cantilever or free-free modes and frequencies) are suitable for any component or the complete system considered in the modal analysis.

13. Weighting matrix, $[W]_i$, $i = 1$, NSUR: (No provisions have been made for entering a $[W]$ matrix for the rigid component; any adjustments to $[C_{ho}]$ must be made before it is input.) The presence or absence of a $[W]$ matrix for surface i is determined by ISW_i , Item 4; if $ISW_i = 1$, a $[W]$ matrix must be input for surface i . When the matrix is input, it consists of two parts if control points for external stores have been reserved ($ISXT_i > 0$, Item 4). The input for $[W]_i$ matrix is as follows, with this exception: omit Parts a and b when $ISXT_i = 0$.

- a. Control card i for external stores $[W]_i$ partition (FORMAT 1814):

Field 1 contains $NXST_i$; $NXST_i = 0$ if no $[W]_i$ matrix is input for the external stores area (the program will use a unit matrix, $[I]$); $NXST_i$ = number of control points reserved ($= ISXT_i$) if the partition is input.

Field 2 is a dummy field and may be left blank.

Field 3 contains $NFORM$; $NFORM = 1$ if the $[W]_i$ matrix elements will be input using FORTRAN (FORMAT 6E12.8); $NFORM = 0$ if the matrix elements will be input using column binary format.

Field 4 contains $NROW$; $NROW = 0$ if the $[W]_i$ matrix elements will be input by column; $NROW = 1$ if the matrix elements will be input by row.

- b. Elements in external stores $[W]_i$ partition (Format given in control card i): Omit this input if $NXST_i = 0$ in control card i . If $NXST_i > 0$, input the matrix elements using the format combination given in control card i ; that is, if $NFORM = 1$ and $NROW = 0$, tabulate the elements

by column and begin each column on a new card, and if NFORM = 1 and NROW = 1 tabulate the elements by row with each row starting on a new card. When the column binary format is used (NFORM = 0 and NROW = 0) Column 1 starts in card Origin 1, Column 2 in Location $(1 + IXST_1)$, Column 3 in Location $(1 + 2 \times ISXT_1)$, etc., and a TRA card must follow the input for this $[W]_i$ partition.

c. Control card $_i$ for component $[W]_i$ (FORMAT 18I4):

Field 1 contains NSIZE $_i$, number of control points used on surface i . (Do not include control points for external stores.)

Field 2 contains NPART, the number of partitions in the component $[W]_i$. NPART = 1 for a full (unpartitioned) matrix. (The $[W]_i$ matrix is often sparse, sometimes diagonal, and may be of rather large order, ≤ 49 ; for this reason, provision is made for partitioning the matrix and entering only the nonzero partitions.)

Field 3 contains NFORM; NFORM is defined above in Part a.

Field 4 contains NROW; NROW is also defined above in Part a.

The input below is repeated for NPART partitions ($j = 1, NPART$).*

d. Partition j size (FORMAT 18I4):

Field 1 contains the order of partition j (if NPART = 1, the order = NSIZE $_i$ used in above control card $_i$).

e. Elements in partition j (Format given in the above control card $_i$): The elements in each partition must be input using the format

* Part d is input only if NFORM = 1 (FORTRAN FORMAT) in the above control card; if NFORM = 0 (column binary format) the unpartitioned matrix is punched by columns and input as Part e.

combination given in Fields 3 and 4 of control card $_j$. See the explanation given in Part b (above) pertaining to the use of the format combinations.

14. Aerodynamic data: The aerodynamic input consists of NAERO (Item 2) sets of AICs. One set of AICs consists of the AICs for each surface which have the same reference $1/k_r$ (see Item 5). If NDENS = 0 (Item 2), a density series will precede each set of AICs. Input the density series (only if Item 7 was omitted) and the AIC matrices for each surface with the following input order. Repeat the order for each $(1/k)_j$, $j = 1, \text{NAERO}$.

a. Density series for $(1/k)_j$: (Omit this input if NDENS > 0.)

(a.1) Control card $_j$ (FORMAT 18I4):

Field 1 contains NRHO, number of densities to be entered for $(1/k)_j$, ≤ 20 .

(a.2) Densities (FORMAT 6E12.8): Use the number of cards (≤ 4) necessary to complete the series.

b. Rigid component $[C_{ho}]_j$ matrix: (Omit this input if NFUS = 0, Item 2.)

(b.1) Reference $1/k$ for $[C_{ho}]_j$ (FORMAT 6E12.8):

Field 1 in this card normally contains the $1/k$ used in computing $[C_{ho}]_j$. The $1/k$ card is used only to help identify $[C_{ho}]_j$ in the input deck and in the printed output; it may be a blank card, but the AICs in $[C_{ho}]_j$ must be computed for a $1/k$ which properly relates them to the j 'th $1/k_r$. (See Items 5 and 6.)

(b.2) Control card $_j$ for $[C_{ho}]_j$ (FORMAT 18I4):

Field 1 contains NSIZE, number of control points on the rigid component (must agree with NSIZE used for $[h_{Ro}]$).

Field 2 contains NPART, the number of partitions in $[C_{ho}]_j$; NPART = 1 is used for an unpartitioned matrix.

Field 3 contains NFORM as defined in Item 13.a.

Field 4 contains NROW also defined in Item 13.a.

The input listed below must be repeated for NPART partitions.*

(b. 3) Partition k size, $k = 1, \text{NPART}$ (FORMAT 18I4):

Field 1 in this card contains N, the order of partition k. If NPART = 1, $N = \text{NSIZE}$ used in control card j .

(b. 4) Elements in partition k (Format given in control card j): All elements in the AIC matrices are complex numbers; but each partition is input as though it is a real matrix, size $N \times 2N$. The imaginary parts of the elements form the even-numbered columns. Input each partition in $[C_{ho}]_j$ with the format combination specified in Fields 3 and 4 of the above control card. The format combinations are explained in Item 13.b.

c. $[C_h]_{ij}$ matrix for flexible surface i: The input order for $[C_h]_{ij}$ will differ from the order given for $[C_{ho}]_j$ only if external stores control points are reserved on surface i ($\text{ISXT}_i > 0$, Item 4). The input given below must be repeated for $i = 1$; NSUR and Parts c. 1 and c. 2 are omitted when no external stores are present on surface i.

*Part b. 3 is input only if NFORM = 1 (FORTRAN FORMAT) in the above control card; if NFORM = 0 (column binary format) the unpartitioned AIC matrix is punched by columns and input as Part b. 4.

- (c. 1) Control card $_{ij}$ for external stores $[C_h]_{ij}$ partition (FORMAT 18I4):

Field 1 contains $NXST_i = 0$ if no $[C_h]_{ij}$ partition will be input. (The program sets the partition to zero.) $NXST_i$ = number of control points reserved for external stores if the external stores $[C_h]_{ij}$ is to be input.

Field 2 is a dummy field and may be left blank.

Field 3 contains NFORM as defined in Item 13. a.

Field 4 contains NROW as defined in Item 13. a.

- (c. 2) Elements in external stores $[C_h]_{ij}$ partition (Format in above control card $_{ij}$): The elements in the partition are complex numbers and will be input in the same manner as the elements in the $[C_{ho}]_j$ partitions (Part b. 4, above), with the exception of format which is given in the above control card.

- (c. 3) Reference l/k for surface i (FORMAT 6E12.8):
(See Part b. 1, above.)

- (c. 4) Control card $_{ij}$ for surface $[C_h]_{ij}$ partition (FORMAT 18I4): (See Part b. 2, above.)

The input below must be repeated for NPART partitions.*

- (c. 5) Partition k size, $k = 1, NPART$ (FORMAT 18I4):
(See Part b. 3, above).

- (c. 6) Elements in partition k : (Format in above control card, Part c. 4).

* See footnote page 37.

Note: Punched cards output from IBM programs based on the sources listed below may be used (with no alterations) for Item 14, Parts b.1, b.2, b.3, and b.4 or Parts c.3, c.4, c.5, and c.6 when the theory is appropriate to the problem. Punched cards output from the program based on slender-body theory may be used for Item 14, Parts c.1 and c.2 if the first card ($1/k_r$) is removed from the deck. The list shown is complete only at this writing; additional compatible sources are currently under development.

W. P. Rodden, E. F. Farkas, H. Malcom, and A. M. Kliszewski.
"Aerodynamic Influence Coefficients from Incompressible Strip Theory: Analytical Development and Computational Procedure." Aerospace Corporation Report No. TDR-169(3230-11)TN-5, 3 September 1962.

W. P. Rodden, E. F. Farkas, H. Malcom, and A. M. Kliszewski.
"Aerodynamic Influence Coefficients from Supersonic Strip Theory: Analytical Development and Computational Procedure." Aerospace Corporation Report No. TDR-169(3230-11)TN-1, 1 August 1962.

W. P. Rodden, E. F. Farkas, H. Malcom, and A. M. Kliszewski.
"Aerodynamic Influence Coefficients from Piston Theory: Analytical Development and Computational Procedure." Aerospace Corporation Report No. TDR-169(3230-11)TN-2, 15 August 1962.

W. P. Rodden, E. F. Farkas, and G. Y. Takata. "Aerodynamic Influence Coefficients from Slender-Body Theory: Analytical Development and Computational Procedure." Aerospace Corporation Report No. TDR-169(3230-11)TN-6, 31 October 1962.

C. Example Keypunch Forms

Example keypunch forms are given on the following pages. Columns 73 through 80 are reserved for data deck identification and sequencing. Only the cards with sequencing are used in the two data decks set up for the example problems; the lines with Columns 73 through 80 left blank are for description of input data.

1	2	3	4	5	6	7	8	9	10	11	12	13	14	15	16	17	18	19	20	21	22	23	24	25	26	27	28	29	30	31	32	33	34	35	36	37	38	39	40	41	42	43	44	45	46	47	48	49	50	51	52	53	54	55	56	57	58	59	60	61	62	63	64	65	66	67	68	69	70	71	72	73	74	75	76	77	78	79	80	
Title Card																																																																																
JET TRANSPORT EXAMPLE PROBLEM - CANTILEVER INPUT MODES																																																																																
General Control Card (FORMAT 1814)																																																																																
1	1	2	1	1	4	0	4	1	0	0																																																																						
ISXT _i and ISW _i , i = 1, NSUR (FORMAT 1814)																																																																																
0	0																																																																															
b _r and (1/k _r) _i Series (FORMAT 6E12.8)																																																																																
5	4	6	8			+16.67																																																																										
b _r _i and s _r _i , i = 0, 1, NSUR (FORMAT 6E12.8)																																																																																
5	4	6	8			18.9585		5.468		37.917																																																																						
Density Series (FORMAT 6E12.8)																																																																																
+2	3	7	8			-02																																																																										
[Δm] (FORMAT 6E12.8)																																																																																
1	7	4	0			+05		137025		+07																																																																						
1	3	7	0			+07		44579072		+10																																																																						
NSIZE _i for [M] _i (FORMAT 1814)																																																																																
1	0																																																																															
LOW _j and LHIGH _j for Each Column in [M] _i , j = 1, NSIZE _i (FORMAT 1814)																																																																																

1	2	3	4	5	6	7	8	9	10	11	12	13	14	15	16	17	18	19	20	21	22	23	24	25	26	27	28	29	30	31	32	33	34	35	36	37	38	39	40	41	42	43	44	45	46	47	48	49	50	51	52	53	54	55	56	57	58	59	60	61	62	63	64	65	66	67	68	69	70	71	72	73	74	75	76	77	78	79	80																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																													
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SECTION IV
PROGRAM OUTPUT

A. Printed Output

1. All input data.
2. The generalized mass matrices ($[m_{FF}]_i$ or $[m'_{FF}]$) corresponding to the input free vibration modal matrices ($[h_F]_i$ or $[h'_F]$).
3. The dynamic matrix if NPUNCH = -1 is used in the general control card.
4. For the vibration analysis or for each $1/k_r$ in the flutter analysis:
 - a. The eigenvalue for each output mode followed by the number of single- and double-precision iterations and the number of Aitken accelerations.
 - b. The eigenvectors for each mode followed by the check eigenvalues and eigenvectors.
 - c. The frequencies in cycles per second and, in a flutter analysis, the structural damping coefficient and the velocity (knots) associated with each frequency.
 - d. The final flutter or vibration modes for each component normalized on the largest element in each mode.
 - e. If NPUNCH = ± 1 , the sequencing numbers used in identifying the punched cards containing the frequencies and the final modes will appear below the respective printed data.
5. In a vibration analysis, the generalized mass corresponding to each output (free vibration) mode will follow the above printed output [see Eqs. (40) and (62)].
6. A number of different statements may be printed indicating machine- or program-detected errors in input data such as wrong format or incompatibility in the number of rows in $[h_R]_i$, $[M]_i$, and $[h_F]_i$.

7. If the program or the machine fails in the iteration portion of the program, a note will be printed stating the type or cause of failure. Some of the various program stops and machine failures are explained in Ref. 1, Appendix A.

8. If convergence is not obtained in the allowable number of iterations, a note will be printed and the program will continue. (In this case, the eigenvalues and eigenvectors should be compared with the check eigenvalues and eigenvectors to determine if the convergence is sufficiently accurate.)

9. The printed output for the two example problems is shown on the following pages.

JET TRANSPORT EXAMPLE PROBLEM-CANTILEVER INPUT MODES
FLUTTER ANALYSIS BY A MODAL METHOD USING AEROYNAMIC INFLUENCE COEFFICIENTS

HMI4-0001

NSUR = 1 NAERO = 1 NRIGIO = 2 NFUS = 1 NOENS = 1 MOES OUT = 4 NFREE = 0 NCANT = 4

CANTILEVER INPUT MODES

B RIGIO COMPONENT = 0.54680000E 01 S RIGIO COMPONENT = 0.18958499E 02

B (REF) = 0.54680000E 01

SURFACE B S EXTERNAL STORES SIZE

1 0.54680000E 01 0.37916999E 02 0

RIGID COMPONENT MASS MATRIX

COLUMN 1 COLUMN 2

1 0.17399999E 05 0.13702500E 07
2 0.13702500E 07 0.44579072E 10

MASS MATRIX									
SURFACE 1,		10 CONTROL POINTS							
COLUMN 1	COLUMN 2	COLUMN 3	COLUMN 4	COLUMN 5	COLUMN 6	COLUMN 7	COLUMN 8	COLUMN 9	COLUMN 10
1 0.53835999E 04	-0.13490000E 03	0.	0.	0.	0.	0.	0.	0.	0.
2 -0.13490000E 03	0.92520000E 03	0.	0.	0.	0.	0.	0.	0.	0.
3 0.	0.	0.20732000E 05	-0.11004999E 05	0.	0.	0.	0.	0.	0.
4 0.	0.	-0.11004999E 05	0.11477999E 05	0.	0.	0.	0.	0.	0.
5 0.	0.	0.	0.	0.31139000E 04	0.13970000E 03	0.	0.	0.	0.
6 0.	0.	0.	0.	0.13970000E 03	0.80660000E 03	0.	0.	0.	0.
7 0.	0.	0.	0.	0.	0.	0.	0.	0.	0.
8 0.	0.	0.	0.	0.	0.	0.	0.	0.	0.
9 0.	0.	0.	0.	0.	0.	0.	0.	0.	0.
10 0.	0.	0.	0.	0.	0.	0.	0.	0.	0.
RIGID COMPONENT MODES,		5 CONTROL POINTS.							
COLUMN 1		COLUMN 2							
1 0.09999999E 01	-0.37329999E 03	0.	0.	0.	0.	0.	0.	0.	0.
2 0.09999999E 01	-0.24829999E 03	0.	0.	0.	0.	0.	0.	0.	0.
3 0.09999999E 01	-0.12329999E 03	0.	0.	0.	0.	0.	0.	0.	0.
4 0.09999999E 01	0.16999999E 01	0.	0.	0.	0.	0.	0.	0.	0.
5 0.09999999E 01	0.12670000E 03	0.	0.	0.	0.	0.	0.	0.	0.

RIGID BODY MODAL MATRIX
SURFACE 1, 10 CONTROL POINTS

	COLUMN 1	COLUMN 2	COLUMN 10
1	0.09999999E 01	0.20249999E 02	
2	0.09999999E 01	-0.80999999E 02	
3	0.09999999E 01	0.17850000E 02	
4	0.09999999E 01	-0.71399999E 02	
5	0.09999999E 01	0.15799999E 02	
6	0.09999999E 01	-0.63199999E 02	
7	0.09999999E 01	0.13299999E 02	
8	0.09999999E 01	-0.53200000E 02	
9	0.09999999E 01	0.11049999E 02	
10	0.09999999E 01	-0.44199999E 02	

FLEXIBLE MODAL MATRIX
SURFACE 1, 10 CONTROL POINTS

4 FREQUENCIES

	COLUMN 1	COLUMN 2	COLUMN 3	COLUMN 4	COLUMN
1	0.45184767E-01	-0.16355576E-00	-0.21968562E-00	0.29141474E-00	
2	0.42502320E-01	0.54498983E 00	-0.24388658E-00	0.27963846E-00	
3	0.14825594E-00	-0.27963124E-00	-0.30546310E-00	0.30680391E-00	
4	0.14372702E-00	0.09999999E 01	-0.36288369E-00	0.35360781E-00	
5	0.37941088E-00	-0.22673982E-00	-0.37182412E-00	-0.55946033E 00	
6	0.37673284E-00	0.94656356E 00	-0.42861269E-00	-0.49016142E-00	
7	0.69299459E 00	-0.13888866E-00	0.20382352E-00	-0.14640462E-00	
8	0.69265460E 00	0.88787997E 00	0.16335642E-00	-0.20284958E-01	
9	0.99896891E 00	-0.39737375E-01	0.09999999E 01	0.83772944E 00	
10	0.09999999E 01	0.82236990E 00	0.98191517E 00	0.09999999E 01	

WEIGHTING MATRIX
SURFACE 1, NO WEIGHTING MATRIX

	COLUMN 1	COLUMN 2	COLUMN 3	COLUMN 4	COLUMN
1	0.31500891E 04	0.15182495E-02	0.91362000E-03	0.90789795E-03	
2	0.14839172E-02	0.21329428E 05	-0.84495544E-03	-0.67901611E-03	
3	0.91552734E-03	-0.89263916E-03	0.27328113E 04	-0.32424927E-03	
4	0.90980530E-03	-0.59700012E-03	-0.29563904E-03	0.334004605E 04	

		RIGID COMPONENT AERO MATRIX,						5 CONTROL POINTS			
		COLUMN 1	COLUMN 2	COLUMN 3	COLUMN 4	COLUMN 5	COLUMN 6				
1		-0.19883168E 01	-0.71028437E 01	-0.31782072E 01	0.99733821E 01	0.17128096E 01	-0.24681993E 01				
2		0.20347044E 01	-0.30576088E 01	-0.89710928E 01	0.67589525E 00	0.25888789E 01	0.32829072E 01				
3		0.	0.	0.25888789E 01	-0.32829072E 01	-0.89710928E 01	-0.67589525E 00				
4		0.	0.	0.	0.	0.18608618E 01	-0.23997266E 01				
5		0.	0.	0.	0.	0.19459838E-00	0.73566824E 00				
1		COLUMN 7	COLUMN 8	COLUMN 9	COLUMN 10						
2		0.	0.	0.	0.						
3		0.20347044E 01	0.30576088E 01	0.	0.						
4		-0.59600403E 01	-0.46448669E-00	0.10709520E 01	0.17538263E 01						
5		0.45456574E-00	-0.31600294E 01	-0.19732837E 01	0.81508698E 00						

	COLUMN 19	COLUMN 20	COLUMN
1	0.	0.	
2	0.	0.	
3	0.	0.	
4	0.	0.	
5	0.	0.	
6	0.	0.	
7	0.	0.	
8	0.	0.	
9	-0.30521601E 03	0.28517172E 02	
10	0.15971068E-00	-0.84300261E 01	

OUTPUT DATA

FLUTTER ANALYSIS BY A MODAL METHOD USING AERODYNAMIC INFLUENCE COEFFICIENTS

DENSITY = 0.23779999E-02 REDUCED VELOCITY = 0.16670000E 02

2 RIGID BODY DEGREES OF FREEDOM, 4 INPUT MODES (TOTAL).

MODE	EIGENVALUE	ITERATIONS	S.P.	D.P.	AITKENS S.P.	D.P.
1	0.19004809E-00	-0.13066806E-00	11	0	3	0
2	0.13305187E-00	-0.90475378E-03	8	0	2	0
3	0.50936355E-02	-0.12321895E-01	11	0	3	0
4	0.89039633E-02	-0.15425324E-02	2	0	0	0

EIGENVECTORS

COLUMN	1	COLUMN	2	COLUMN	3	COLUMN	4	COLUMN	5	COLUMN	6
1	0.09999999E 01	0.	0.09999999E 01	-0.	0.09999999E 01	-0.	0.23126761E-00	0.55586841E 00			
2	-0.52882797E-02	0.26563946E-02	0.75557064E-02	-0.41647033E-01	0.75557064E-02	-0.41647033E-01	0.18934382E-01	0.35449203E-02			
3	0.46232747E-01	-0.26897484E-01	0.20981800E-01	-0.40106017E-01	0.20981800E-01	-0.40106017E-01	0.09999999E 01	0.			
4	-0.15751113E-02	-0.54060456E-02	-0.10034602E-01	-0.32852156E-02	-0.10034602E-01	-0.32852156E-02	0.35542952E-00	-0.13060754E-00			
	COLUMN 7	COLUMN 8	COLUMN								
1	-0.50736564E 00	-0.28387121E-00									
2	0.31773394E-01	0.13081321E-01									
3	0.09999999E 01	0.									
4	-0.15375598E-00	-0.25941110E-00									

CHECK EIGENVALUES AND EIGENVECTORS

0.19204808E-00	-0.13066801E-00	0.13305186E-00	-0.90471179E-03	0.50936326E-02	-0.12321895E-01
0.89039642E-02	-0.15425376E-02				
COLUMN 1	COLUMN 2	COLUMN 3	COLUMN 4	COLUMN 5	COLUMN 6
1 0.09999999E 01	0.	0.09999999E 01	0.	0.23126674E-00	0.55586720E 00
2 -0.52882672E-02	0.26563954E-02	0.75557120E-02	-0.41647052E-01	0.18934361E-01	0.35450576E-02
3 0.46232741E-01	-0.26897469E-01	0.20981800E-01	-0.40105984E-01	0.09999999E 01	0.
4 -0.15751122E-02	-0.54060404E-02	-0.10034598E-01	-0.32852045E-02	0.35542956E-00	-0.13060750E-00
COLUMN 7	COLUMN 8	COLUMN			
1 -0.50736279E 00	-0.28387356E-00				
2 0.31773206E-01	0.13081333E-01				
3 0.09999999E 01	0.				
4 -0.15375577E-00	-0.25941128E-00				

MODE OMEGA (CPS) DAMPING VELOCITY (KNOTS)

1	0.22938669E 01	-0.68755261E 00	0.77786975E 03
2	0.27415078E 01	-0.6800076E-02	0.92966856E 03
3	0.14011545E 02	-0.24190768E 01	0.47514340E 04
4	0.10597619E 02	-0.17324109E-00	0.35937425E 04

FINAL FLUTTER MODES

	SURFACE 1,		10 CONTROL POINTS		
	COLUMN 1	COLUMN 2	COLUMN 3	COLUMN 4	
1	-0.19844364E-00	0.78853738E-01	-0.93420087E-01	-0.97846834E-01	-0.41824473E-01
2	-0.21468721E-00	0.83936904E-01	-0.90291424E-01	-0.13011442E-00	-0.40065185E-01
3	-0.79901066E-01	0.77532722E-01	0.15015688E-01	-0.68443904E-01	-0.26054742E-02
4	-0.10292312E-00	0.85141466E-01	0.24666292E-01	-0.12520704E-00	-0.42543685E-02
5	0.19292901E-00	0.77061356E-01	0.27690162E-00	-0.16246606E-01	0.17809848E-00
6	0.17350294E-00	0.83907188E-01	0.28711402E-00	-0.67938515E-01	0.017440537E-00
7	0.59565143E 00	0.43911391E-01	0.63678547E 00	0.18358172E-01	0.14763121E-00
8	0.58159092E 00	0.49055874E-01	0.64766992E 00	-0.27171946E-01	0.13596687E-00
9	0.09999999E 01	-0.	0.98947141E 00	0.38904484E-01	0.18204748E-01
10	0.9054588E 00	0.33664029E-02	0.09999999E 01	0.	-0.

COLUMN

1	-0.50261029E-01
2	-0.55549506E-01
3	-0.16105451E-00
4	-0.18329110E-00
5	-0.35805747E-00
6	-0.38073238E-00
7	0.17717902E-00
8	0.17168011E-00
9	0.98417973E 00
10	0.09999999E 01

MODE	OMEGA (CPS)	DAMPING	VELOCITY (KNOTS)
1	1.00	0.00	0.00
2	1.00	0.00	0.00
3	1.00	0.00	0.00
4	1.00	0.00	0.00
5	1.00	0.00	0.00
6	1.00	0.00	0.00
7	1.00	0.00	0.00
8	1.00	0.00	0.00
9	1.00	0.00	0.00
10	1.00	0.00	0.00
11	1.00	0.00	0.00
12	1.00	0.00	0.00
13	1.00	0.00	0.00
14	1.00	0.00	0.00
15	1.00	0.00	0.00
16	1.00	0.00	0.00
17	1.00	0.00	0.00
18	1.00	0.00	0.00
19	1.00	0.00	0.00
20	1.00	0.00	0.00
21	1.00	0.00	0.00
22	1.00	0.00	0.00
23	1.00	0.00	0.00
24	1.00	0.00	0.00
25	1.00	0.00	0.00
26	1.00	0.00	0.00
27	1.00	0.00	0.00
28	1.00	0.00	0.00
29	1.00	0.00	0.00
30	1.00	0.00	0.00
31	1.00	0.00	0.00
32	1.00	0.00	0.00
33	1.00	0.00	0.00
34	1.00	0.00	0.00
35	1.00	0.00	0.00
36	1.00	0.00	0.00
37	1.00	0.00	0.00
38	1.00	0.00	0.00
39	1.00	0.00	0.00
40	1.00	0.00	0.00
41	1.00	0.00	0.00
42	1.00	0.00	0.00
43	1.00	0.00	0.00
44	1.00	0.00	0.00
45	1.00	0.00	0.00
46	1.00	0.00	0.00
47	1.00	0.00	0.00
48	1.00	0.00	0.00
49	1.00	0.00	0.00
50	1.00	0.00	0.00
51	1.00	0.00	0.00
52	1.00	0.00	0.00
53	1.00	0.00	0.00
54	1.00	0.00	0.00
55	1.00	0.00	0.00
56	1.00	0.00	0.00
57	1.00	0.00	0.00
58	1.00	0.00	0.00
59	1.00	0.00	0.00
60	1.00	0.00	0.00
61	1.00	0.00	0.00
62	1.00	0.00	0.00
63	1.00	0.00	0.00
64	1.00	0.00	0.00
65	1.00	0.00	0.00
66	1.00	0.00	0.00
67	1.00	0.00	0.00
68	1.00	0.00	0.00
69	1.00	0.00	0.00
70	1.00	0.00	0.00
71	1.00	0.00	0.00
72	1.00	0.00	0.00
73	1.00	0.00	0.00
74	1.00	0.00	0.00
75	1.00	0.00	0.00
76	1.00	0.00	0.00
77	1.00	0.00	0.00
78	1.00	0.00	0.00
79	1.00	0.00	0.00
80	1.00	0.00	0.00
81	1.00	0.00	0.00
82	1.00	0.00	0.00
83	1.00	0.00	0.00
84	1.00	0.00	0.00
85	1.00	0.00	0.00
86	1.00	0.00	0.00
87	1.00	0.00	0.00
88	1.00	0.00	0.00
89	1.00	0.00	0.00
90	1.00	0.00	0.00

MODE	OMEGA (CPS)	DAMPING	VELOCITY (KNOTS)
1	0.20945049E 01	-0.68642659E 00	0.710264438E 03
2	0.27472688E 01	0.42585590E-02	0.93162217E 03
3	0.48110338E 01	0.26282316E-00	0.16314623E 04
4	0.11158799E 02	-0.69910804E 00	0.37840436E 04
5	0.12543309E 02	-0.21431879E-00	0.42535426E 04

FINAL FLUTTER MODES

	SURFACE 1.			CONTROL POINTS		COLUMN 5	COLUMN 6
	COLUMN 1	COLUMN 2	COLUMN 3	COLUMN 4			
1	-0.22536646E-00	0.92572200E-01	-0.10430135E-00	-0.95571166E-01	-0.10669166E-00	-0.66641671E-02	
2	-0.24172064E-00	0.96812297E-01	-0.10552717E-00	-0.12630315E-00	-0.13305146E-00	-0.51599176E-02	
3	-0.10655960E-00	0.8932177E-01	0.75712425E-02	-0.67028530E-01	0.59504019E-01	-0.36529737E-01	
4	-0.12957601E-00	0.95967677E-01	0.97156047E-02	-0.12113088E-00	0.20609725E-01	-0.37150735E-01	
5	0.16918588E-00	0.8353867E-01	0.27401933E-00	-0.16638520E-01	0.42422621E-00	-0.10293659E-00	
6	0.14333673E-00	0.94240405E-01	0.28014200E-00	-0.67000752E-01	0.42554127E-00	-0.12906936E-00	
7	0.57887563E-00	0.49874357E-01	0.63759924E-00	0.19681936E-01	0.74136435E-00	-0.51632738E-01	
8	0.54986458E-00	0.66028306E-01	0.64893137E-00	-0.26138963E-01	0.792228053E-00	-0.10830468E-00	
9	0.09999999E-01	-0.	0.98852029E-00	0.39756269E-01	0.95080607E-00	-0.52753998E-01	
10	0.97653205E-00	0.13983364E-01	0.09999999E-01	0.	0.09999999E-01	-0.	
		COLUMN 7	COLUMN 8	COLUMN 9	COLUMN 10	COLUMN	
1	-0.48560822E-03	-0.97199512E-01	0.55720084E-00	0.71647885E-01	0.59014091E-01		
2	-0.54546969E-02	-0.86510264E-01	0.52352165E-00	0.16462570E-01	0.		
3	-0.91827672E-01	-0.14610815E-00	0.90787979E-00	-0.			
4	-0.97049317E-01	-0.15178238E-00	0.09999999E-01	-0.			
5	-0.47107571E-00	-0.21387362E-01	-0.11198328E-00	-0.33678662E-00			
6	-0.48660678E-00	-0.30987718E-01	-0.66459754E-01	-0.34726656E-00			
7	0.10400940E-00	0.10713105E-00	-0.39143629E-00	0.14244039E-00			
8	0.98177937E-01	0.96218284E-01	-0.39663463E-00	0.16197237E-00			
9	0.98493147E-00	0.29076613E-01	-0.28143807E-00	0.76281424E-00			
0	0.09999999E-01	0.	-0.26482295E-00	0.79094338E-00			

FOR EIGHT CANTILEVER INPUT MODES: REDUCED VELOCITY = 16.67

MODE	OMEGA (CPS)	DAMPING	VELOCITY (KNOTS)						
				FINAL FLUTTER MODES					
				SURFACE 1, 10 CONTROL POINTS					
				1	2	3	4	5	6
				COLUMN	COLUMN	COLUMN	COLUMN	COLUMN	COLUMN
1	0.20905757E 01	-0.69425995E 00	0.70893196E 03						
2	0.27376900E 01	-0.92221314E-02	0.92837390E 03						
3	0.46615379E 01	0.26804059E-00	0.15807670E 04						
4	0.79630522E 01	-0.79058085E 00	0.27003385E 04						
5	0.92712380E 01	-0.22886848E-00	0.31439554E 04						

FOR TEN CANTILEVER INPUT MODES: REDUCED VELOCITY = 16.67

MODE	OMEGA (CPS)	DAMPING	VELOCITY (KNOTS)
1	0.20648006E 01	-0.69443911E 00	0.70019143E 03
2	0.27409278E 01	-0.12059832E-01	0.92947187E 03
3	0.44713013E 01	0.27819017E-00	0.15162562E 04
4	0.69985107E 01	-0.61700474E 00	0.23732543E 04
5	0.95550425E 01	-0.26846905E-00	0.32401959E 04

FINAL FLUTTER MODES

COLUMN 1		SURFACE 2		SURFACE 1, 10 CONTROL POINTS		COLUMN 4		COLUMN 5		COLUMN 6	
COLUMN 1		COLUMN 2		COLUMN 3		COLUMN 4		COLUMN 5		COLUMN 6	
1	-0.22672141E-00	0.92801609E-01	-0.10980555E-00	-0.10249897E-00	-0.11565813E-00	-0.14201696E-00	0.10797655E-02				
2	-0.24322546E-00	0.96231766E-01	-0.11252964E-00	-0.13723268E-00	-0.14201696E-00	0.41647484E-03					
3	-0.10896499E-00	0.90128804E-01	0.36838599E-02	-0.72438644E-01	0.39346464E-01	-0.37219568E-01					
4	-0.13140173E-00	0.95914537E-01	0.38688469E-02	-0.12866598E-00	0.62708063E-02	-0.41508544E-01					
5	0.16390745E-00	0.84669594E-01	0.27385022E-00	-0.15872125E-01	0.38962284E-00	-0.14034666E-00					
6	0.13893208E-00	0.93805630E-01	0.27711197E-00	-0.70519595E-01	0.38136988E-00	-0.15814506E-00					
7	0.57352667E 00	0.51462835E-01	0.63861454E 00	0.20940823E-01	0.72148301E 00	-0.93108028E-01					
8	0.54571546E 00	0.66242959E-01	0.64748624E 00	-0.26460116E-01	0.75146683E 00	-0.13747214E-00					
9	0.09999999E 01	-0.09999999E-01	0.98693995E 00	0.36698003E-01	0.94384036E 00	0.67121437E-01					
10	0.97106527E 00	0.19206535E-01	0.09999999E 01	0.09999999E 01	0.09999999E 01	-0.09999999E 01					
COLUMN 7		COLUMN 8		COLUMN 9		COLUMN 10		COLUMN		COLUMN	
1	0.55524993E-01	-0.28636505E-01	0.10481205E-00	-0.75341270E-01							
2	0.37311652E-01	-0.33436552E-02	0.67771085E 00	-0.13639605E-00							
3	-0.61298352E-01	-0.81925987E-01	0.32304791E-00	-0.65253254E-01							
4	-0.72266892E-01	-0.58644155E-01	0.20867915E-00	-0.67861656E-01							
5	-0.53633323E 00	-0.13744034E-00	0.09999999E 01	-0.09999999E 01							
6	-0.51012327E 00	-0.16303784E-00	0.93896377E 00	0.10471679E-00							
7	-0.12072749E-00	-0.69964735E-01	-0.37512479E-00	0.53744053E 00							
8	-0.82757482E-01	-0.11890169E-00	-0.21398329E-00	0.38646627E-00							
9	0.09999999E 01	0.09999999E 01	-0.69760381E 00	-0.62525453E 00							
10	0.94309023E 00	-0.89771795E-02	-0.77933718E 00	-0.50264002E 00							

JET TRANSPORT EXAMPLE PROBLEM USING FREE-FREE INPUT MODES
FLUTTER ANALYSIS BY A MODAL METHOD USING AEROYNAMIC INFLUENCE COEFFICIENTS

HM140001

NSUR = 1 NAERO = 1 NRGIO = 2 NFUS = 1 NOENS = 1 MODES OUT = 4 NFREE = 4 NCANT = 0

FREE-FREE INPUT MODES, 0 CANTILEVER MODES INCLUDED.

B RIGID COMPONENT = 0.54680000E 01 S RIGID COMPONENT = 0.18958499E 02

B (REF) = 0.54680000E 01

SURFACE B S EXTERNAL STORES SIZE

1 0.54680000E 01 0.37916999E 02 -0

RIGID COMPONENT MASS MATRIX

	COLUMN 1	COLUMN 2
1	0.17399999E 05	0.13702500E 07
2	0.13702500E 07	0.44579072E 10

MASS MATRIX									
SURFACE 1,		10 CONTROL POINTS							
COLUMN	1	COLUMN	2	COLUMN	3	COLUMN	4	COLUMN	5
1	0.53835999E 04	-0.13490000E 03	0.	0.	0.	0.	0.	0.	0.
2	-0.13490000E 03	0.92520000E 03	0.	0.	0.	0.	0.	0.	0.
3	0.	0.	0.	0.20732000E 05	05	-0.11004999E 05	05	0.	0.
4	0.	0.	-0.11004999E 05	05	0.	0.11477999E 05	05	0.	0.
5	0.	0.	0.	0.	0.	0.	0.	0.31139000E 04	0.
6	0.	0.	0.	0.	0.	0.	0.	0.13970000E 03	0.13970000E 03
7	0.	0.	0.	0.	0.	0.	0.	0.13970000E 03	0.80660000E 03
8	0.	0.	0.	0.	0.	0.	0.	0.	0.
9	0.	0.	0.	0.	0.	0.	0.	0.	0.
10	0.	0.	0.	0.	0.	0.	0.	0.	0.
COLUMN		7	COLUMN	8	COLUMN	9	COLUMN	10	COLUMN
1	0.	0.	0.	0.	0.	0.	0.	0.	0.
2	0.	0.	0.	0.	0.	0.	0.	0.	0.
3	0.	0.	0.	0.	0.	0.	0.	0.	0.
4	0.	0.	0.	0.	0.	0.	0.	0.	0.
5	0.	0.	0.	0.	0.	0.	0.	0.	0.
6	0.	0.	0.	0.	0.	0.	0.	0.	0.
7	0.26387999E 04	-0.20999999E 02	0.	0.	0.	0.	0.	0.	0.
8	-0.20999999E 02	0.80329999E 03	0.	0.	0.	0.	0.	0.	0.
9	0.	0.	0.48750000E 03	0.	0.	0.	0.72999999E 01	0.	0.
10	0.	0.	0.72999999E 01	0.	0.	0.	0.17789999E 03	0.	0.
RIGID COMPONENT MODES,									
COLUMN		1	COLUMN	2	COLUMN	3	COLUMN	4	5
1	0.09999999E 01	-0.37329999E 03	0.	0.	0.	0.	0.	0.	0.
2	0.09999999E 01	-0.24829999E 03	0.	0.	0.	0.	0.	0.	0.
3	0.09999999E 01	-0.12329999E 03	0.	0.	0.	0.	0.	0.	0.
4	0.09999999E 01	0.16999999E 01	0.	0.	0.	0.	0.	0.	0.
5	0.09999999E 01	0.12670000E 03	0.	0.	0.	0.	0.	0.	0.

5 CONTROL POINTS.

RIGID BODY MODAL MATRIX
SURFACE 1, 10 CONTROL POINTS

	COLUMN 1	COLUMN 2	COLUMN 2
1	0.09999999E 01	0.20249999E 02	02
2	0.09999999E 01	-0.80999999E 02	02
3	0.09999999E 01	0.17850000E 02	02
4	0.09999999E 01	-0.71399999E 02	02
5	0.09999999E 01	0.15799999E 02	02
6	0.09999999E 01	-0.63199999E 02	02
7	0.09999999E 01	0.13299999E 02	02
8	0.09999999E 01	-0.53200000E 02	02
9	0.09999999E 01	0.11049999E 02	02
10	0.09999999E 01	-0.44199999E 02	02

FLEXIBLE MODAL MATRIX
SURFACE 1, 10 CONTROL POINTS

4 FREQUENCIES

	COLUMN 1	COLUMN 2	COLUMN 3	COLUMN 4	COLUMN
1	-0.12972128E-00	-0.91809043E-01	-0.35610154E-01	0.30597945E-00	
2	-0.13133238E-00	0.56810184E 00	-0.47615869E-01	0.28780279E-00	
3	-0.17536534E-01	-0.22162712E-00	-0.16294216E-00	0.44284272E-00	
4	-0.15998174E-01	0.09999999E 01	-0.18977508E-00	0.50378348E 00	
5	0.24780568E-00	-0.21537105E-00	-0.44938723E-00	-0.56146173E 00	
6	0.25089253E-00	0.90554103E 00	-0.47773996E-00	-0.44347260E-00	
7	0.62266540E 00	-0.20513079E-00	0.10694845E-00	-0.22509799E-00	
8	0.62794039E 00	0.77614732E 00	0.97215801E-01	-0.62188134E-02	
9	0.99374234E 00	-0.18689994E-00	0.98440876E 00	0.74102239E 00	
10	0.09999999E 01	0.63668663E 00	0.09999999E 01	0.09999999E 01	

	COLUMN 1	COLUMN 2	COLUMN 3	COLUMN 4	COLUMN
1	0.28519310E 04	-0.19187927E-02	-0.44631958E-03	-0.54168701E-03	
2	-0.19111633E-02	0.18691394E 05	0.18501282E-02	0.18329620E-02	
3	-0.44631958E-03	0.18501282E-02	0.23241071E 04	0.85449219E-03	
4	-0.53405762E-03	0.18367767E-02	0.86212158E-03	0.54105940E 04	

WEIGHTING MATRIX
SURFACE 1, NO WEIGHTING MATRIX

		RIGID COMPONENT AERO MATRIX,						5 CONTROL POINTS	
		COLUMN 1	COLUMN 2	COLUMN 3	COLUMN 4	COLUMN 5	COLUMN 6	COLUMN 4	COLUMN 5
1		-0.19883168E 01	-0.71028437E 01	-0.31782072E 01	0.99733821E 01	0.17128096E 01	-0.24681993E 01		
2		0.20347044E 01	-0.30576088E 01	-0.89710928E 01	0.67589525E 00	0.25888789E 01	0.32829072E 01		
3		0.	0.	0.25888789E 01	-0.32829072E 01	-0.89710928E 01	-0.67589525E 00		
4		0.	0.	0.	0.	0.18608618E 01	-0.23997266E 01		
5		0.	0.	0.	0.	0.19459838E-00	0.73566824E 00		
1		0.	0.	COLUMN 9	COLUMN 10	COLUMN			
2		0.	0.	0.	0.	0.			
3		0.20347044E 01	0.30576088E 01	0.	0.	0.			
4		-0.59000403E 01	-0.46448669E-00	0.10709520E 01	0.17538263E 01	0.			
5		0.45456574E-00	-0.31600294E 01	-0.19732837E 01	0.81508698E 00	0.			

AERODYNAMIC MATRIX
SURFACE 1, 10 CONTROL POINTS
1./K R = 0.16670000E 02

	COLUMN 1	COLUMN 2	COLUMN 3	COLUMN 4	COLUMN 5	COLUMN 6
1	0.30099700E 03	-0.60092673E 02	-0.30579615E 03	0.32278208E 02	0.	0.
2	0.19111816E-00	0.16514119E 02	0.57333544E 00	-0.16514119E 02	0.	0.
3	0.	0.	0.	0.	0.29334980E 03	-0.55152775E 02
4	0.	0.	0.	0.	0.14211353E-00	0.13930788E 02
5	0.	0.	0.	0.	0.	0.
6	0.	0.	0.	0.	0.	0.
7	0.	0.	0.	0.	0.	0.
8	0.	0.	0.	0.	0.	0.
9	0.	0.	0.	0.	0.	0.
10	0.	0.	0.	0.	0.	0.
	COLUMN 7	COLUMN 8	COLUMN 9	COLUMN 10	COLUMN 11	COLUMN 12
1	0.	0.	0.	0.	0.	0.
2	0.	0.	0.	0.	0.	0.
3	-0.29728241E 03	0.31246942E 02	0.	0.	0.	0.
4	0.42634058E-00	-0.13930788E 02	0.	0.	0.	0.
5	0.	0.	0.30483988E 03	-0.53868061E 02	-0.30827594E 03	0.31871554E 02
6	0.	0.	0.11384775E-00	0.12607991E 02	0.34154329E-00	-0.12607991E 02
7	0.	0.	0.	0.	0.	0.
8	0.	0.	0.	0.	0.	0.
9	0.	0.	0.	0.	0.	0.
10	0.	0.	0.	0.	0.	0.
	COLUMN 13	COLUMN 14	COLUMN 15	COLUMN 16	COLUMN 17	COLUMN 18
1	0.	0.	0.	0.	0.	0.
2	0.	0.	0.	0.	0.	0.
3	0.	0.	0.	0.	0.	0.
4	0.	0.	0.	0.	0.	0.
5	0.	0.	0.	0.	0.	0.
6	0.	0.	0.	0.	0.	0.
7	0.32483251E .03	-0.52278329E 02	-0.32767460E 03	0.32541166E 02	0.	0.
8	0.84216231E-01	0.11079562E 02	0.25264870E-00	-0.11079562E 02	0.	0.
9	0.	0.	0.	0.	0.30321474E 03	-0.43827882E 02
10	0.	0.	0.	0.	0.53236897E-01	0.84300261E 01

	COLUMN 19	COLUMN 20	COLUMN
1	0.	0.	
2	0.	0.	
3	0.	0.	
4	0.	0.	
5	0.	0.	
6	0.	0.	
7	0.	0.	
8	0.	0.	
9	-0.30521601E 03	0.28517172E 02	
10	0.15971068E-00	-0.84300261E 01	

OUTPUT DATA

FLUTTER ANALYSIS BY A MODAL METHOD USING AERODYNAMIC INFLUENCE COEFFICIENTS

DENSITY = 0.23779999E-02 REDUCED VELOCITY = 0.16670000E 02

2 RIGID BODY DEGREES OF FREEDOM, 4 INPUT MODES (TOTAL).

MODE	EIGENVALUE	ITERATIONS	S.P.	D.P.	AITKENS S.P.	O.P.
1	0.18916545E-00	-0.13115912E-00	11	0	3	0
2	0.13292938E-00	-0.86249603E-03	8	0	2	0
3	0.47592145E-02	-0.12187260E-01	14	0	4	0
4	0.98496465E-02	-0.14207529E-02	2	0	0	0

EIGENVECTORS

COLUMN	1	COLUMN	2	COLUMN	3	COLUMN	4	COLUMN	5	COLUMN	6
1	0.09999999E 01	0.	0.09999999E 01	-0.	0.44428806E-00	0.50267780E 00					
2	-0.17784069E-01	0.34797354E-02	-0.17137665E-02	-0.48118412E-01	0.96034528E-03	-0.22564518E-02					
3	0.15017853E-01	-0.29673016E-01	-0.13965693E-01	-0.43682259E-01	0.09999999E 01	0.					
4	0.17615897E-02	-0.10034480E-02	-0.19148218E-02	0.23242460E-02	0.14828939E-00	-0.69422592E-01					
	COLUMN 7	COLUMN 8	COLUMN								
1	-0.12961736E-00	-0.28763704E-00									
2	0.18731757E-01	0.18930750E-01									
3	0.09999999E 01	0.									
4	-0.17362598E-00	-0.19799793E-00									

CHECK EIGENVALUES AND EIGENVECTORS

0.18916556E-00 -0.13115912E-00 0.13292948E-00 -0.86250604E-03 0.47592180E-02 -0.12187257E-01
0.98496455E-02 -0.14207555E-02

	COLUMN 1	COLUMN 2	COLUMN 3	COLUMN 4	COLUMN 5	COLUMN 6
1	0.09999999E 01	0.	0.09999999E 01	0.	0.44428635E-00	0.50268287E 00
2	-0.17784079E-01	0.34797155E-02	-0.17138086E-02	-0.48118390E-01	0.96057817E-03	-0.22566880E-02
3	0.15017867E-01	-0.29672489E-01	-0.13965622E-01	-0.43682230E-01	0.09999999E 01	0.
4	0.17615926E-02	-0.10034367E-02	-0.19148137E-02	0.23242421E-02	0.14828938E-00	-0.69422607E-01
	COLUMN 7	COLUMN 8	COLUMN			
1	-0.12961851E-00	-0.28764046E-00				
2	0.18731755E-01	0.18930972E-01				
3	0.09999999E 01	0.				
4	-0.17362595E-00	-0.19799802E-00				

MODE OMEGA (CPS) DAMPING VELOCITY (KNOTS)

1	0.22992123E 01	-0.69333566E 00	0.77968240E 03
2	0.27427707E 01	-0.64883775E-02	0.93009681E 03
3	0.14495472E 02	-0.25607714E 01	0.49155374E 04
4	0.10076035E 02	-0.14424405E-00	0.34168689E 04

FINAL FLUTTER MODES

	SURFACE 1, 10		CONTROL POINTS			
	COLUMN 1	COLUMN 2	COLUMN 3	COLUMN 4	COLUMN 5	COLUMN 6
1	-0.19798382E-00	0.79322939E-01	-0.93759178E-01	-0.98772941E-01	-0.38185357E-01	-0.39426726E-01
2	-0.21410581E-00	0.84471434E-01	-0.90696646E-01	-0.13123662E-00	-0.46656333E-01	-0.37693791E-01
3	-0.79324013E-01	0.78083789E-01	0.15005647E-01	-0.68983589E-01	-0.64005999E-01	-0.28086398E-02
4	-0.10239267E-00	0.85738368E-01	0.24328708E-01	-0.12634239E-00	-0.74012243E-01	-0.39014272E-02
5	0.19340336E-00	0.77492066E-01	0.27696662E-00	-0.16448494E-01	-0.21635852E-00	0.16610579E-00
6	0.17411190E-00	0.84444531E-01	0.28701860E-00	-0.68477032E-01	-0.22248337E-00	0.16209281E-00
7	0.59585255E 00	0.44114260E-01	0.63675676E 00	0.18233940E-01	0.25981362E-00	0.13876332E-00
8	0.58214916E 00	0.49438009E-01	0.64768282E 00	-0.27302944E-01	0.27279337E-00	0.12600318E-00
9	0.09999999E 01	-0.	0.98938049E 00	-0.38838806E-01	0.96899516E 00	0.18922722E-01
10	0.99091579E 00	0.35426569E-02	0.09999999E 01	0.	0.09999999E 01	-0.
COLUMN 7						
1	-0.42771401E-01	-0.12797226E-00	COLUMN 8			
2	-0.50017554E-01	-0.11050270E-00				
3	-0.15618632E-00	-0.30295999E-00				
4	-0.17657918E-00	-0.30384143E-00				
5	-0.37818480E-00	-0.29088473E-00				
6	-0.40172405E-00	-0.31386279E-00				
7	0.16239082E-00	-0.13932644E-00				
8	0.15330823E-00	-0.18515402E-00				
9	0.98711184E 00	0.45538359E-01				
10	0.09999999E 01	0.				

FOR SIX FREE-FREE INPUT MODES: REDUCED VELOCITY = 16.67

MODE	OMEGA (CPS)	DAMPING	VELOCITY (KNOTS)
1	0.20944695E 01	-0.68661743E 00	0.71025240E 03
2	0.27474146E 01	0.47400708E-02	0.93167160E 03
3	0.48123113E 01	0.26238423E-00	0.16318955E 04
4	0.11123481E 02	-0.69183040E 00	0.37720668E 04
5	0.12642173E 02	-0.22247387E-00	0.42870682E 04

FINAL FLUTTER MODES

COLUMN		SURFACE 2		SURFACE 1,		10 CONTROL POINTS		COLUMN		COLUMN		COLUMN	
1		2		1,		3		4		5		6	
1	-0.22536416E-00	0.92623165E-01	-0.10428888E-00	-0.95617969E-01	-0.10694816E-00	-0.13273688E-00	-0.64947930E-02						
2	-0.24158062E-00	0.76910723E-01	-0.10536446E-00	-0.12621076E-00	-0.13273688E-00	-0.52423403E-02							
3	-0.10638769E-00	0.90016660E-01	0.77892566E-02	-0.66937186E-01	0.60163083E-01	-0.36864151E-01							
4	-0.12939467E-00	0.96067116E-01	0.99430882E-02	-0.12104959E-00	0.21114814E-01	-0.37425315E-01							
5	0.16926584E-00	0.83706391E-01	0.27414474E-00	-0.16616217E-01	0.42480908E-00	-0.10347813E-00							
6	0.14343195E-00	0.94309451E-01	0.28028685E-00	-0.66988336E-01	0.42607864E-00	-0.12961110E-00							
7	0.57890615E 00	0.49899049E-01	0.63765150E 00	0.19684771E-01	0.74169123E 00	-0.51988563E-01							
8	0.54990779E 00	0.66068787E-01	0.64900336E 00	-0.26153403E-01	0.79264993E 00	-0.10870654E-00							
9	0.09999999E 01	-0.	0.98849916E 00	0.39769159E-01	0.95074619E 00	0.52796511E-01							
10	0.97654707E 00	0.13997710E-01	0.09999999E 01	0.	0.09999999E 01	-0.							
COLUMN 7		COLUMN 8		COLUMN 9		COLUMN 10		COLUMN		COLUMN		COLUMN	
1	-0.18877905E-02	-0.96578937E-01	0.55185816E 00	0.72051873E-01									
2	-0.74029310E-02	-0.87815752E-01	0.53127283E 00	0.59908347E-01									
3	-0.95516861E-01	-0.14733771E-00	0.91057707E 00	0.15560011E-01									
4	-0.10089871E-00	-0.15260053E-00	0.09999999E 01	-0.									
5	-0.47069521E-00	-0.22004030E-01	-0.13254518E-00	-0.34762463E-00									
6	-0.48633242E-00	-0.31481614E-01	-0.88586425E-01	-0.35831629E-00									
7	0.10496839E-00	0.10617295E-00	-0.38397461E-00	0.14641586E-00									
8	0.99158280E-01	0.95083151E-01	-0.38881926E-00	0.16508962E-00									
9	0.98502792E 00	0.29177163E-01	-0.23295638E-00	0.77594201E 00									
10	0.09999999E 01	0.	-0.21485166E-00	0.80241252E 00									

1	0.20907349E 01	-0.69434052E 00	0.70898596E 03
2	0.27376843E 01	-0.94513867E-02	0.92837197E 03
3	0.46601459E 01	0.26838358E-00	0.15802950E 04
4	0.79644649E 01	-0.79144046E 00	0.27008176E 04
5	0.92728179E 01	-0.22534730E-00	0.31444912E 04

SURFACE 1, 10 CONTROL POINTS

-72-

FOR TEN FREE-FREE INPUT MODES: REDUCED VELOCITY = 16.67

MODE OMEGA (CPS) DAMPING VELOCITY (KNOTS)

1	0.20647985E 01	-0.69443611E 00	0.70019069E 03
2	0.27409285E 01	-0.12062252E-01	0.92947210E 03
3	0.44713092E 01	0.27819197E-00	0.15162589E 04
4	0.6985186E 01	-0.61700870E 00	0.23732570E 04
5	0.95550377E 01	-0.26846989E-00	0.32401942E 04

FINAL FLUTTER MODES

	SURFACE 1,		10 CONTROL POINTS		COLUMN 5	COLUMN 6
	COLUMN 1	COLUMN 2	COLUMN 3	COLUMN 4		
1	-0.22672210E-00	0.92801184E-01	-0.10980548E-00	-0.10249859E-00	-0.11565738E-00	0.10802589E-02
2	-0.24322628E-00	0.96231290E-01	-0.11252954E-00	-0.13723224E-00	-0.14201599E-00	0.41718773E-03
3	-0.10896556E-00	0.90128426E-01	0.36838976E-02	-0.72438279E-01	0.39346816E-01	-0.37219467E-01
4	-0.13140251E-00	0.95914066E-01	0.38689399E-02	-0.12866554E-00	0.62714847E-02	-0.41508120E-01
5	0.16390713E-00	0.84669287E-01	0.27385023E-00	-0.15871787E-01	0.38962222E-00	-0.14034727E-00
6	0.13893154E-00	0.93805225E-01	0.27711201E-00	-0.70519144E-01	0.38136957E-00	-0.15814554E-00
7	0.57352652E 00	0.51462667E-01	0.63861449E 00	0.20940991E-01	0.72148240E 00	-0.93108623E-01
8	0.54571509E 00	0.66242675E-01	0.64748625E 00	-0.26459806E-01	0.75146637E 00	-0.13747277E-00
9	0.09999999E 01	-0.	0.98693993E 00	0.36697865E-01	0.94384047E 00	0.67121518E-01
10	0.97106511E 00	0.19206408E-01	0.09999999E 01	0.	0.09999999E 01	-0.
COLUMN 7						
1	0.55525421E-01	0.28637560E-01	0.10480621E-00	-0.75338227E-01	COLUMN 10	
2	0.37312139E-01	-0.33451135E-02	0.67769676E 00	-0.13640756E-00		
3	-0.61299051E-01	-0.81927439E-01	0.32303939E-00	-0.65244763E-01		
4	-0.72267612E-01	-0.58646142E-01	0.20867298E-00	-0.67849804E-01		
5	-0.53633708E 00	-0.13744275E-00	0.09999999E 01	-0.		
6	-0.51012741E 00	-0.16304075E-00	0.93896465E 00	0.10472411E-00		
7	-0.12073137E-00	-0.69966570E-01	-0.37512115E-00	0.53746350E 00		
8	-0.82761717E-01	-0.11890361E-00	-0.21398164E-00	0.38648304E-00		
9	0.09999999E 01	0.	-0.69761330E 00	-0.62528355E 00		
10	0.94308997E 00	-0.89770386E-02	-0.77934656E 00	-0.50266519E 00		

B. Punched Output

1. A deck of punched cards is output from the vibration analysis portion of this program that is suitable as input to other programs requiring free-free vibration modes and frequencies. The flutter modes and frequencies also may be punched in cards if it is so desired. If a printout of the dynamic matrix is needed (NPUNCH = -1) but no punched-card output is wanted, the machine operator must be instructed to ignore the output punch tape.

2. All punched output is sequenced in order on Column 73 through 80 starting with HM140000. Each mode will have NSUR (number of surfaces) partitions, and the output data will be in the following order.

a. Control card for surface 1 (FORMAT 18I4):

Field 1 contains $NSIZE_1$, number of control points on component 1.

Field 2 contains NMODES, number of output modes punched.

b. Frequencies (column binary format): The frequencies are punched in the order in which they appear in the frequency column in the printed output. The column binary card(s) (only one card if $NMODES \leq 22$) containing the frequencies are followed by a TRA card.

c. Modes for component 1 (column binary format): The component 1 partition of each vibration mode is punched as a column with a TRA card following the output for each column. If the final flutter modes are punched in cards, a TRA card will follow both the output for the real column and the output for the imaginary column forming each mode.

d. Control cards and modes for component i , $i = 2$, NSUR: Parts a and c (above) are repeated for NSUR surfaces when $NSUR > 1$.

SECTION V
PROCESSING INFORMATION

A. Operation

FORTRAN II MONITOR system (uses chain job capability).

B. Estimated Machine Time

Because of the many variables involved such as number of input modes, number of surfaces, number of iterations required for eigenvector convergence, etc., it is not possible to give a practical method of estimating machine time. Complete but independent decks may be stacked to save program read-in time. The two data decks for the example problems were stacked and the machine time was 1.84 minutes.

C. Machine Components Used

The following machine logical tape units were used; each is set equal to the symbolic tape unique to the program.

TAPE 2 (input tape) = NTAPE2

TAPE 3 (output print tape) = NTAPE3

TAPE 11 (chain tape) = NTAPE1

TAPE 12 (output punch tape) = NTAPE0

TAPES 13, 5, 6, 7, 8, and 14 (utility tapes) = NTAPES4, 5, 6, 7, 8, and 9, respectively.

NTAPE1 contains two chains. Chain 1 uses six utility tapes for storing matrices; the number of rows and number of columns precedes each matrix. The symbolic utility tapes and the matrices stored on each are listed below (i = 1, NSUR on all tapes except NTAPE 9).

NTAPE4: 1) $[M]_i$
 If NRIGID > 0
 2) $[h_R]$ for the system

If NRIGID > 0 and NFREE > 0

3) $[a_{RF}]$

NTAPE5: 1) $[h_F]_i$

NTAPE6: If NRIGID > 0 and NFUS > 0

1) $[h_{Ro}]$

If any $ISW_i > 0$

2) $[W]_i$

NTAPE7: 1) $[W]_i[C_h]_i$ or $[C_h]_i$

NTAPE8: If NRIGID > 0

1) $[\bar{M}]_i = ([M]_i + 32.174\rho_j b_{ri}^2 s_i [W]_i [C_h]_i)$
 $j = 1, NDENS \text{ or } 1, NRHO$

2) $[\bar{M}]_i [h_F]_i$

3) $[h_F]_i^T [\bar{M}]_i$

NTAPE9: 1) $[\bar{m}_{FF}] = [h_F]_i^T [\bar{M}]_i [h_F]_i$, $i = 1$ for $[h_F]_i$
 if any free-free modes are input;

If NRIGID > 0

2) $[\bar{m}_{RF}]_i = [h_R]_i^T [\bar{M}]_i [h_F]_i$

3) $[\bar{m}_{FR}]_i = [h_F]_i^T [\bar{M}]_i [h_R]_i$

If NRIGID > 0 and NFREE > 0

4) $[\Delta\bar{m}][a_{RF}]$

5) $[a_{RF}]^T [\Delta\bar{m}]$

6) $[a_{RF}]^T [\Delta\bar{m}][a_{RF}]$

If NRIGID > 0

7) $[\bar{m}_{RR}]^{-1} = ([h_R]^T [\bar{M}][h_R] + [\Delta\bar{m}])^{-1}$

8) $[h_R][\bar{m}_{RR}]^{-1}$

Chain2 also uses NTAPE8 for storing computations used in the MITERS subroutine and for storing the normalized vibration or flutter modes for each surface.

SECTION VI

PROGRAM NOTES

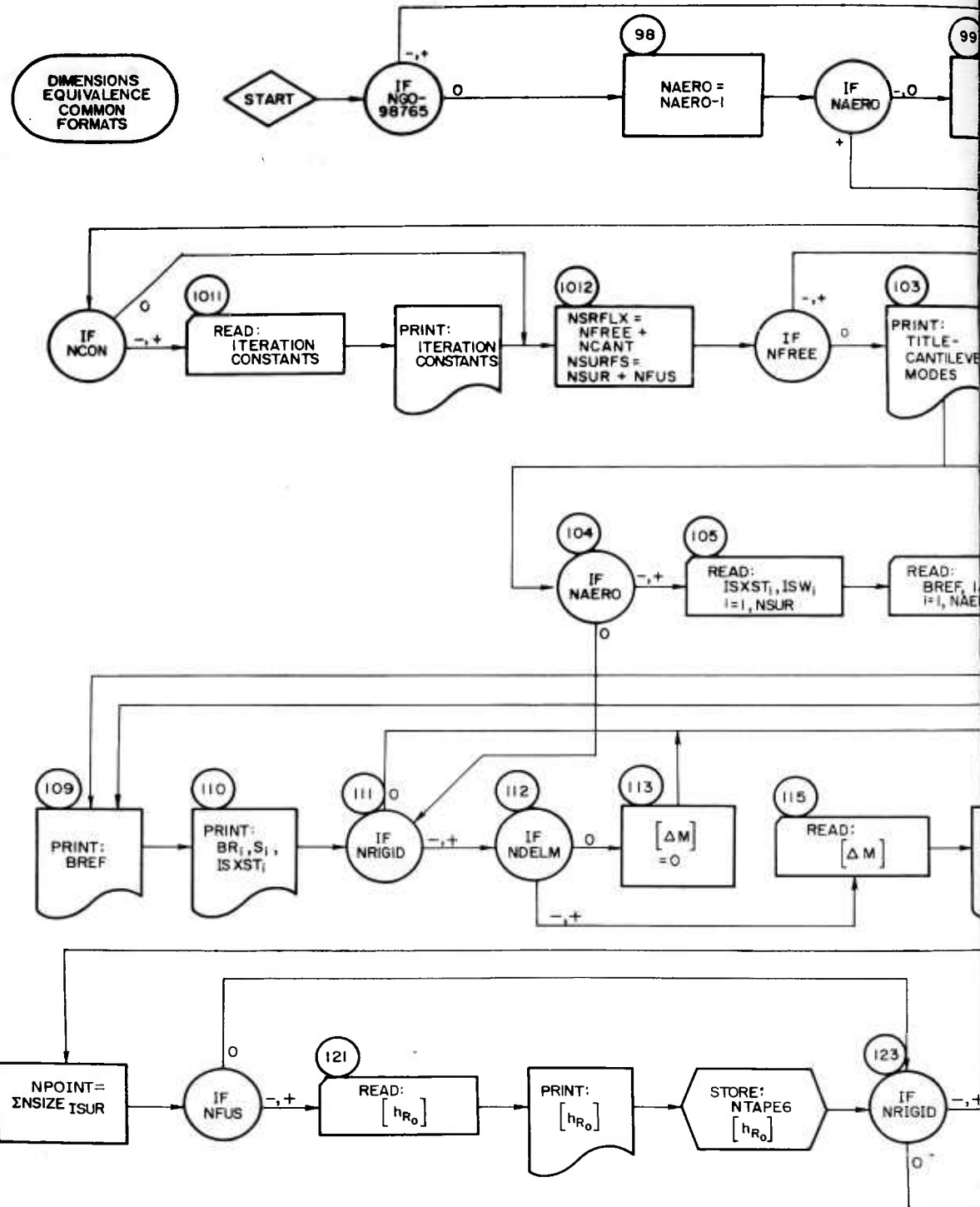
A. Subroutine Used

In addition to the main program (page 91), the following subroutines were used.

1. RDLN, reads and prints the title card (page 127).
2. MMULTD, computes the product $[C]$ of two matrices $[A]$ and $[B]$ (page 129).
3. MPRINT, prints matrix in matrix format (page 132).
4. MREAD, reads a matrix in either column binary or FORTRAN format (page 135).
5. BINRD, reads column binary cards from tape (page 140).
6. MNVRSX, computes the inverse of a complex matrix (page 150).
7. INVERS, computes the inverse of a real matrix (page 153).
8. BINPU, writes column binary cards on tape (page 158).
9. MPUNCH, punches column binary cards (page 168).
10. MITERS, real or complex matrix iteration (page 172).
11. NPNRMX, vector normalization (page 185).
12. SWEEPX, computes the true mode and sweeps it from the related matrix (page 190).
13. CLOSE, matrix iteration using double-precision arithmetic (page 196).
14. DPFORM, converts double-precision numbers to single-precision numbers and conversely (page 207).
15. DPMLTX, matrix multiplication using single- or double-precision numbers (page 211).

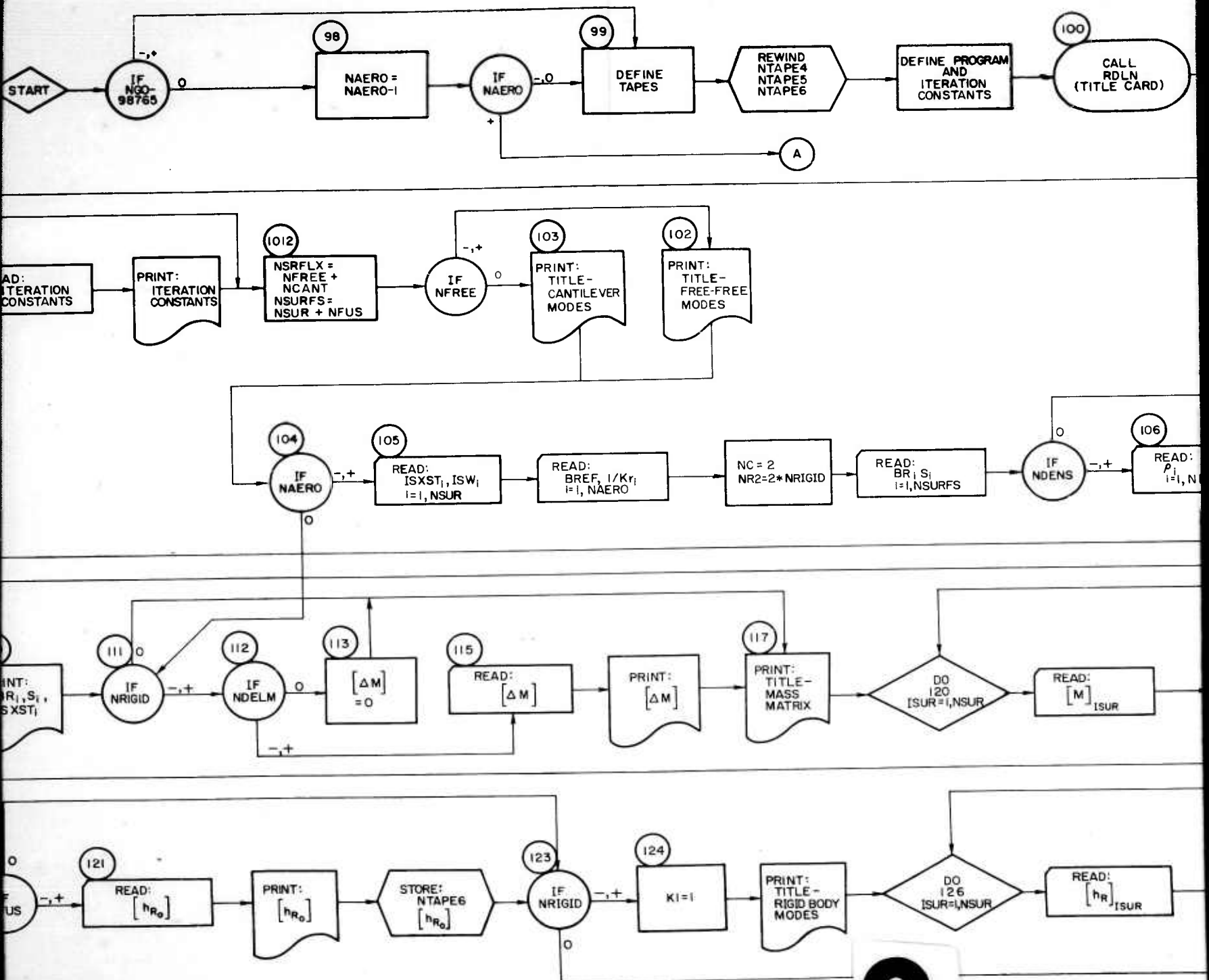
B. Generalized Tapes

The Aerospace IBM 7090 computer logical tape units used in the program are TAPES 2, 3, 5, 6, 7, 8, 11, 12, 13 and 14, but these may be altered by placing the desired machine logical tape units on symbolic cards HM140716 through HM140725. These symbolic cards appear in the main program, and the logical tape units are defined in the comments preceding the cards.



1

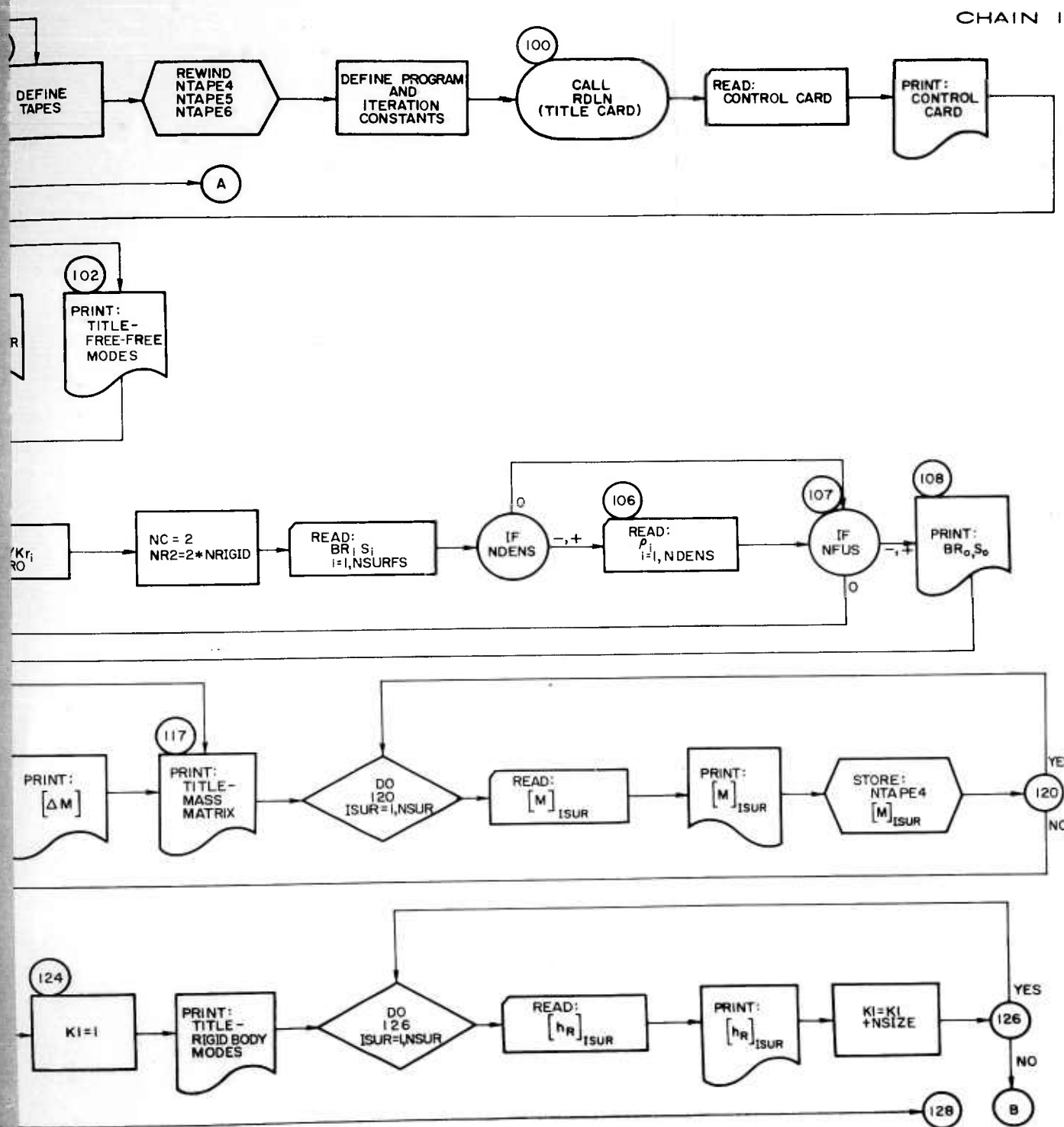
SECTION VII
FLOW DIAGRAM -- MAIN PROGRAM



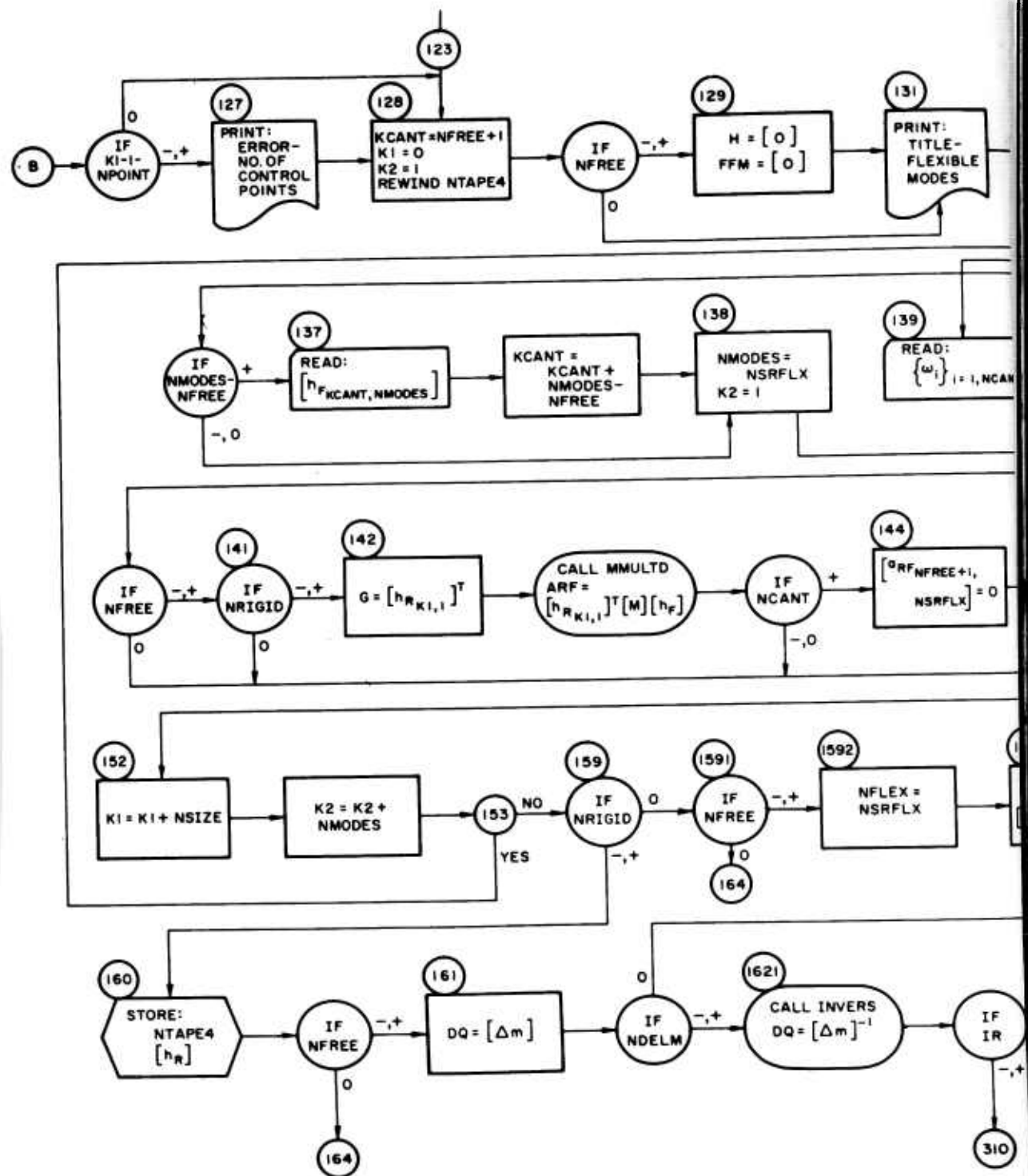
2

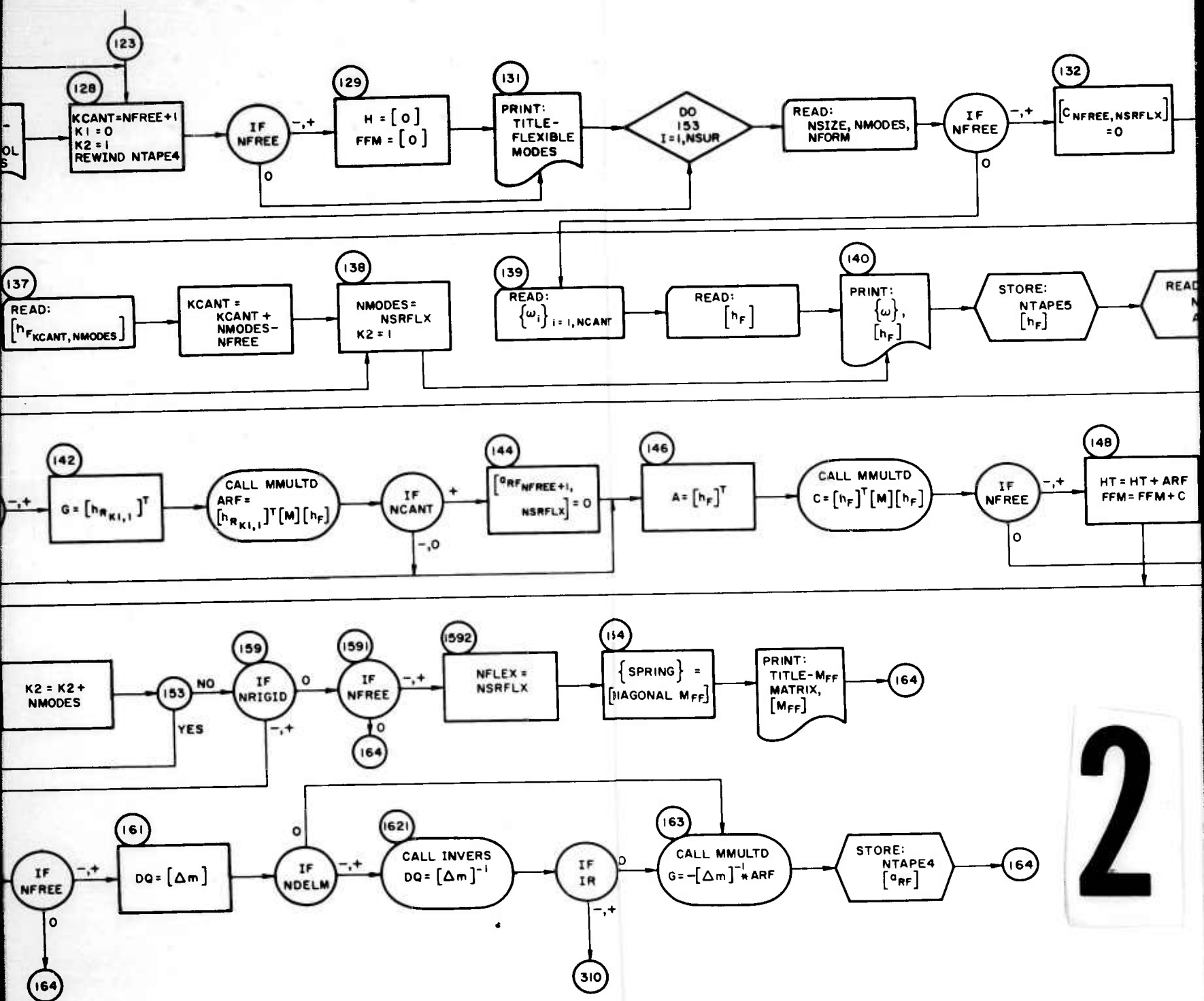
SECTION VII

PROGRAM -- MAIN PROGRAM



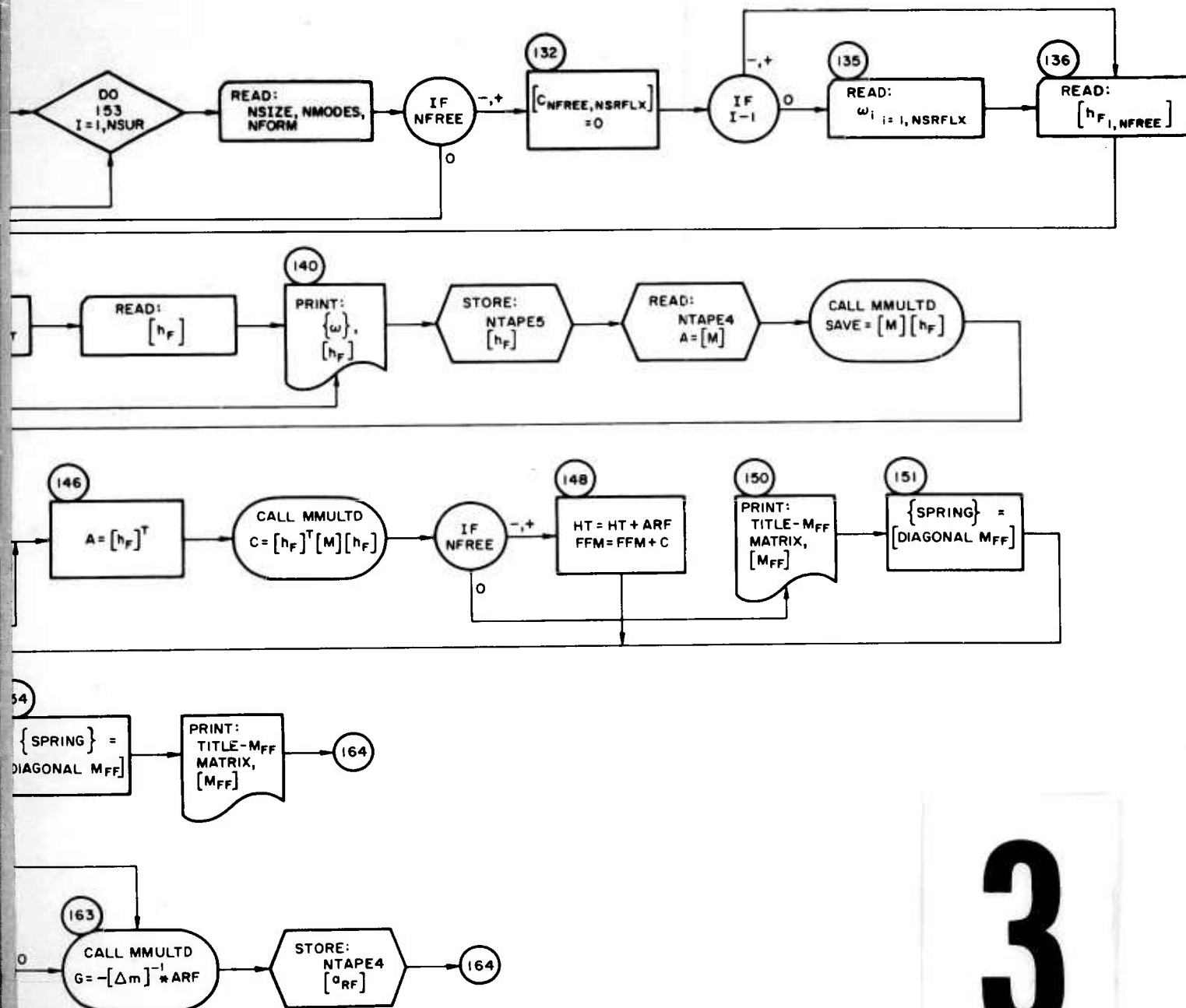
3





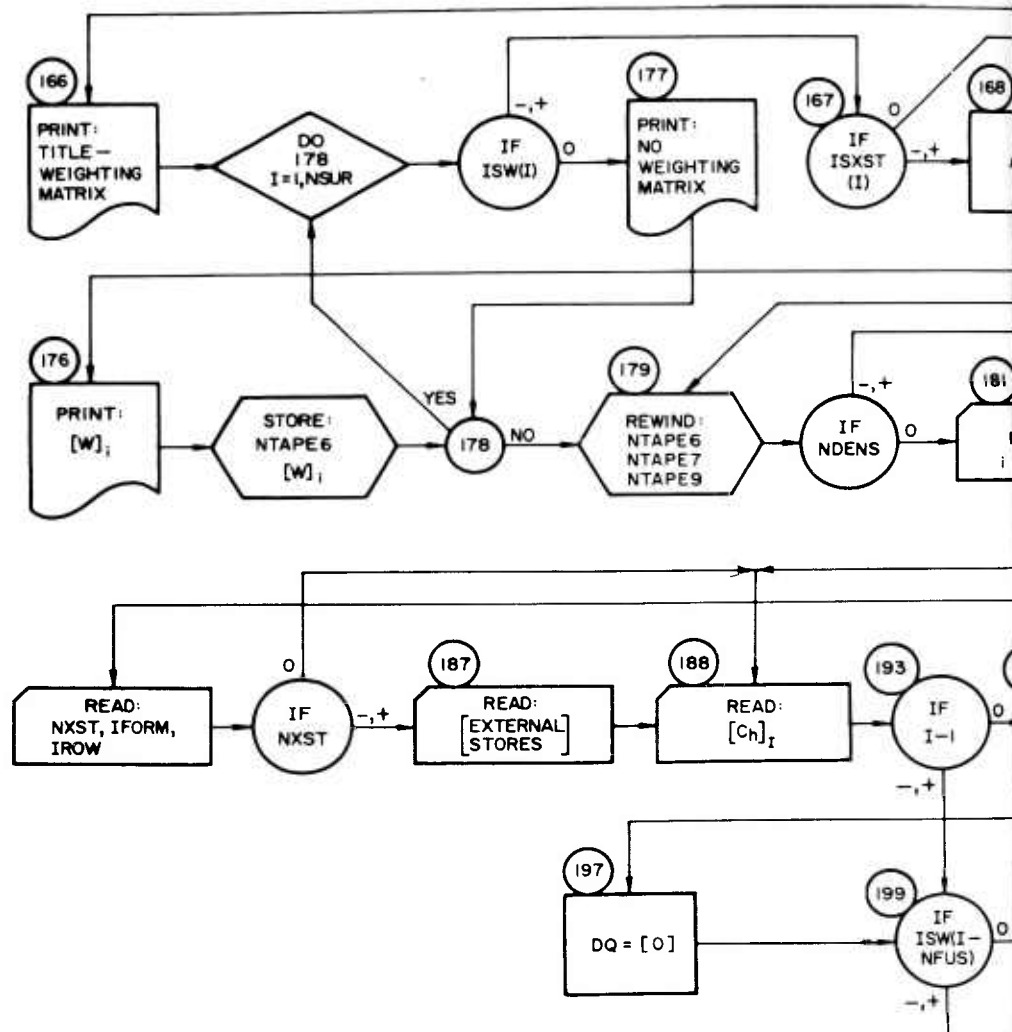
2

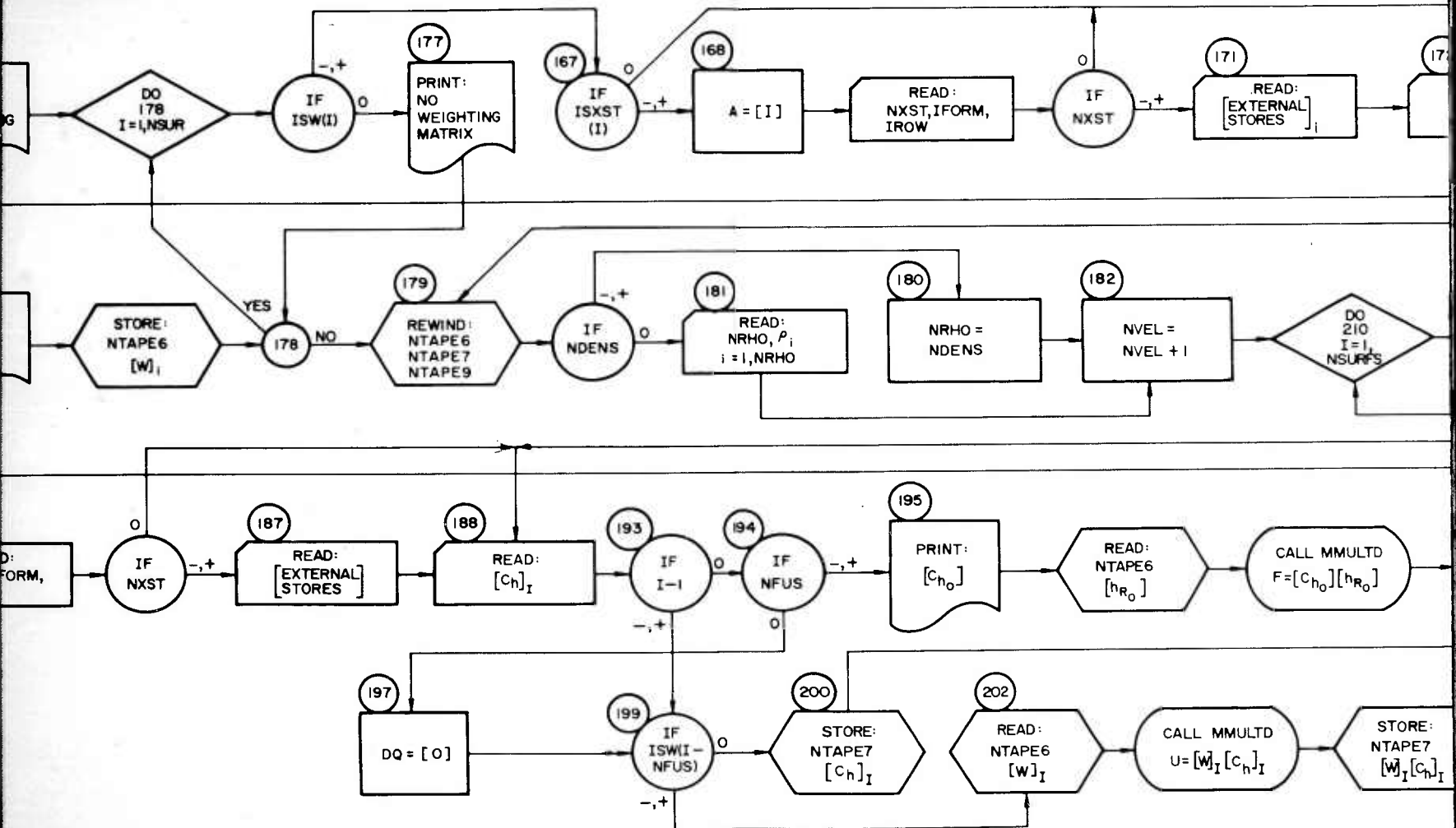
CHAIN 1 (CONT.)



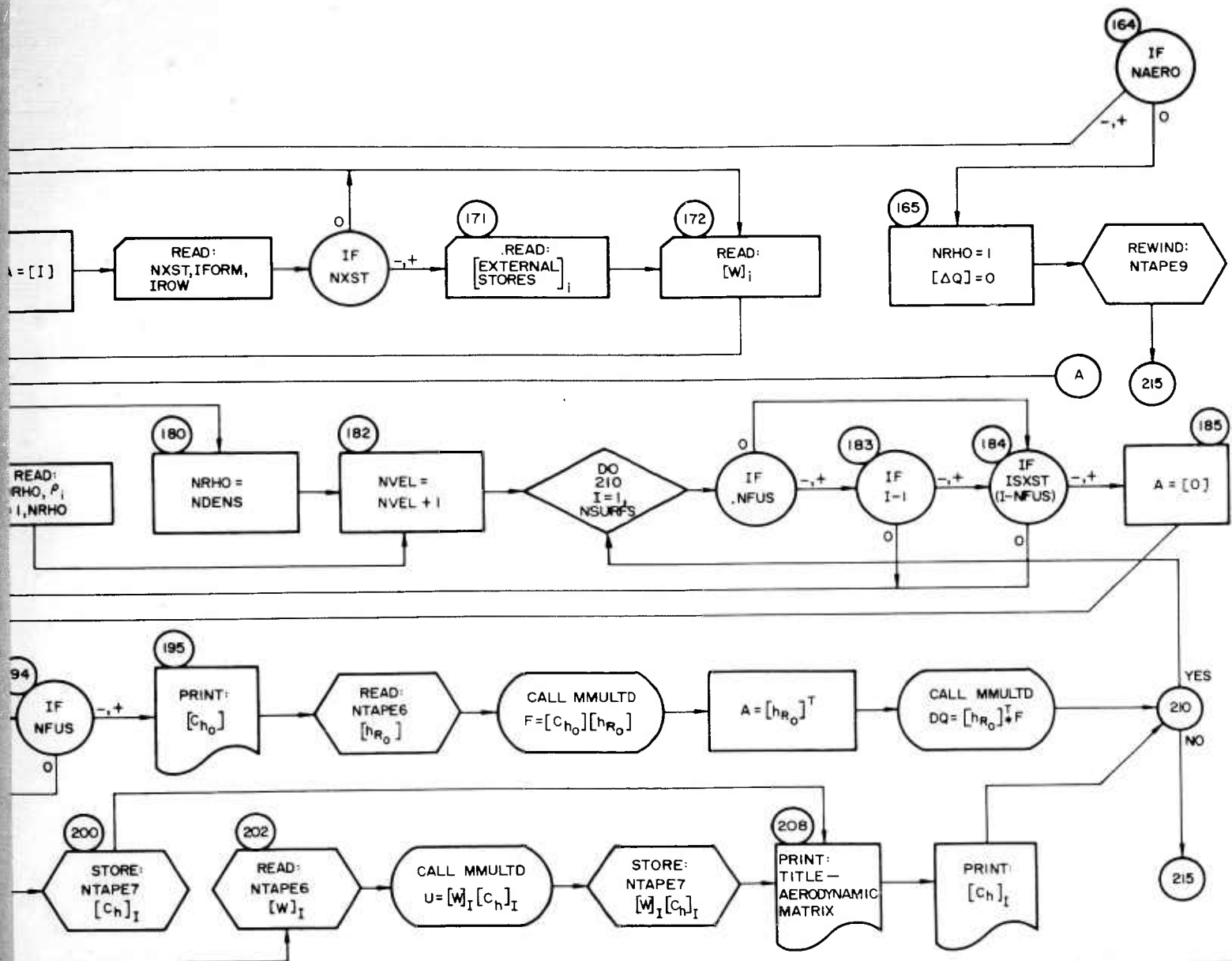
3

1

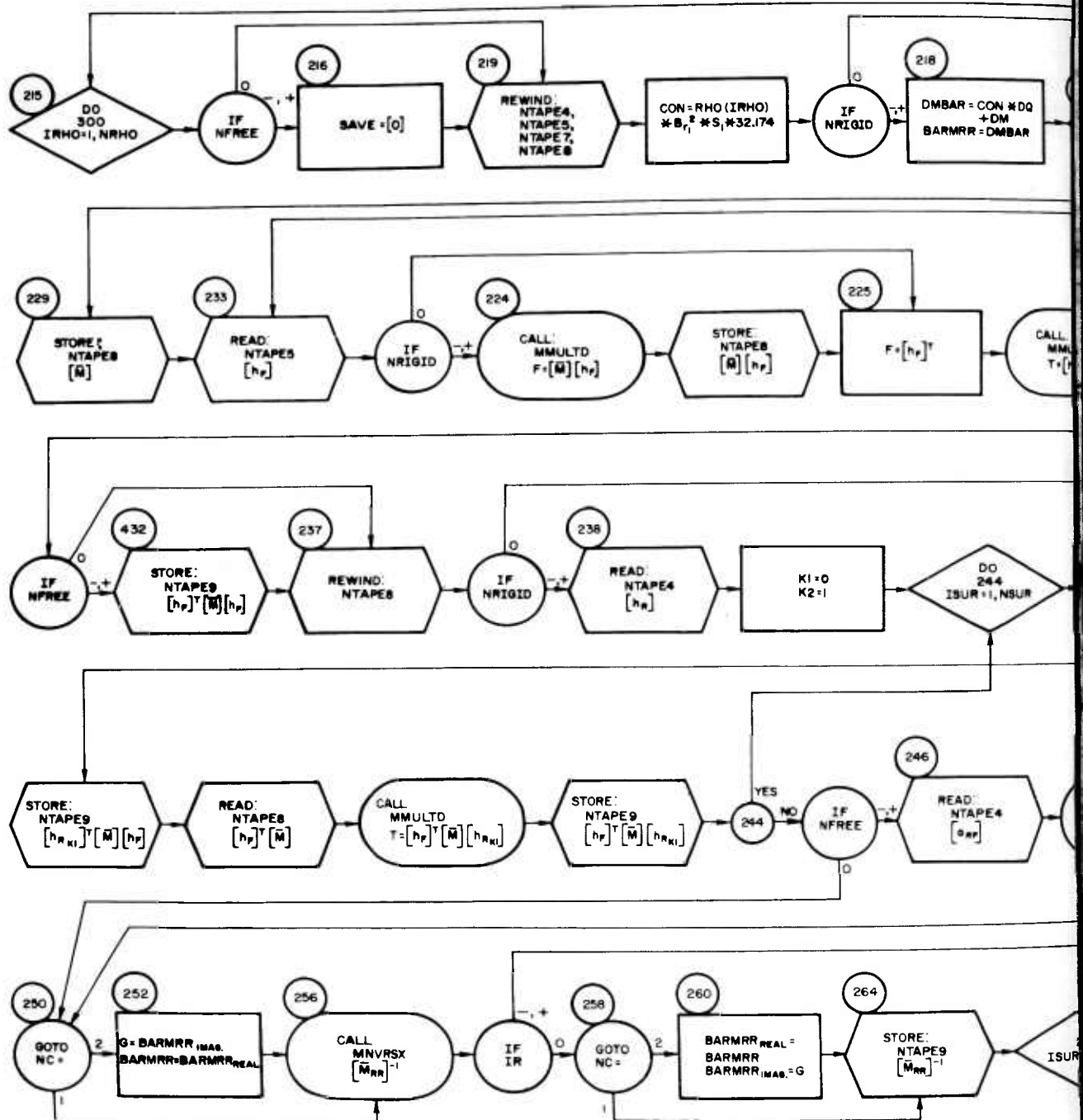


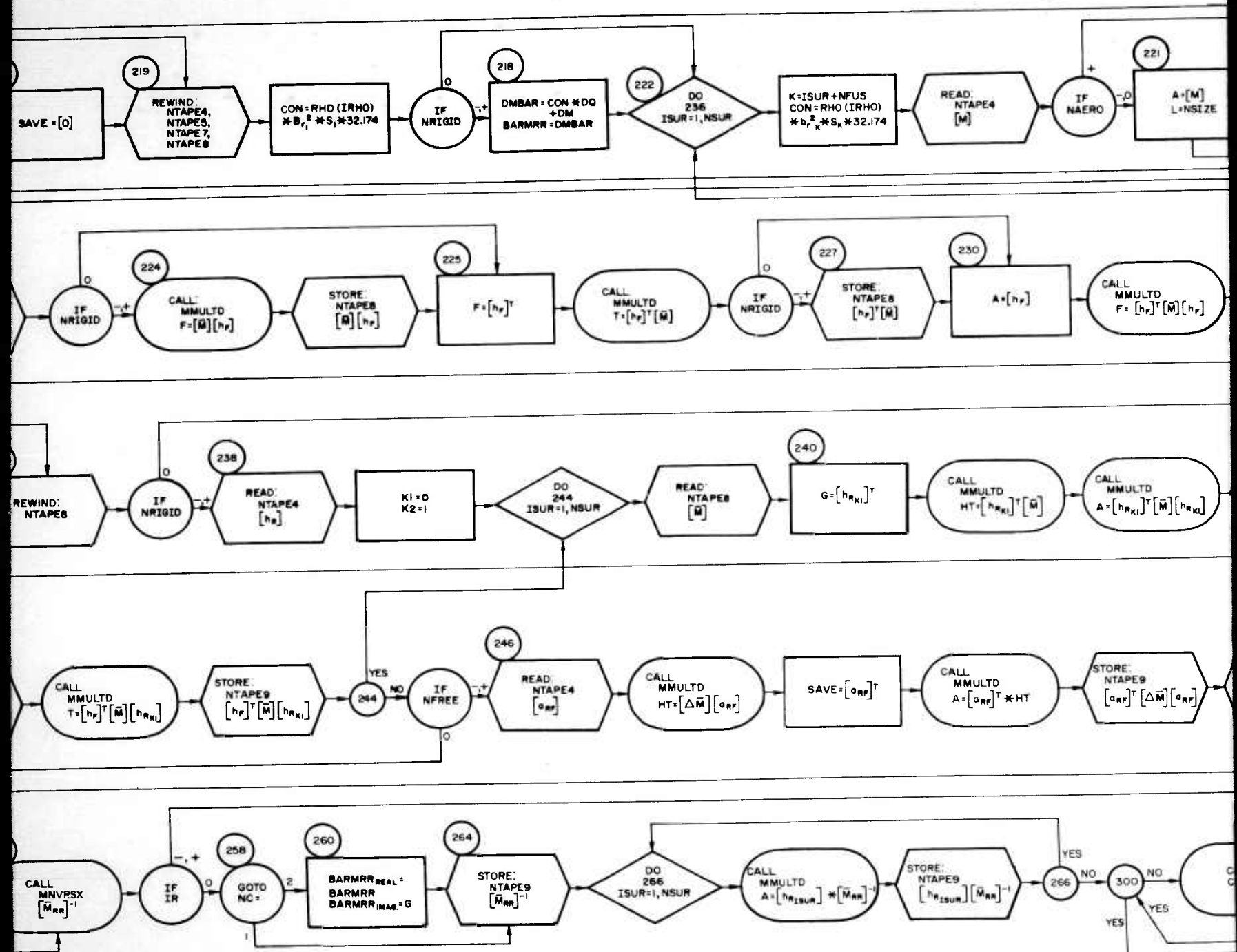


CHAIN 1 (CONT.)



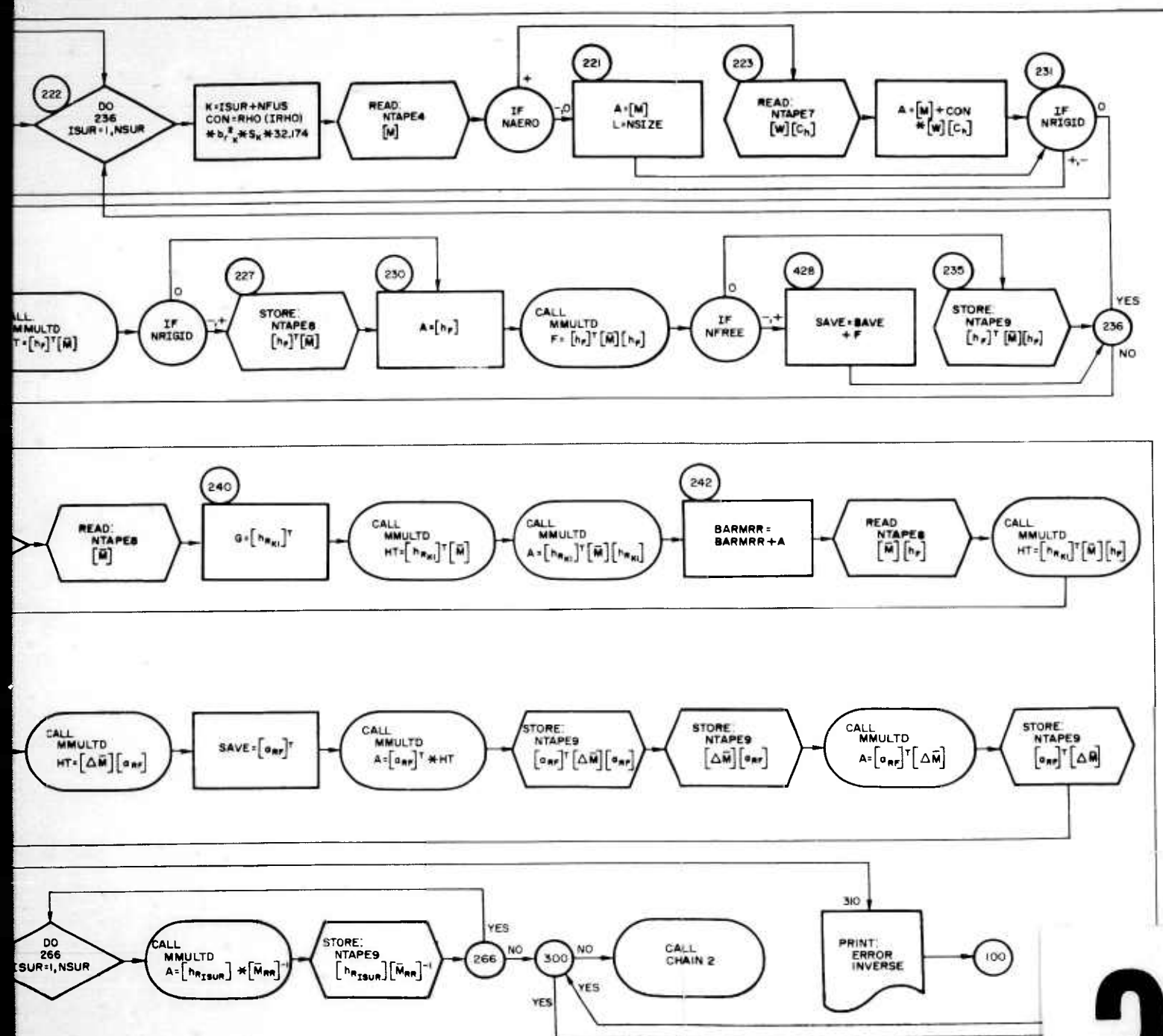
1

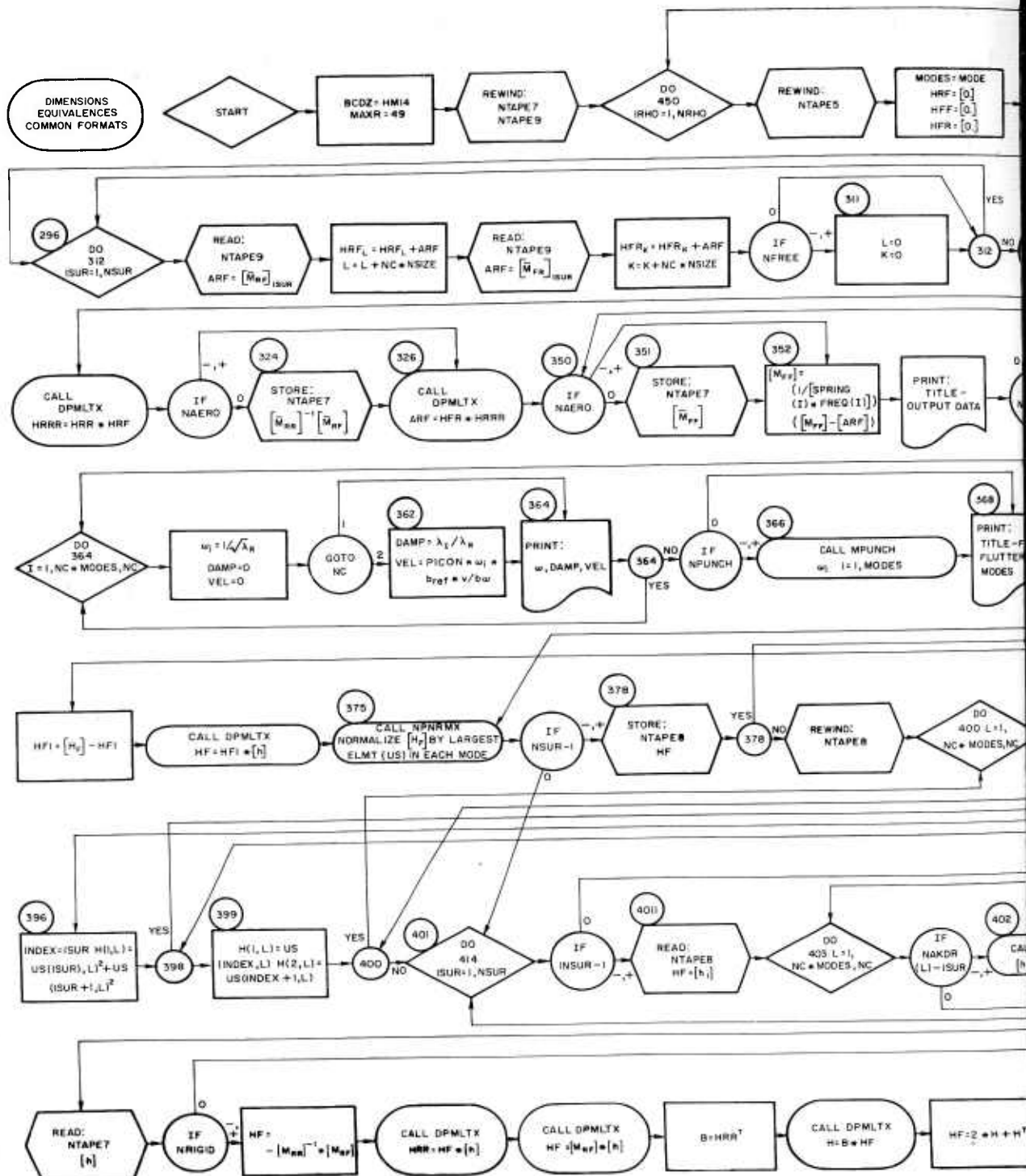




2

CHAIN (1, 11)





1





SECTION VIII

SYMBOLIC LISTING

Some of the program FORTRAN symbols which were not defined in Section III, Part B are defined below.

<u>FORTRAN Symbols</u>	<u>Definitions</u>
BARMRR(I, J)	$[\bar{m}_{RR}]$ and $[\bar{m}_{RR}]^{-1}$
BOOLT	BCD titles
BREF	b_r (reference)
BR(I)	b_{ri} series
DM(I, J)	$[\Delta m]$
DMBAR(I, J)	$[\Delta \bar{m}]$
DQ(I, J)	$[h_{Ro}]^T [C_{ho}] [h_{Ro}]$
MAXS, MAXR, MAXP, MAXQ	Dimensional number of rows in working matrices
NFLEX, NFLEX2	Total number of flexible modes
NGO	Chain 1 entry control number
NPOINT	Total number of control points on all flexible surfaces
NRIGID, NR2	Total number of rigid-body modes
NSIZES(I)	Number of control points on surface i
NSRFLEX	NFREE + NCANT
NSURFS	NSUR + NFUS, (NFUS = 0 or 1)
PICON	$2 \times \pi \times 0.5921$
RHO(I)	ρ_i , $i = 1, NDENS$

SPRING

$$[K/4\pi^2]$$

T, G, A, SAVE, H, ARF,
C, FFM, F, HR, HT

Working areas of core

VELCTY(I)

$$(1/k_r)_i = (V/b_r \omega)_i$$

The complete symbolic listing is given on the following pages.

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```

C      NSUR=TOTAL NUMBER OF SURFACES ALLOWED.
C      NDENS=TOTAL NUMBER OF DENSITIES ALLOWED.
C      NRIGID=TOTAL NUMBER OF RIGID BODIES ALLOWED
C      NSIZE=TOTAL NUMBER CONTROL POINTS ON ANY ONE SURFACE ALLOWED
C      NMDES=TOTAL NUMBER MODES INPUT ON ANY ONE SURFACE.
C      DIMENSION
C      1      ISXST(NSUR), ISW(NSUR), RHD(NDENS), BR(NSUR+1),
C      2      S(NSUR+1), DM(NRIGID), LOW(NSIZE), LHIGH(NSIZE),
C      3      HR(NSUR*NSIZE,NRIGID), FREQ(NSUR*NMDES),
C      4      DM(NRIGID,NRIGID), DMBAR(NRIGID,2*NRIGID), QO(NRIGID,
C      5      2*NRIGID)
C
C      DIMENSION
C      1      ISXST(20), ISW(20), 8R(21), S(21), RHO(20), DM(6,6),
C      2      DQ(6,12), DMBAR(6,12), BARMRR(6,12), FREQ(40),
C      3      LOW(50), LHIGH(50),      NSIZES(20), IT(520),
C      4      A(50,100), C(50,50), HR(1000,6), F(50,100), G(6,50),
C      5      HT(6,100), H(50,12), ARF(6,40), T(50,100)
C      ,SPRING(40),SAVE(50,50),VELCTY(20),FFM(50,50)
C
C      EQUIVALENCE
C      1      (IT(1),ISXST(1)), (IT(21),ISW(1)), (IT(41),NSIZES)
C      2      , (IT(61),NTAPE1), (IT(62),NTAPE2), (IT(63),NTAPE3),
C      3      (IT(64),NTAPE4), (IT(65),NTAPE5), (IT(66),NTAPE6),
C      4      (IT(67),NTAPE7), (IT(68),NTAPE8), (IT(69),NTAPE9),
C      5      (IT(70),NSUR), (IT(71),NRIGID), (IT(72),NFREE),
C      6      (IT(73),NCANT), (IT(74),NFUS), (IT(75),NDENS),
C      7      (IT(76),MDES), (IT(77),NPDINT), (IT(78),NLEX),
C      8      (IT(79),MAXR), (IT(80),MAXQ), (IT(81),MAXS),
C      9      (IT(82),MAXP), (IT(83),NSURES), (IT(84),NR2),
C      10     (IT(85),NFLEX2), (IT(86),NC), (IT(87),NGO)
C      EQUIVALENCE
C      1      (IT(88),NPUNCH), (IT(89),BDOLT), (IT(91),RHO),
C      2      (IT(111),FREQ), (IT(151),BREF), (IT(152),NSRFLX),
C      3      (IT(153),8R), (IT(174),S), (IT(195),DM), (IT(231),
C      4      DQ), (IT(303), DMBAR), (IT(375),BARMRR),
C      5      (IT(447),NAERD), (IT(448),NRHO), (IT(449),PICON),
C      6      (IT(450),SPRING), (IT(490),VELCTY), (IT(511),NVEL),
C      7      (IT(512),NTAPE0), (IT(513),NCARDS), (IT(514),EPSPI),
C      8      (IT(515),EPDP), (IT(516),NITRSP), (IT(517),NITROP),

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```

      8      (IT(1518),AITKEN), (A(2501),C), (HR,F), (ARF,T,LOW),
      9      (T(241),HT,H), (T(1841),G,HIGH)
COMMON      IT, A, HR, T ,SAVE,FFM
1 FORMAT    (1814)
2 FORMAT    (6E12.8)
3 FORMAT    (1H 16X, 41H FLUTTER ANALYSIS BY A MODAL METHOD USING
1          35H AEROODYNAMIC INFLUENCE COEFFICIENTS /// 10H NSUR =
2          112, 10H NAERO = 114, 11H NRIGID = 112, 9H NFUS =
3          112, 10H NOENS = 114, 14H MODES OUT = 112, 7H NFREE
4          3H = 113, 10H NCANT = 113 )
4 FORMAT    (1H0 38X, 11H B (REF) = 1E20.8, / 1H0 20X, 7H SURFACE
1          17X, 2H B 18X, 2H S 10X, 20H EXTERNAL STORES SIZE ///)
5 FORMAT    (1H0 10X, 21H B RIGID COMPONENT = 1E18.8, 5X, 8H S RIGID
1          13H COMPONENT = 1E18.8 )
6 FORMAT    (1H 1125, 2( 5X, 1E20.8), 1112 )
7 FORMAT    (1H0 20X, 23H FREE-FREE INPUT MODES, 116, 11H CANTILEVER
1          16H MODES INCLUDED. )
8 FORMAT    (1H0 28X, 24H CANTILEVER INPUT MODES )
9 FORMAT    (1H0 45X, 116, 13H FREQUENCIES // 11H 45X, 1E 20.8)
10 FORMAT   (1H1 48X, 12H MASS MATRIX )
11 FORMAT   (41H0 NUMBER OF CONTROL POINTS THIS MATRIX, ( 114,
1          48H) AND TOTAL NUMBER OF CONTROL POINTS EXPECTED, ( 114,
2          37H) DO NOT AGREE, PROGRAM CONTINUED.... )
12 FORMAT   (1H1 42X, 24H RIGID BODY MODAL MATRIX )
13 FORMAT   (1H 38X, 8H SURFACE 112, 1H, 116, 15H CONTROL POINTS)
14 FORMAT   (1H1 43X, 23H FLEXIBLE MODAL MATRIX )
15 FORMAT   (1H1 46X, 18H WEIGHTING MATRIX )
16 FORMAT   (1H1 28X, 29H RIGID COMPONENT AERO MATRIX, 119, 8H CONTROL
1          7H POINTS )
17 FORMAT   (1H1 33X, 20H AERODYNAMIC MATRIX 3X, 5X,
1          10H 1./K R = 1E20.8 )
18 FORMAT   (1H0 30X, 23H RIGID COMPONENT MODES, 1110,
1          17H CONTROL POINTS. )
19 FORMAT   (1H0 30X, 29H RIGID COMPONENT MASS MATRIX )
20 FORMAT   (47H1 ERROR IN INVERSE ROUTINE, PROGRAM TERMINATED )
21 FORMAT   ( 1H 38X, 8H SURFACE 112, 1H, 6X, 13H NO WEIGHTING

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1      7H MATRIX )
22 FORMAT (1H0 40X, 21H M FF MATRIX,SURFACE 116 )
23 FORMAT ( 3E12.8, 214 )
24 FORMAT ( 8H0EPSP = 1E16.8, 1X, 8H EPDP = 1E16.8, 1X,
1      10H AITKEN = 1E16.8, 1X, 10H NITRSP = 114, 1X,
2      10H NITRDP = 114 )

C      NTAPE0 = OUTPUT PUNCH TAPE
C      NTAPE1 = CHAIN TAPE
C      NTAPE2 = INPUT TAPE
C      NTAPE3 = OUTPUT PRINT TAPE
C      NTAPE4 = / ARE UTILITY TAPES
C      NTAPE5 = /
C      NTAPE6 = /
C      NTAPE7 = /
C      NTAPE8 = /
C      NTAPE9 = /

IF ( NGO-98765 ) 99,98,99
98 NAERO = NAERO-1
IF ( NAERO ) 99,99,179

99 NTAPE1 = 11
NTAPE2 = 2
NTAPE3 = 3
NTAPE4 = 13
NTAPE5 = 5
NTAPE6 = 6
NTAPE7 = 7
NTAPE8 = 8
NTAPE9 = 14
NTAPE0 = 12
CALL SETH(NTAPE5,NTAPE6,NTAPE7,NTAPE8,NTAPE4,NTAPE9)

MAXR = 50
MAXQ = 6
MAXS = 1000
MAXP = 40

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VIBRATION AND FLUTTER ANALYSIS BY A MODAL METHOD, USING AICS.

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NVEL=0
NCARDS = 0
EPSP = .5E-06
EPDP = .5E-07
AITKEN = .9
NITRSP = 40
NITKDP = 100
PICON=.5921*2.*3.14159
NC=1

REWIND NTAPE4
REWIND NTAPE5
REWIND NTAPE6

C*****
C READ IN TITLE, CONTROLS AND CONSTANTS AND PRINT.

100 CALL RDLN (NTAPE2,NTAPE3,1)
   READ INPUT TAPE NTAPE2, 1,
   ,NFREE, NCANT, NDELM, NPUNCH, NCON
1   WRITE OUTPUT TAPE NTAPE3, 3, NSUR, NAERO, NRIGID, NFUS, NDENS,
   ,MODES, NFREE, NCANT
1   IF ( NCON ) 1011,1012,1011
1011 READ INPUT TAPE NTAPE2, 23, EPSP, EPDP, AITKEN, NITRSP, NITROP
   WRITE OUTPUT TAPE NTAPE3, 24, EPSP, EPDP, AITKEN, NITRSP, NITROP
1012 NSRFLX=NFREE+NCANT
   NSURFS=NSUR+NFUS
   NR2= NRIGID *NC
   IF ( NFREE ) 102,103,102
   GOTO 104
102 WRITE OUTPUT TAPE NTAPE3, 7, NCANT
   GOTO 104
103 WRITE OUTPUT TAPE NTAPE3, 8
104 IF (NAERO) 105,111,105
105 READ INPUT TAPE NTAPE2, 1, (ISKST(1),ISW(1),I=1,NSUR)
   READ INPUT TAPE NTAPE2, 2, BREF, (VELCTY(1),I=1,NAERO)
   NC=2
   NR2= NRIGID *NC
   READ INPUT TAPE NTAPE2, 2, (BR(1),S(1),I=1,NSURFS)

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HML40738
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HML40743
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HML40746
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HML40748
HML40749
HML40750
HML40751
HML40752
HML40753
HML40754
HML40755
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HML40758
HML40759
HML40760
HML40761
HML40762
HML40763
HML40764
HML40765
HML40766
HML40767
HML40768
HML40769

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```

      IF ( NDENS )      106,107,106
106 READ INPUT TAPE NTAPE2, 2, (RHO(I),I=1,NDENS)
      107 IF ( NFUS )      108,109,108
107 WRITE OUTPUT TAPE NTAPE3, 5, BR(1), S(1)
109 WRITE OUTPUT TAPE NTAPE3, 4, 8REF
      DO 110 I=1,NSUR
      J=1+NFUS
110 WRITE OUTPUT TAPE NTAPE3, 6, (I, BR(J), S(J), ISXST(I))

C*****
C READ IN FUSELAGE MASS CHARACTERISTICS
111 IF ( NRIGID )      112,117,112
112 IF ( NOELM )      115,113,115
113 DO 114 I=1,NRIGID
      DO 114 J=1,NRIGID
114 DM(I,J)=0.
      GOTU 117
115 DO 116 I=1,NRIGID
116 READ INPUT TAPE NTAPE2, 2, (DM(J,I),J=1,NRIGID)
      WRITE OUTPUT TAPE NTAPE3, 19,
      CALL MPRINT (DM,NRIGID,NRIGID,MAXQ,NTAPE3)

C*****
C READ MASS MATRIX FOR EACH SURFACE, STORE SYSTEM MASS MATRIX ON NTAPE4
      117 K1=0
      WRITE OUTPUT TAPE NTAPE3, 10
      DO 120 I=1,NSUR
      READ INPUT TAPE NTAPE2, 1, NSIZE
      READ INPUT TAPE NTAPE2, 1, (LDW(I),LHIGH(J),J=1,NSIZE)
      DO 119 J=1,NSIZE
      DO 118 K=1,NSIZE
118 A(K,J)=0.
      N1=LDW(I)

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VIBRATION AND FLUTTER ANALYSIS BY A MODAL METHOD, USING AICS.

```

N2=HIGH(J)
119 READ INPUT TAPE NTAPE2, 2, (A(N,J),N=N1,N2)
WRITE TAPE NTAPE4, NSIZE, NSIZE, ((A(N,J),N=1,NSIZE),J=1,NSIZE)
HML40808
HML40809
HML40810
HML40811
HML40812
HML40813
HML40814
HML40815
HML40816
HML40817
HML40818
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HML40839
HML40840
HML40841
HML40842
HML40843
HML40844
HML40845

WRITE OUTPUT TAPE NTAPE3, 13, 1, NSIZE
CALL MPRINT (A,NSIZE,NSIZE,MAXR,NTAPE3)
NSIZES(I)=NSIZE
120 K1=K1+NSIZE

C*****
C NPOINT = TOTAL NUMBER OF CONTROL POINTS ON ALL SURFACES.
NPOINT=K1

C*****
C READ IN RIGID BODY MODAL MATRIX FOR EACH SYSTEM. (HR)
IF ( NFUS )
121 READ INPUT TAPE NTAPE2, 1, NSIZE
DO 122 J=1,NRIGIO
122 READ INPUT TAPE NTAPE2, 2, (HR(I,J),I=1,NSIZE)
WRITE OUTPUT TAPE NTAPE3, 18, NSIZE
CALL MPRINT (HR,NSIZE,NRIGIO,MAXS,NTAPE3)
WRITE TAPE NTAPE6, NSIZE, NRIGIO, ((HR(I,J),I=1,NSIZE),J=1,NRIGIO)
123 IF ( NRIGIO )
124 K1=1
WRITE OUTPUT TAPE NTAPE3, 12
DO 126 ISUR=1,NSUR
READ INPUT TAPE NTAPE2, 1, NSIZE
K2=K1+NSIZE-1
DO 125 J=1,NRIGIO
125 READ INPUT TAPE NTAPE2, 2, (HR(I,J),I=K1,K2)
WRITE OUTPUT TAPE NTAPE3, 13, ISUR,NSIZE
CALL MPRINT (HR(K1,1),NSIZE,NRIGIO,MAXS,NTAPE3)
126 K1=K1+NSIZE

IF ( K1-NPOINT-1 )
127 WRITE OUTPUT TAPE NTAPE3, 11, K1, NPOINT
C*****

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VIBRATION AND FLUTTER ANALYSIS BY A MODAL METHOD, USING AICS.

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C READ IN FLEXIBLE MODAL MATRIX FOR EACH SURFACE, STORE SYSTEM ON NTAPES
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HMI40862
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HMI40864
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HMI40867
HMI40868
HMI40869
HMI40870
HMI40871
HMI40872
HMI40873
HMI40874
HMI40875
HMI40876
HMI40877
HMI40878
HMI40879
HMI40880
HMI40881
HMI40882
HMI40883

128 K1=0
K2=1
KS=0
KCANT = NFREE+1
REWIND NTAPES4
NSRFLX = NSRFLX
IF ( NFREE ) 129,131,129
129 DO 130 J=1,NSRFLX
DO 130 I=1,MAXR
H(I,J)=0.
FFM(I,J)=0.
130

131 WRITE OUTPUT TAPE NTAPES3, 14
DO 153 I=1,NSUR

READ INPUT TAPE NTAPES2, 1, NSIZE, NMODES, NFORM

IF ( NFREE ) 132,139,132
132 L=NFREE+1
IF (L-NSRFLX) 1321,1321,134
1321 DO 133 M=L,NSRFLX
DO 133 J=1,NSIZE
C(J,M)=0.
133
134 IF ( I-1 ) 136,135,136
135 CALL MREAD (FREQ,NSRFLX,1,NFORM,0,0,1,G,NSRFLX,NTAPES2,NTAPES3)
136 CALL MREAD (C,NSIZE,NFREE,NFORM,0,1,1,G,MAXR,NTAPES2,NTAPES3)
L=NMODES-NFREE
IF ( L ) 138,138,137
137 CALL MREAD (C(I,KCANT), NSIZE,L,NFORM,0,1,1,G,MAXR,NTAPES2,NTAPES3)
KCANT=KCANT+L
138 NMODES=NSRFLX
K2=1
GOTO 140

139 CALL MREAD (FREQ(K2),NMODES,1,NFORM,0,0,1,G,NMODES,NTAPES2,NTAPES3)

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VIBRATION AND FLUTTER ANALYSIS BY A MODAL METHOD, USING AICS.

```

      CALL MREAD (C,NSIZE,NMODES,NFORM,0,1,1,G,MAXR,NTAPE2,NTAPE3)
140  WRITE OUTPUT TAPE NTAPE3, 13, 1, NSIZE
      K=K2+NMODES -1
      WRITE OUTPUT TAPE NTAPE3, 9, NMODES, (FREQ(J),J=K2,K)
      CALL MPRINT (C,NSIZE,NMODES,MAXR,NTAPE3)
      WRITE TAPE NTAPE5, NSIZE, NMODES, ((C(J,K),J=1,NSIZE),K=1,
      NMODES)
1  C*****
C READ MASS (M), AND COMPUTE (M)*(H F)
      READ TAPE NTAPE4, J,J, ((A(K,J),K=1,NSIZE),J=1,NSIZE)

      CALL MMULTD (A,0,0,0,SAVE,NSIZE,NSIZE,NMODES,MAXR,MAXR,MAXR)
C*****
C IF REQUESTED, COMPUTE (HR)T * (M) * (H F)
      IF ( NFREE ) 141,146,141
141 IF ( NRIGID) 142,146,142
142 L=NFREE+1
      DO 143 N=1,NSIZE
      DO 143 J=1,NRIGID
      K=K1+N
      G(J,N)=HR(K,J)
143

      CALL MMULTD (G,0,0,SAVE,0,ARF,NRIGID,NSIZE,NFREE,MAXQ,MAXR,MAXQ)
      IF ( NCANT ) 146,146,144
144 DO 145 N=1,NRIGID
      DO 145 J=L,NSRFLX
      ARF(N,J)=0.
145

C*****
C COMPUTE (M FF) = (H F)TRANPOSED * (M) * (H F)
146 DO 147 M=1,NSIZE
      DO 147 J=1,NMODES
      A(J,M)=C(M,J)
147
      CALL MMULTD (A,0,0,SAVE,0,C,NMODES,NSIZE,NMODES,MAXR,MAXR,MAXR)
      IF ( NFREE ) 148,150,148
148 DO 149 M=1,NSRFLX
      DO 1481 J=1,NRIGID

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 HMI40921

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1481 HT(J,M)=HT(J,M)+ARF(J,M)
      DO 149 J=1,NSRFLX
149 FFM(M,J)=FFM(M,J)+C(M,J).
      GOTO 152
150 WRITE OUTPUT TAPE NTAPE3, 22, I
      CALL MPRINT (C,NMODES,NMODES,MAXR,NTAPE3)
      DO 151 J=1,NMODES
        KS=KS+1
151 SPRING(KS)=C(J,J)
152 K1=K1+NSIZE
153 K2=K2+NMODES
      NFLEX=K2-1

      IF ( K1-NPOINT ) 158,159,158
158 WRITE OUTPUT TAPE NTAPE3, 11, K1, NPOINT
159 IF ( NRIGID ) 160,1591,160
1591 IF ( NFREE ) 1592,164,1592
1592 NFLEX=NSRFLX
      GOTO 154

160 WRITE TAPE NTAPE4, K1, NRIGID, ((HR(N,J),N=1,K1),J=1,NRIGID)
C*****
C COMPUTE (A RF) = (DELTA M)INVERSE * (H R)TRANPOSED *(M) * (H F)
      IF ( NFREE ) 161,164,161
161 DO 162 I=1,NRIGID
      DO 162 J=1,NRIGID
        QQ(I,J)=DM(I,J)
162 NFLEX=NSRFLX
      IF ( NDELM ) 1621,163,1621
1621 CALL INVERS (DQ,NRIGID,IR)
      IF ( IR ) 310,163,310
163 CALL MMULTD (DQ,O,HT,O,G,NRIGID,NRIGID,NFLEX,MAXQ,MAXQ,MAXQ)
      DO 1630 N=1,NRIGID
        DO 1630 J=1,NFLEX
          G(N,J)=G(N,J)
1630 WRITE TAPE NTAPE4, NRIGID, VFLEX, ((G(N,J),N=1,NRIGID),J=1,NFLEX)
      DO 1631 I=1,NFLEX

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```

1631      DO 1631 J=1,NRIGID
           A(I,J)=G(J,I)
           CALL MMULTD(A,O,OM,O,C,NSRFLX,NRIGID,NRIGID,MAXR,MAXQ,MAXR)
           CALL MMULTD(C,O,G,O,A,NSRFLX,NRIGID,NSRFLX,MAXR,MAXQ,MAXR)
           DO 1632 I=1,NSRFLX
               DO 1632 J=1,NSRFLX
                   FFM(I,J)=FFM(I,J)+A(I,J)
1632      DO 1632 J=1,NSRFLX
           FFM(I,J)=FFM(I,J)+A(I,J)
154 DO 155 M=1,NSRFLX
155   SPRING(M)=FFM(M,M)
           WRITE OUTPUT TAPE NTAPE3, 22,
           CALL MPRINT (FFM,NSRFLX,NSRFLX,MAXR,NTAPE3)

164 NFLEX2=NC*NFLEX
           IF ( NAERO ) 166,165,166
165 NRHO=1
           DO 500 I=1,NRIGID
               DO 500 J=1,NR2
                   DQ(I,J)=0.
500   NCX=NC
           REWIND NTAPE9
           GOTO 215
C*****
C READ IN WEIGHTING MATRIX FOR EACH SURFACE.....STORE ON NTAPE6,
166   WRITE OUTPUT TAPE NTAPE3, 15
           DO 178 I=1,NSUR
               IF ( ISW(I) ) 167,177,167
167   N1=ISXST(I)
               IF ( N1 ) 168,172,168
168   DO 170 J=1,N1
                   DO 169 L=1,MAXR
                       A(J,L)=0.
                       A(L,J)=0.
169   A(J,J)=1.
170   A(J,J)=1.

           READ INPUT TAPE NTAPE2, 1, NXST, J, IFORM, IROW
           IF ( NXST ) 171,172,171
171   CALL MREAD (A,NXST,NXST,IFORM,IROW,0,1,F,MAXR,NTAPE2,NTAPE3)

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VIBRATION AND FLUTTER ANALYSIS BY A MODAL METHOD, USING AICS.

```

172 K=N1+1
    READ INPUT TAPE NTAPE2, 1, NSIZE, NPART, IFORM, IROW
    IF ( IFORM ) 174,173,174
173 CALL MREAD (A(K,K),NSIZE,NSIZE,0,0,0,1,F,MAXR,NTAPE2,NTAPE3)
    K=K+NSIZE
    GOTO 176

174 DO 175 J=1,NPART
    READ INPUT TAPE NTAPE2, 1, NSIZE
    N=K+NC
    DD 1741 M=K,N
    DD 1741 L=1,K
    A(L,M)=0.
1741 CALL MREAD (A(K,K),NSIZE,NSIZE,1,IRDW,0,0,F,MAXR,NTAPE2,NTAPE3)
175 K=K+NSIZE

176 K=K-1
C*****
    WRITE OUTPUT TAPE NTAPE3, 13, 1, K
    CALL MPRINT (A,K,K,MAXR,NTAPE3)
    WRITE TAPE NTAPE6, K, K, ((A(I,J),J=1,K),L=1,K)
    GOTO 178
177 WRITE OUTPUT TAPE NTAPE3, 21, 1
178 CONTINUE
179 REWIND NTAPE6
    REWIND NTAPE7
    REWIND NTAPE9
    NCX=NC
C*****
C IF SERIES OF DENSITIES FOR EACH V/8 OMEGA, READ IN THAT SERIES.
    IF ( NDENS ) 180,181,180
180 NRHD=NDENS
    GOTO 182
181 READ INPUT TAPE NTAPE2, 1, NRHD
    READ INPUT TAPE NTAPE2, 2, (RHD(I),I=1,NRHD)

```

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VIBRATION AND FLUTTER ANALYSIS BY A MODAL METHOD, USING AICS.

```

C*****
C READ IN COMPLEX AERODYNAMIC MATRIX FOR EACH SURFACE
182 NVEL=NVEL+1
  DO 210 I=1,NSURFS
    K=1
    K2=1
    IF ( NFUS )      183,184,183
    183 IF ( I-1 )    184,188,184
    184 L=1-NFUS
    IF ( ISXST(L) )  185,188,185
    185 K=ISXST(L)+1
    K2=2*ISXST(L)+1
    DO 186 J=1,K
      DO 186 L=1,K2
        186 A(I,J,L)=0.

      READ INPUT TAPE NTAPE2, 1, NXST, J, IFORM, IROW
      IF ( NXST ) 187,188,187

      N= NXST *NC
      CALL MREAD (A,NXST,N,IFORM,IROW,0,1,F,MAXR,NTAPE2,NTAPE3)

      187
      READ INPUT TAPE NTAPE2, 2, VELC
      READ INPUT TAPE NTAPE2, 1, NSIZE, NPART, IFORM, IROW
      IF ( IFORM ) 190,189,190

      N= NSIZE *NC
      CALL MREAD (A(K,K2),NSIZE,N,0,0,1,F,MAXR,NTAPE2,NTAPE3)
      NSIZE=NSIZE+K-1
      GOTO 193

      188
      DO 192 J=1,NPART
        READ INPUT TAPE NTAPE2, 1, NSIZE
        N=K2+NC
        DO 191 M=K2,N
          DO 191 L=1,K
            A(L,M)=0.
            N= NSIZE *NC
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VIBRATION AND FLUTTER ANALYSIS BY A MODAL METHOD, USING AICS.

```

192 CALL MREAD (A(K,K2),NSIZE,N,1,IROW,0,0,F,MAXR,NTAPE2,NTAPE3)
      K=K+NSIZE
      K2=K2+N
193 NSIZE=K-1
      N=K2-1
194 IF (I-1) 199,194,199
195 IF (NFUS) 195,197,195
196 WRITE OUTPUT TAPE NTAPE3, 16, NSIZE
197 CALL MPRINT (A,NSIZE,N,MAXR,NTAPE3)
      C*****
      C COMPUTE (DQ) = (HR)T * (CH) * (HR)
      K1=1
      K2=1
      M=NC-1
      READ TAPE NTAPE6, NSIZE, L, ((H(L,J),L=1,NSIZE),J=1,NRIGID)
      CALL MMULTD (A,M,H,O,F,NSIZE,NSIZE,NRIGID,MAXR,MAXR,MAXR)
      DO 196 L=1,NSIZE
      DO 196 J=1,NRIGID
196 A(J,L)=H(L,J)
      CALL MMULTD (A,O,F,M,DQ,NRIGID,NSIZE,NRIGID,MAXR,MAXR,MAXR)
      GOTO 210
197 DO 198 L=1,NRIGID
      DO 198 J=1,NR2
198 DQ(L,J)=0.
199 L=1-NFUS
      IF (ISW(L)) 202,200,202
200 WRITE TAPE NTAPE7, VSIZE, N, ((A(J,M),J=1,NSIZE),M=1,N)
      GOTO 208
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      HML41076
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      HML41109
      HML41110
      HML41111

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```

202 READ TAPE NTAPE6, L, L, ((T(J,M),J=1,L),M=1,L)
      IF ( NSIZE=L )      204,206,204
204 WRITE OUTPUT TAPE NTAPE3, 11, NSIZE,L
206 CALL MMULD (I,O,A,NC-1,F,L,L,MAXR,MAXR,MAXR)
      WRITE TAPE NTAPE7, NSIZE, N, ((F(J,M),J=1,NSIZE),M=1,N)
208 WRITE OUTPUT TAPE NTAPE3, 17, VELC
      L=-NFUS
      WRITE OUTPUT TAPE NTAPE3, 13, L, NSIZE
      CALL MPRINT (A,NSIZE,N,MAXR,NTAPE3)
210 CONTINUE

C*****
C CARRY ON FROM HERE TO END ONCE FOR EACH DENSITY.

215 DO 300 IKHO=1,NRHO
      IF ( NFREE )      216,219,216
216 DO 217 I=1,NFLEX
      DO 217 J=1,NFLEX2
217   SAVE(I,J)=0.
219 REWIND NTAPE4
      REWIND NTAPE5
      REWIND NTAPE7
      REWIND NTAPE8
      CON=RHO(I,RHO)*BR(1)**2*S(1) *32.174
      IF ( NRIGID )      218,222,218
218 DO 220 I=1,NRIGIO
      DO 220 J=1,NR2,NCX
      DMBAR(I,J+1)=CON*OQ(I,J+1)
      K=J/2+1
      DMBAR(I,J)=OM(I,K)+CON*OQ(I,J)
      BARMR(I,J)=DMBAR(I,J)
      BARMR(I,J+1)=DMBAR(I,J+1)
220
222 DO 236 ISUR=1,NSUR
      K=ISUR+NFUS

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HML41112
HML41113
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VIBRATION AND FLUTTER ANALYSIS BY A MODAL METHOD, USING AICS.

```

CON=RH0*(IRHO)*8R(K)**2*S(K) *32.174
C*****
C      READ (M) I
C*****
      READ TAPE NTAPE4, NSIZE, NSIZE, ((F(I,J),I=1,NSIZE),J=1,NSIZE)
      IF ( NAERD )
        221 DO 226 I=1,NSIZE
        DO 226 J=1,NSIZE
        226 A(I,J)=F(I,J)
        L=NSIZE
        GOTO 231
C*****
C      READ (W)*(CH) I
C*****
      223 READ TAPE NTAPE7, NSIZE, L, ((A(I,J),I=1,NSIZE),J=1,L)
C*****
C      COMPUTE (M BAR) =((M)+RH0*BR**2*S*(W)*(CH) ) I
C*****
      DO 228 I=1,NSIZE
      DO 228 J=1,L,NCX
      K=J/2+1
      A(I,J)=CON*A(I,J)+F(I,K)
      228 A(I,J+1)=CON*A(I,J+1)
      231 IF ( NRIGID ) 229,233,229
      229 WRITE TAPE NTAPE8, NSIZE, L, ((A(I,J),I=1,NSIZE),J=1,L)
C*****
C      READ (H F) INTO CORE STORAGE
C*****
      233 READ TAPE NTAPE5, NSIZE, NMODES, ((T(I,J),I=1,NSIZE),
      1 J=1,NMODES)
C*****
C      COMPUTE (M BAR) * (H F)
      N=NC-1

```

VIBRATION AND FLUTTER ANALYSIS BY A MODAL METHOD, USING AICS.

```

M= NMODES *NC
IF ( NRIGIO ) 224,225,224

224 CALL MMULTD (A,N,T,O,F,NSIZE,NSIZE,NMODES,MAXR,MAXR,MAXR)
225 WRITE TAPE NTAPE8, NSIZE, M, ((F(I,J),I=1,NSIZE),J=1,M)
DO 232 I=1,NSIZE
232 DO 232 J=1,NMODES
F(J,I)=F(I,J)

C*****
C COMPUTE (H F T) * (M BAR)
CALL MMULTD (F,O,A,N,T,NMODES,NSIZE,NSIZE,MAXR,MAXR,MAXR)
IF ( NRIGIO ) 227,230,227
227 WRITE TAPE NTAPE8, NMODES, L, ((T(I,J),I=1,NMODES),J=1,L)
230 DO 234 I=1,NSIZE
DO 234 J=1,NMODES
234 A(I,J)=F(J,I)

C*****
C THEN (H F T) * (M BAR) * (H F)
CALL MMULTD (T,N,A,O,F,NMODES,NSIZE,NMODES,MAXR,MAXR,MAXR)
IF ( NFREE ) 428,235,428
428 DO 430 I=1,NFLEX
DO 430 J=1,NFLEX2
430 SAVE(I,J)=SAVE(I,J)+F(I,J)
GOTO 236
235 WRITE TAPE NTAPE9, NMODES, M, ((F(I,J),I=1,NMODES),J=1,M)
236 CONTINUE
IF ( NFREE ) 432,237,432
432 WRITE TAPE NTAPE9, NFLEX, NFLEX2, ((SAVE(I,J),I=1,NFLEX),
1 J=1,NFLEX2)

C*****
C IF NRIGIO=0, THATS ALL FOR THIS DENSITY.
C IF NOT, READ HR MATRIX FROM NTAPE4
237 REWIND NTAPE8
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VIBRATION AND FLUTTER ANALYSIS BY A MODAL METHOD, USING AICS.

```

      IF (NRIGID) 238,300,238
      READ TAPE NTAPE4, NPOINT, NRIGID, ((HR(I,J),I=1,NPOINT),
1      J=1,NRIGID)
      KI=0
      K2=1
      DO 244 ISUR=1,NSUR
      C*****
      C READ (M BAR) AND COMPUTE (H R)TRANPOSED * (M BAR) * (H R) + (DELTA M)
      READ TAPE NTAPE8, NSIZE, M, ((A(I,J),I=1,NSIZE),J=1,M)
      DO 240 I=1,NSIZE
      DO 240 J=1,NRIGID
      K=K1+1
      G(J,I)=HR(K,J)
      CALL MMULTD (G,O,A,N,HT,NRIGID,NSIZE,NSIZE,MAXQ,MAXR,MAXQ)
      CALL MMULTD (HT,N,HR(K1+1,1),O,A,NRIGID,NSIZE,NRIGID,MAXQ,
1      MAXS,MAXR)
      DO 242 I=1,NRIGID
      DO 242 J=1,NR2
      BARMRR(I,J)=BARMRR(I,J)+A(I,J)
      C*****
      C FIND (H R)TRANPOSED * (M BAR) * (H F)
      (H F)TRANPOSED * (M BAR) * (H R) AND STORE ON NTAPE9
      READ TAPE NTAPE8, NSIZE, M, ((A(I,J),I=1,NSIZE),J=1,M)
      CALL MMULTD (G,O,A,N,HT,NRIGID,NSIZE,NFLEX,MAXQ,MAXR,MAXQ)
      WRITE TAPE NTAPE9, NRIGID,M,((HT(I,J),I=1,NRIGID),J=1,M)
      READ TAPE NTAPE8, NMODES, L, ((A(I,J),I=1,NMODES),J=1,L)
      CALL MMULTD (A,N,HR(K1+1,1),O,T,NMODES,NSIZE,NRIGID,MAXR,MAXS,
1      MAXR)
      WRITE TAPE NTAPE9, NMODES, NR2, ((T(I,J),I=1,NMODES),J=1,NR2)
      KI=KI+NSIZE
      C*****
      IF (NFREE) 246,250,246
      READ TAPE NTAPE4, NSIZE, NMODES, ((ARF(I,J),I=1,NSIZE),J=1,NMODES)
      CALL MMULTD (DMBAR,N,ARF,O,HT,NRIGID,NRIGID,NMODES,MAXQ,MAXR,MAXQ)
      DO 248 I=1,NRIGID

```

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VIBRATION AND FLUTTER ANALYSIS BY A MODAL METHOD, USING AICS.

```

      DD 248 J=1,NMODES
      SAVE(J,1) = ARF(I,J)
248  CALL MMULTD (SAVE,O,HT,N,A,NMODES,NRIGID,NMODES,MAXR,MAXQ,MAXR)
      M= NMODES *NC
      WRITE TAPE NTAPE9, NMODES, M, ((A(I,J),I=1,NMODES),J=1,M)
      WRITE TAPE NTAPE9, NRIGID, M, ((HT(I,J),I=1,NRIGID),J=1,M)
      CALL MMULTD (SAVE,O,DMBAR,N,A,NMODES,NRIGID,NRIGID,MAXR,MAXQ,MAXR)
      WRITE TAPE NTAPE9, NMODES, NR2, ((A(I,J),I=1,NMODES),J=1,NR2)
      HML41264
      HML41265
      HML41266
      HML41267
      HML41268
      HML41269
      HML41270
      HML41271
      HML41272
      HML41273
      HML41274
      HML41275
      HML41276
      HML41277
      HML41278
      HML41279
      HML41280
      HML41281
      HML41282
      HML41283
      HML41284
      HML41285
      HML41286
      HML41287
      HML41288
      HML41289
      HML41290
      HML41291
      HML41292
      HML41293
      HML41294
      HML41295
      HML41296
      HML41297
      HML41298
      HML41299
      HML41300
      HML41301

C*****
C COMPUTE (M BAR RR) INVERSE, AND STORE ON TAPE NTAPE9.
250 GDTU (256,252),NCX
252 DD 254 I=1,NRIGID
      DD 254 J=1,NR2,NCX
      K=J/2+1
      G(I,K)=BARMRR(I,J+1)
254  BARMRR(I,K)=BARMRR(I,J)

256 CALL MNVRX (BARMRR,G,ARF(1,1),ARF(1,MAXQ),NRIGID,IR,NC-1)

      IF ( IR ) 310,258,310
258 GOTO (264,260),NCX
260 DD 262 I=1,NRIGID
      DD 262 J=1,NR2,NCX
      K=NR2-J
      L=K/2+1
      BARMRR(I,K)=BARMRR(I,L)
262  BARMRR(I,K+1)=G(I,L)

264 WRITE TAPE NTAPE9, NRIGID, NR2, ((BARMRR(I,J),I=1,NRIGID),J=1,NR2)
      HML41293
      HML41294
      HML41295
      HML41296
      HML41297
      HML41298
      HML41299
      HML41300
      HML41301

      K=1
      DD 266 ISUR=1,NSUR
      L=NSIZES(ISUR)
      CALL MMULTD (HR(K,1),O,BARMRR,NC-1,A, L ,NRIGID,NRIGID,
      1  MAXS,MAXQ,MAXR)
      K=K+L
266  WRITE TAPE NTAPE9, L, NR2, ((A(I,J),I=1,L),J=1,NR2)

```

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VIBRATION AND FLUTTER ANALYSIS BY A MODAL METHOD, USING AICS.

HMI41302
HMI41303
HMI41304
HMI41305
HMI41306
HMI41307

300 CONTINUE

CALL CHAIN (2,NTAPE1)

310 WRITE OUTPUT TAPE NTAPE3, 20

GOTO 100

END(1,0,0,0,0,0,0,0,0,1,0,0,0,0,0)

STORAGE NOT USED BY PROGRAM

DEC OCT
11041 25441

STORAGE INDICATORS FOR VARIABLES APPEARING IN COMMON STATEMENTS

	DEC	OCT	DEC	OCT	DEC	OCT	DEC	OCT	DEC	OCT
AITKEN	32044	76454	ARF	21041	51061	A	32041	76451	BARMR	32187
BOOLT	32473	77331	BREF	32411	77233	BR	32409	77231	C	29541
DBHAR	32259	77003	DM	32367	77157	DQ	32331	77113	EPDP	32047
EPSP	32048	76460	FFM	13541	32345	FREQ	32451	77303	F	27041
G	20201	47351	HR	27041	64641	H	20801	50501	HT	20801
ISW	32541	77435	ISXT	32561	77461	IT	32561	77461	LHIGH	20201
LOW	21041	51061	MAXP	32480	77340	MAXP	32482	77342	MAXR	32483
MAXS	32481	77341	MODES	32486	77346	NAERD	32115	76563	NCANT	32489
NCARDS	32049	76461	NC	32476	77334	NDENS	32487	77347	NFLEX2	32477
NFLX	32484	77344	NFREE	32490	77352	NFUS	32488	77350	NGO	32475
NITRDP	32045	76455	NITRSP	32046	76456	NPOINT	32485	77345	NPUNCH	32474
NR2	32478	77336	NRHO	32114	76562	NRIGID	32491	77353	NSIZES	32521
NSRFLX	32410	77232	NSURFS	32479	77337	NSUR	32492	77354	NTAPE0	32050
NTAPE1	32501	77365	NTAPE2	32500	77364	NTAPE3	32499	77356	NTAPE4	32498
NTAPE5	32497	77361	NTAPE6	32496	77360	NTAPE7	32495	77357	NTAPE8	32494
NTAPE9	32493	77355	NVEL	32051	76463	PICON	32113	76561	RHO	32471
SAVE	16041	37251	SPRING	32112	76560	S	32388	77204	T	21041
VELTY	32072	76510								51061

STORAGE LOCATIONS FOR VARIABLES NOT APPEARING IN COMMON. DIMENSION. OR EQUIVALENCE STATEMENTS

	CON	DEC	OCT	IFORM	DEC	OCT	IRHO	DEC	UCT	LROM	DEC	OCT
	KR	4745	11211		4744	11210		4743	11207		4742	11206
	IK	4741	11205	I	4740	11204	ISUR	4739	11203	J	4738	11202
	KI	4737	11201	K2	4736	11200	KCANT	4735	11177	K	4734	11176
	KS	4733	11175	L	4732	11174	M	4731	11173	N1	4730	11172
	N2	4729	11171	NCON	4728	11170	NCX	4727	11167	NDELM	4726	11166
	NFORM	4725	11165	NMODES	4724	11164	NPART	4723	11163	N	4722	11162
	NPSIZE	4721	11161	NXST	4720	11160	VELC	4719	11157			

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VIBRATION AND FLUTTER ANALYSIS BY A MODAL METHOD, USING AICS.

SYMBOLS AND LOCATIONS FOR SOURCE PROGRAM FORMAT STATEMENTS

EFN	LOC	EFN	LOC	EFN	LOC
8J1	1 11100	8J2	2 11076	8J3	3 11074
8J5	5 10765	8J6	6 10744	8J7	7 10735
8J9	9 10706	8JA	10 10674	8JB	11 10666
8JD	13 10621	8JE	14 10607	8JF	15 10600
8JH	17 10553	8JI	18 10532	8JK	19 10512
8JL	21 10467	8JM	22 10451	8JN	23 10441
				8JO	24 10436

LOCATIONS FOR OTHER SYMBOLS NOT APPEARING IN SOURCE PROGRAM

DEC	LOC	DEC	LOC	DEC	LOC
1J	4673 11101	2J	4325 10345	3J	4342 10366
6J	4351 10377	AJ108	4229 10205	AJ108	4238 10216
AJ10M	4264 10250	AJ100	4277 10265	AJ10P	4290 10302
AJ10T	4312 10330	CJ60	4677 11105	CJ62	4678 11106
CJ64	4680 11110	CJ66	4681 11111	CJ67	4682 11112
CJ69	4684 11114	CJ6A	4685 11115	CJ100	4686 11116
CJ105	4688 11120	CJ106	4689 11121	CJ108	4690 11122
CJ108	4692 11124	CJ10K	4693 11125	CJ10M	4694 11126
CJ10P	4696 11130	CJ10S	4697 11131	CJ10T	4698 11132
CJ201	4700 11134	CJ202	4701 11135	CJ203	4702 11136
CJ205	4704 11140	CJ206	4705 11141	CJ207	4706 11142
CJ209	4708 11144	CJ20A	4709 11145	CJ20B	4710 11146
CJ20D	4712 11150	CJ20E	4713 11151	CJ20F	4714 11152
CJ20H	4716 11154	CJ20I	4717 11155	CJ20J	4718 11156
DJ12M	953 01671	DJ161	2163 04163	DJ190	2984 05650
DJ1CK	4030 07676	DJ25A	1683 03223	DJ263	1997 03715
DJ205	1752 03330	DJ2C9	3890 07462	DJ2CH	3996 07634
DJ2AR	3402 06512	DJ2CT	4133 10045	DJ361	2162 04162
DJ2CO	4084 07764	DJ3AR	3401 06511	DJ3CD	4083 07763
DJ390	2983 05647	DJ418	484 00744	DJ436	1174 02226
DJ403	98 00142	DJ45L	1821 03435	DJ461	1992 03710
DJ45I	1752 03330	DJ47V	2642 05122	DJ488	2720 05240
DJ47T	2634 05112	DJ48S	2930 05562	DJ4CG	3992 07630
DJ48I	2841 05431	DJ588	2719 05237	DJ505	4220 10174
DJ56L	2224 04260	DJ761	2161 04161	DJ790	2982 05646
DJ67V	2641 05121				

DEC	LOC	DEC	LOC	DEC	LOC
4J	32767 77777	4J	32767 77777	4J	32767 77777
AJ10K	4251 10233	AJ10K	4251 10233	AJ10K	4251 10233
AJ10S	4303 10317	AJ10S	4303 10317	AJ10S	4303 10317
CJ63	4679 11107	CJ63	4679 11107	CJ63	4679 11107
CJ68	4683 11113	CJ68	4683 11113	CJ68	4683 11113
CJ101	4687 11117	CJ101	4687 11117	CJ101	4687 11117
CJ10A	4691 11123	CJ10A	4691 11123	CJ10A	4691 11123
CJ100	4695 11127	CJ100	4695 11127	CJ100	4695 11127
CJ10U	4699 11133	CJ10U	4699 11133	CJ10U	4699 11133
CJ204	4703 11137	CJ204	4703 11137	CJ204	4703 11137
CJ208	4707 11143	CJ208	4707 11143	CJ208	4707 11143
CJ20C	4711 11147	CJ20C	4711 11147	CJ20C	4711 11147
CJ20G	4715 11153	CJ20G	4715 11153	CJ20G	4715 11153
DJ100	356 00544	DJ100	356 00544	DJ100	356 00544
DJ1AM	3360 06440	DJ1AM	3360 06440	DJ1AM	3360 06440
DJ26R	2329 04431	DJ26R	2329 04431	DJ26R	2329 04431
DJ2CI	4000 07640	DJ2CI	4000 07640	DJ2CI	4000 07640
DJ36R	2328 04430	DJ36R	2328 04430	DJ36R	2328 04430
DJ3CT	4132 10044	DJ3CT	4132 10044	DJ3CT	4132 10044
DJ43F	1271 02367	DJ43F	1271 02367	DJ43F	1271 02367
DJ46L	2225 04261	DJ46L	2225 04261	DJ46L	2225 04261
DJ48C	2775 05327	DJ48C	2775 05327	DJ48C	2775 05327
DJ405	4221 10175	DJ405	4221 10175	DJ405	4221 10175
DJ636	1173 02225	DJ636	1173 02225	DJ636	1173 02225
EJ25	776 01410	EJ25	776 01410	EJ25	776 01410

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VIBRATION AND FLUTTER ANALYSIS BY A MODAL METHOD, USING AICS.

E12J	907 01613	E12L	928 01640	E12M	952 01670	E12O	991 01737
E134	1155 02203	E13R	1351 02507	E15B	1686 03226	E17C	2477 04655
E18N	2865 05461	E196	3022 05716	E19G	3115 06053	E19Q	3185 06161
E1A4	3244 06254						

LOCATIONS OF NAMES IN TRANSFER VECTOR

DEC	UCT	DEC	OCT	DEC	OCT	DEC	OCT
CHAIN	17 00021	INVERS	15 00017	MMULTO	14 00016	MNVRX	16 00020
MPRINT	8 00010	MREAD	11 00013	ROLN	3 00003	SETH	1 00001
(FIL)	7 00007	(FPT)	0 00000	(RLK)	13 00015	(RTN)	5 00005
(WLT)	2 00002	(STB)	9 00011	(STH)	6 00006	(TSH)	12 00014
(TSH)	4 00004	(WLR)	10 00012				

ENTRY POINTS TO SUBROUTINES NOT OUTPUT FROM LIBRARY

CHAIN	INVERS	MMULTO	MNVRX	MPRINT	MREAD	RDLN	SETH
(FIL)	(FPT)	(RLK)	(RTN)	(RWT)	(STB)	(STH)	(TSH)
(TSH)	(WLR)						

EXTERNAL FORMULA NUMBERS WITH CORRESPONDING INTERNAL FORMULA NUMBERS AND OCTAL LOCATIONS

EFN	IFN	LOC	EFN	IFN	LOC	EFN	IFN	LOC
98	31 00034	99	33 00042	100	61 00143	1011	68 00241	
1012	72 00305	102	76 00321	103	79 00334	104	80 00342	
105	81 00344	106	100 00437	107	105 00456	108	106 00460	
109	108 00474	110	112 00520	111	115 00545	112	116 00547	
113	117 00553	114	119 00566	115	121 00576	116	122 00614	
117	131 00653	118	143 00745	119	146 00756	120	165 01074	
121	168 01111	122	171 01135	123	189 01230	124	190 01232	
125	197 01314	126	206 01364	127	208 01377	128	210 01411	
129	217 01451	130	220 01471	131	221 01502	132	226 01534	
1321	228 01545	133	230 01601	134	231 01614	135	232 01621	
136	234 01641	137	238 01672	138	241 01724	139	244 01740	
140	248 01775	141	280 02204	142	281 02206	143	285 02256	
144	289 02323	145	291 02360	146	292 02370	147	294 02406	
148	298 02440	1481	300 02465	149	302 02473	150	304 02510	

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VIBRATION AND FLUTTER ANALYSIS BY A MODAL METHOD, USING AICS.

151	310	02542	152	311	02546	153	312	02553	158	315	02575
159	317	02605	1591	318	02611	1592	319	02613	160	321	02616
161	331	02656	162	333	02671	1621	336	02704	163	339	02714
1630	343	02747	1631	355	03032	1632	362	03113	154	363	03123
155	364	03127	164	369	03146	165	371	03156	500	374	03173
166	378	03211	167	381	03227	168	383	03233	169	386	03254
170	387	03263	171	391	03311	172	393	03331	173	397	03372
174	401	03431	1741	407	03507	175	410	03541	176	411	03561
177	426	03675	178	428	03711	179	429	03716	180	434	03732
181	436	03735	182	443	03766	183	448	04026	184	449	04031
185	451	04040	186	455	04103	187	459	04136	188	462	04164
189	467	04222	190	472	04254	191	478	04333	192	483	04406
193	486	04432	194	487	04437	195	488	04441	196	508	04570
197	512	04623	198	514	04636	199	515	04645	200	517	04656
202	527	04715	204	537	04761	206	539	04771	208	550	05054
210	557	05113	215	558	05123	216	560	05131	217	562	05144
219	563	05153	218	569	05177	220	575	05304	222	576	05330
221	589	05432	226	591	05445	223	594	05462	228	607	05623
231	608	05651	229	609	05653	233	618	05717	224	630	05772
225	641	06054	232	643	06072	227	647	06122	230	656	06162
234	658	06200	428	662	06230	430	664	06243	235	666	06255
236	675	06313	432	677	06320	237	686	06356	238	688	06362
240	712	06542	242	719	06645	244	760	07127	246	762	07142
248	775	07240	250	808	07463	252	809	07466	254	813	07563
256	814	07605	258	817	07641	260	818	07644	262	823	07742
264	824	07765	266	839	10121	300	848	10165	310	851	10176

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PART 2.....VIBRATION AND FLUTTER ANALYSIS BY A MODAL METHOD.

DIMENSION	ISXST(20), ISW(20), BR(21), S(21), RHO(20), DM(6,6),	HMI41627
1	DQ(6,12), OMBAR(6,12), BARMRR(6,12), FREQ(40)	HMI41628
2	, NSIZES(20), IT(520), SPRING(40), VELCTY(20),	HMI41629
3	HFF(40,80), HRR(6,80), HFR(40,12), ARF(40,80),	HMI41630
4	HRR(6,12), HRRR(6,80), A(40,160), GUESS(40,2),	HMI41631
5	VECTOR(40,50), EIGVAL(50), NITER(25), US(40,50),	HMI41632
6	H(40,58), NAKSR(25), NAKOR(25), BOOLT(2)	HMI41633
7	, HFI(49,100), HF(49,40), H1(49,46), OMEGA(25)	HMI41634
8	, B(40,6)	HMI41635
		HMI41636
EQUIVALENCE	(IT(1),ISXST(1)), (IT(21),ISW(1)), (IT(41),NSIZES)	HMI41637
1	, (IT(61),NTAPE1), (IT(62),NTAPE2), (IT(63),NTAPE3),	HMI41638
2	(IT(64),NTAPE4), (IT(65),NTAPE5), (IT(66),NTAPE6),	HMI41639
3	(IT(67),NTAPE7), (IT(68),NTAPE8), (IT(69),NTAPE9),	HMI41640
4	(IT(70),NSUR), (IT(71),NRIGID), (IT(72),NFREE),	HMI41641
5	(IT(73),NCANT), (IT(74),NFUS), (IT(75),NDENS),	HMI41642
6	(IT(76),MODE), (IT(77),NPOINT), (IT(78),NFLEX),	HMI41643
7	(IT(79),MAXR), (IT(80),MAXQ), (IT(81),MAXS),	HMI41644
8	(IT(82),MAXP), (IT(83),NSURFS), (IT(84),NR2),	HMI41645
9	(IT(85),NFLEX2), (IT(86),NC), (IT(87),NGO)	HMI41646
EQUIVALENCE	(IT(88),NPUNCH), (IT(89),BOOLT), (IT(91),RHO),	HMI41647
1	(IT(111),FREQ), (IT(151),BREF), (IT(152),NSRFLX),	HMI41648
2	(IT(153),PR), (IT(174),S), (IT(195),DM), (IT(231),	HMI41649
3	DQ), (IT(303),DMBAR), (IT(375),BARMRR),	HMI41650
4	(IT(447),NAERO), (IT(448),NRHO), (IT(449),PICON),	HMI41651
5	(IT(450),SPRING), (IT(490),VELOCITY), (IT(511),NVEL),	HMI41652
6	(IT(512),NTAPE0), (IT(513),NCARDS), (IT(514),EPS),	HMI41653
7	(IT(515),EPDP), (IT(516),NITRSP), (IT(517),NITROP),	HMI41654
8	(IT(518),AITKEN), (A,HFF,HFI), (US,HFR), (H1,H),	HMI41655
9	(A(4901),HF,ARF), (VECTOR,HRRR), (VECTOR(961),HRR)	HMI41656
EQUIVALENCE	(NAKSR,OMEGA), (B,HFF)	HMI41657
		HMI41658
COMMON	IT, A, VECTOR, HRF, GUESS, EIGVAL, US, NITER, H, NAKSR,	HMI41659
1	NAKCR	HMI41660
200	FORMAT (1H1, 20X, 5H MODE 7X, 12H OMEGA (CPS) 10X, 8H DAMPING	HMI41661
1	4X, 17H VELOCITY (KNOTS) ///)	HMI41662
201	FORMAT (1H 21X, 114, 3E20.8)	HMI41663

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PART 2.....VIBRATION AND FLUTTER ANALYSIS BY A MODAL METHOD.

```

202 FORMAT (1H1 48X, 12H OUTPUT DATA // 1H 16X, 7HFLUTTER
1 55H ANALYSIS BY A MODAL METHOD USING AEROYNAMIC INFLUENCEHMI41664
2 13H COEFFICIENTS // 1H0 14X, 11H DENSITY = 1E20.8, HMI41665
3 5X, 20H REDUCED VELOCITY = 1E20.8, // 1H0 12X, 116, HMI41666
4 32H RIGID BODY DEGREES OF FREEDOM, 12X, 116, 6H INPUT HMI41667
5 16H MODES (TOTAL). //// ) HMI41668
203 FORMAT (1H0 40X, 21H FINAL FLUTTER MODES //// ) HMI41669
204 FORMAT (1H0 38X, 8HSURFACE 112, 1H, 116, 15H CONTROL POINTS ) HMI41670
205 FORMAT ( 214, 62X, 1A6, 114 ) HMI41671
206 FORMAT ( 1H0 33X, 23H PUNCHED CARDS NUMBERS 1A6, 114, 6H THRU HMI41672
1 1A6, 114 ) HMI41673
207 FORMAT ( 1H0 45X, 16H DYNAMIC MATRIX ) HMI41674
208 FORMAT (5H MOOE 116, 32H, GIVES AN IMAGINARY FREQUENCY. ) HMI41675
209 FORMAT (1H0 /// 1H0 40X, 20H GENERALIZED MASSES //// HMI41676
1 (1H 30X, 5H MASS 114, 3H = 1E16.8 ) ) HMI41677
8 BCOZ = 606030440104 HMI41678
MAXR = 49 HMI41679
HMI41680
HMI41681
HMI41682
HMI41683
HMI41684
HMI41685
HMI41686
HMI41687
HMI41688
HMI41689
HMI41690
HMI41691
HMI41692
HMI41693
HMI41694
HMI41695
HMI41696
HMI41697
HMI41698
HMI41699
HMI41700
HMI41701

00 301 I=1,NRIGID
00 301 J=1,NFLEX2
HRF(I,J)=0.
301 00 304 I=1,NFLEX
00 302 J=1,NFLEX2
HFF(I,J)=0.
302 00 304 J=1,NR2
HFR(I,J)=0.
304

L=0
K=C

```

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PART 2.....VIBRATION AND FLUTTER ANALYSIS BY A MODAL METHOD.

```

IF ( NFREE )      290,294,290
290 READ TAPE NTAPE9, NSIZE, N, ((HFF(I,J),I=1,NSIZE ),J=1,N)
GOTO 296
C*****
C READ AND POSITION ALL ( M BAR FF )
294 DO 307 ISUR=1,NSUR
READ TAPE NTAPE9, NSIZE, N, ((ARF(I,J),I=1,NSIZE),J=1,N)
DO 306 I=1,NSIZE
K=K+1
DO 305 J=1,N
L=L+1
HFF(K,L)=ARF(I,J)
305
306 L=L+N
307 L=L+N
IF ( NRGID )      296,297,296
297 DO 298 I=1,NFLEX
DO 298 J=1,NFLEX2
298 ARF(I,J)=0.
GOTO 350
296 K=0
L=0
C*****
C READ AND POSITION ALL ( M BAR FR ) AND ( M BAR RF )
DO 312 ISUR=1,NSUR
READ TAPE NTAPE9, NSIZE, N, ((ARF(I,J),I=1,NSIZE ),J=1,N)
DO 309 I=1,NSIZE
DO 308 J=1,N
L=L+1
HRF(I,L)=HRF(I,L)+ARF(I,J)
L=L+N
308
309 L=L+N
READ TAPE NTAPE9, NSIZE, N, ((ARF(I,J),I=1,NSIZE),J=1,N)
DO 310 I=1,NSIZE
K=K+1
DO 310 J=1,N
HFR(K,J)=HFR(K,J)+ARF(I,J)
310

```

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PART 2.....VIBRATION AND FLUTTER ANALYSIS BY A MODAL METHOD.

```

IF ( NFREE )      311,312,311
311 L=0
K=D
312 CONTINUE

C NFLEX = TOTAL NUMBER OF FLEXIBLE MODES.

IF ( NFREE )      314,322,314
314 READ TAPE NTAPE9, NSIZE, N, ((ARF(I,J),I=1,NSIZE),J=1,N)
DO 316 I=1,NFLEX
  DO 316 J=1,NFLEX2
    HFF(I,J)=HFF(I,J)+ARF(I,J)
316 READ TAPE NTAPE9, NSIZE, N, ((ARF(I,J),I=1,NSIZE),J=1,N)
DO 318 I=1,NRIGID
  DO 318 J=1,NFLEX2
    HRF(I,J)=HRF(I,J)+ARF(I,J)
318 READ TAPE NTAPE9, NSIZE, N, ((ARF(I,J),I=1,NSIZE),J=1,N)
DO 320 I=1,NFLEX
  DO 320 J=1,NR2
    HFR(I,J)=HFR(I,J)+ARF(I,J)
320 HFR(I,J)=HFR(I,J)+ARF(I,J)

322 READ TAPE NTAPE9, NSIZE, N, ((HRR(I,J),I=1,NSIZE),J=1,N)
C*****
C CDMPUTE (M BAR FR) * (M BAR RR)INV * (M BAR RF)
CALL DPMLTX (HRR,NC ,HRF,NC ,HRRR ,NRIGID,NFLEX,MAXQ,
1 MAXQ,MAXQ,1)
IF ( NAERO )      326,324,326
324 WRITE TAPE NTAPE7, ((HRRR (I,J),I=1,NRIGID),J=1,NFLEX)
326 CONTINUE
CALL DPMLTX (HFR,NC ,HRRR ,NC ,ARF,NFLEX,NRIGID,NFLEX,MAXP,
1 MAXQ,MAXP,1)
350 IF ( NAERO )      352,351,352
351 WRITE TAPE NTAPE7, ((HFF(I,J),I=1,NFLEX),J=1,NFLEX)
352 DO 354 I=1,NFLEX
  FFM=SPRING(I)*FREQ(I)**2
  DO 354 J=1,NFLEX2
    HFF(I,J)= (HFF(I,J)-ARF(I,J)) / FFM
354

```

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PART 2.....VIBRATION AND FLUTTER ANALYSIS BY A MODAL METHOD.

```

C*****
WRITE OUTPUT TAPE NTAPE3, 202, RHO(IRHO), VELOCITY(NVEL), NRGID,
1
DO 355 I=1,MODES
NAKSR(I)=0
355 NAKDR(I)=0
IF ( NPUNCH ) 356,358,358
356 WRITE OUTPUT TAPE NTAPE3, 207
CALL MPRINT (HFF,NFLEX,NFLEX2,MAXP,NTAPE3)
C*****
356 CALL MITERS (HFF,NTAPE8,NFLEX,GUESS,O,MODES,VECTOR,EIGVAL,NITER,
1
NITRSP, NITROP, EPSP, EPOP, IR, US, H, MAXP, NC,
2
AITKEN, NAKSR, NAKDR, NTAPE3 )
IF ( NAERO ) 3584,3582,3584
3582 WRITE TAPE NTAPE7, ((VECTOR(I,J),I=1,NFLEX),J=1,MODES)
REWIND NTAPE7
3584 CONTINUE
REWIND NTAPE8
WRITE OUTPUT TAPE NTAPE3, 200
MODES2=NC*MODES
DO 364 I=1,MODES2,NC
J=I/NC+NC-1
IF ( EIGVAL(I) ) 359,360,360
359 WRITE OUTPUT TAPE NTAPE3, 208, J
OMEGA(J)=0.
GOTO 361
360 OMEGA(J)=1. / SQRT( EIGVAL(I) )
DAMP=0.
VELC=0.
361 GOTO (364,362),NC
362 DAMP= EIGVAL(I+1) / EIGVAL(I)
VELC = PICON*OMEGA(J)*REF*VELOCITY(NVEL)
364 WRITE OUTPUT TAPE NTAPE3, 201, J, OMEGA(J), OAMP, VELC
IF ( NPUNCH ) 366,368,366
366 WRITE OUTPUT TAPE NTAPE0, 205, NSIZES(1), MODES,BCDZ,NCARDS
NCARDS=NCARDS+1

```


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PART 2.....VIBRATION AND FLUTTER ANALYSIS BY A MODAL METHOD.

```

CALL MPUNCH (OMEGA,J,1,0,1,1,BCDZ,MAXR,NTAPEO,NCRDS )
WRITE OUTPUT TAPE NTAPE3, 206, 8CDZ, NCRDS, NCRDS
NCRDS=NCRDS+1
368 L=1
K1=0
C*****
WRITE OUTPUT TAPE NTAPE3, 203
DO 378 ISUR=1,NSUR
IF ( NRIGID ) 371,369,371
369 READ TAPE NTAPE5, NSIZE, NMODES, ((HF1(I,J),I=1,NSIZE),J=1,NMODES)
CALL DPMLTX (HF1,1,VECTOR,NC,HF,NSIZE,NMODES,MODES,MAXR,MAXP,
1 MAXR,1)
NCMODE=NC*NMODES
GOTO 375
371 READ TAPE NTAPE9, NSIZE, I,((H1(I,J),I=1,NSIZE),J=1,NR2)
READ TAPE NTAPE5, I,NMODES, ((HF1(I,J),I=1,NSIZE), J=1,NMODES)
NCMODE=NC*NMODES
CALL DPMLTX (H1,NC,HF,NC,HF1,NSIZE,NRIGID,NFLEX,MAXR,MAXQ,MAXR,1)
DO 374 I=1,NSIZE
DO 372 J=1,NFLEX2
372 HF1(I,J)=-HF1(I,J)
DO 374 J=1,NCMODE,NC
K= J/NC +NC -1
L=K1+J
374 HF1(I,L)=HF1(I,L)+HF(I,K)
IF ( NFREE ) 3742,3741,3742
3741 K1=K1+NCMODE
3742 CALL DPMLTX (HF1,NC,VECTOR,NC,HF,NSIZE,NFLEX,MODES,MAXR,
1 MAXP,MAXR,1)
375 DO 376 I=1,MODES2,NC
INDEX=0
K=ISUR*NC-NC+1
L=I/NC+NC-1
376 CALL NPNRMX (HF1(I,1),HF(1,1),NSIZE,US(K,L),INDEX,MAXR,NC,1)
IF ( NSUR-1 ) 378,377,378
377 DO 3775 L=1,MODES2,NC
I=L/NC +NC -1
H(1,1)=US(1,1)

```

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PART 2.....VIBRATION AND FLUTTER ANALYSIS BY A MODAL METHOD.

```

3775 H(2,I)=US(2,I)
      GOTO 401
378 WRITE TAPE NTAPE8, NSIZE, MODES2, ((HF(I,J),I=1,NSIZE),J=1,MODES2)
      REWIND NTAPE8
      DO 400 L=1,MODES2,NC
        I=L/NC + NC - 1
        NAKOR(I)=1
        GOTO (384,392),NC
384 H(1,I)=US(1,I)
      DO 390 ISUR=2,NSUR
        IF (H(1,I)-US(ISUR,I)) 388,390,390
388 H(1,I)=US(ISUR,I)
        NAKOR(I)=ISUR
390 CONTINUE
      GOTO 400
392 INCX=1
      H(1,I)=US(1,I)**2+US(2,I)**2
      J=2*NSUR
      DO 398 ISUR=3,J,2
        IF (H(1,I)-(US(ISUR,I)**2+US(ISUR+1,I)**2)) 396,398,398
396 INDEX=ISUR
        H(1,I)=(US(ISUR,I)**2+US(ISUR+1,I)**2)
        NAKOR(I)=ISUR/2+1
398 CONTINUE
399 H(1,I)=US(INDEX,I)
      H(2,I)=US(INDEX+1,I)
      400 CONTINUE

401 DO 414 ISUR=1,NSUR
      IF (NSUR-1) 4011,4012,4011
4011 READ TAPE NTAPE8, NSIZE, MODES2, ((HF(I,J),I=1,NSIZE),J=1,
1      MODES2)
4012 DO 403 L=1,MODES2,NC
      I=L/NC+NC-1
      IF (NSUR-1) 4018,4016,4018
4018 IF (NAKOR(I)-ISUR) 402,403,402
402 GOTO (4021,4022),NC
4021 GUESS(L,I)=H(1,I) / US(ISUR,I)

```

HMI41854
 HMI41855
 HMI41856
 HMI41857
 HMI41858
 HMI41859
 HMI41860
 HMI41861
 HMI41862
 HMI41863
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 HMI41865
 HMI41866
 HMI41867
 HMI41868
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 HMI41884
 HMI41885
 HMI41886
 HMI41887
 HMI41888
 HMI41889
 HMI41890
 HMI41891

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PART 2.....VIBRATION AND FLUTTER ANALYSIS BY A MODAL METHOD.

```

4022 GOTO 4023
4023 GUESS(L,1)=( H(1,1)*US(2*ISUR-1,1)+H(2,1)*US(2*ISUR,1) ) /
      ( US(2*ISUR-1,1)**2 + US(2*ISUR,1)**2 )
      1 GUESS(L+1,1)=( H(2,1)*US(2*ISUR-1,1) - H(1,1)*US(2*ISUR,1) ) /
      1 ( US(2*ISUR-1,1)**2 + US(2*ISUR,1)**2 )
4023 CALL NPNRMX (HF(1,L),HF(1,L),NSIZE,GUESS(L,1),-1,MAXR,NC,1 )
4016 GUESS(I,1)=H(1,I)
      GUESS(I,2)= H(2,I)
403 CONTINUE
405 WRITE OUTPUT TAPE NTAPE3, 204, ISUR, NSIZE
      CALL MPRINT (HF,NSIZE,MODES2,MAXR,NTAPE3)
      IF ( NPUNCH ) 406,414,406
406 IF ( ISUR-1 ) 408,410,408
408 WRITE OUTPUT TAPE NTAPE0, 205, NSIZE, MODES2, BCDZ, NCARDS
      NCRDS=NCARDS+1
      GOTO 412
410 NCRDS=NCARDS
412 CALL MPUNCH (HF,NSIZE,MODES2,0,1,1,BCDZ,MAXR,NTAPE0,NCRDS)
      WRITE OUTPUT TAPE NTAPE3, 206, BCDZ, NCARDS, BCDZ, NCRDS
      NCRDS=NCRDS+1
414 CONTINUE
450 CONTINUE
      IF ( NAERG ) 480,460,480
460 IF ( NRIGID ) 461,462,461
461 READ TAPE NTAPE7, ((HF(I,J),I=1,NRIGID),J=1,NFLEX)
462 READ TAPE NTAPE7, ((HFF(I,J),I=1,NFLEX),J=1,NFLEX)
      READ TAPE NTAPE7, ((US(I,J),I=1,NFLEX),J=1,MODES)
      IF ( NRIGID ) 463,467,463
463 DO 464 I=1,NRIGID
      DO 464 J=1,NFLEX
464 HF(I,J)=-HF(I,J)
      CALL DPMLTX (HF,1,US,1,HRRR ,NRIGID,NFLEX,MODES,MAXR,MAXP,MAXQ,
      1)
      CALL DPMLTX (HRF,1,US,1,HF,NRIGID,NFLEX,MODES,MAXQ,MAXP,MAXR,1)
      HML41892
      HML41893
      HML41894
      HML41895
      HML41896
      HML41897
      HML41898
      HML41899
      HML41900
      HML41901
      HML41902
      HML41903
      HML41904
      HML41905
      HML41906
      HML41907
      HML41908
      HML41909
      HML41910
      HML41911
      HML41912
      HML41913
      HML41914
      HML41915
      HML41916
      HML41917
      HML41918
      HML41919
      HML41920
      HML41921
      HML41922
      HML41923
      HML41924
      HML41925
      HML41926
      HML41927
      HML41928
      HML41929

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STORAGE NOT USED BY PROGRAM

DEC OCT
2598 05046

DEC OCT
16936 41050

STORAGE LOCATIONS FOR VARIABLES APPEARING IN COMMON STATEMENTS

AITKEN 32044 76454	DEC OCT	DEC OCT	DEC OCT
BOOLT 32473 77331		ARF 27141 65005	A 32041 76451
OMBAR 32259 77003		8REF 32411 77233	BR 32409 77231
EPOP 32047 76457		DM 32367 77157	DQ 32331 77113
H1 21941 52665		EPSP 32048 76460	FREQ 32451 77303
HF 21941 52665		HFI 32041 76451	HFF 32041 76451
H 21941 52665		HRR 17596 42274	HRR 18661 44345
MAXP 32480 77340		ISW 32541 77435	ISXST 32561 77461
MODE 32486 77346		MAXQ 32482 77342	MAXS 32481 77341
NCANT 32489 77351		NAERO 32115 76563	NAKSR 17621 42325
NFLX2 32477 77335		NCARDS 32049 76461	NDENS 32487 77347
NGO 32475 77333		NFLX 32484 77344	NFUS 32488 77350
NPOINT 32485 77345		NITER 16986 41132	NITRSP 32046 76456
NRIGIO 32491 77353		NPUNCH 32474 77332	NRHO 32114 76562
NSUR 32492 77354		NSIZES 32521 77411	NSURFS 32479 77337
NTAPE3 32499 77363		NTAPE0 32050 76462	NTAPE2 32500 77364
NTAPE7 32495 77357		NTAPE4 32498 77362	NTAPE6 32496 77360
OMEGA 17621 42325		NTAPE8 32494 77356	NVEL 32051 76463
S 32388 77204		PICON 32113 76561	SPRING 32112 76560
		US 23941 56605	VELCTY 32072 76510
		VECTOR 19621 46245	

STORAGE LOCATIONS FOR VARIABLES NOT APPEARING IN COMMON, DIMENSION, OR EQUIVALENCE STATEMENTS

8COZ 2597 05045	DEC OCT	DEC OCT	DEC OCT
IR 2593 05041		DAMP 2596 05044	INDEX 2594 05042
K1 2589 05035		I 2592 05040	J 2590 05036
MUDES 2585 05031		K 2588 05034	MODES2 2586 05032
NMODES 2581 05025		NAERO 2584 05030	NCRODS 2582 05026
		N 2580 05024	VELC 2578 05022
		N 2580 05024	
		NSIZE 2579 05023	
		FFM 2595 05043	
		ISUR 2591 05037	
		L 2587 05033	
		NCMODE 2583 05027	

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PART 2.....VIBRATION AND FLUTTER ANALYSIS BY A MODAL METHOD.

SYM8DLS AND LDCATDINS FOR SOURCE PROGRAM FDRMAT STATEMENTS

EFN	LOC	EFN	LOC	EFN	LOC
8168	200 04757	816A	202 04731	8168	203 04635
816C	204 04625	816E	206 04607	816F	207 04571
816G	208 04563				

LDCATIONS FDR OTHER SYM8DLS NOT APPEARING IN SOURCE PRDGRAM

DEC	DCT	DEC	DCT	DEC	DCT
1)	2544 04760	2)	2372 04504	3)	2381 04515
6)	2384 04520	A1102	2270 04336	A1103	2283 04353
A1107	2309 04405	A1109	2322 04422	A110A	2331 04433
A110D	2357 04465	C1G0	2548 04764	C1G1	2549 04765
C1G3	2551 04767	C1G4	2552 04770	C1101	2553 04771
C1103	2555 04773	C1105	2556 04774	C1106	2557 04775
C1109	2559 04777	C110A	2560 05000	C1108	2561 05001
C110D	2563 05003	C110E	2564 05004	C110F	2565 05005
C1201	2567 05007	C1202	2568 05010	C1203	2569 05011
C1205	2571 05013	C1206	2572 05014	C1207	2573 05015
C1209	2575 05017	C120A	2576 05020	C1208	2577 05021
D111F	415 00637	D114J	1420 02614	D114V	1547 03013
D115I	1750 03326	D122Q	725 01325	D123K	973 01715
D1252	1589 03065	D1254	1623 03127	D125J	1753 03331
D125M	1822 03436	D134T	1535 02777	D1352	1588 03064
D135L	1793 03401	D135M	1821 03435	D1401	32 00040
D144A	1276 02374	D144F	1341 02475	D144G	1355 02513
D145N	1830 03446	D147B	2251 04313	D1547	1260 02354
D175H	1744 03320	E12P	705 01301	E13G	930 01642
E13N	1043 02023	E14U	1541 03005	E153	1600 03100

LDCATIONS OF NAMES IN TRANSFER VECTOR

DEC	DCT	DEC	DCT	DEC	DCT
CHAIN	14 00016	DPMLTX	4 00004	MTERS	10 00012
MPUNCH	12 00014	NPNRMX	13 00015	SQRT	11 00013
(FPT)	0 00000	(RLR)	3 00003	(RWT)	1 00001
(ISTH)	7 00007	(TSH)	2 00002	(WLR)	6 00006
				MPRINT	9 00011
				(FIL)	8 00010
				(STB)	5 00005

ENTRY POINTS TO SUBROUTINES NOT OUTPUT FROM LIBRARY

CHAIN (FPT)	OPMLTX (RLR)	MTERS (RWT)	MPRINT (STB)	MPUNCH (STH)	NPNRMX (TSB)	SHORT (WLR)	(FIL)
EXTERNAL FORMULA NUMBERS WITH CORRESPONDING INTERNAL FORMULA NUMBERS AND SOCIAL LOCATIONS							
EFN	IFN	LOC	EFN	IFN	LOC	EFN	IFN
301	27	00066	302	30	00121	290	36
294	46	00220	305	60	00315	307	62
297	64	00352	298	66	00365	308	83
309	84	00531	310	98	00650	312	102
314	104	00706	316	115	00763	320	139
322	140	01155	324	152	01242	350	163
351	164	01332	352	172	01370	355	180
356	182	01462	358	185	01476	3584	197
359	204	01644	360	208	01661	362	212
364	214	01716	366	217	01743	369	230
371	243	02136	372	266	02356	3741	272
3742	273	02455	375	275	02476	377	282
3775	285	02642	378	287	02650	388	304
390	306	03014	392	308	03021	398	316
399	317	03135	400	319	03141	4011	322
4012	331	03250	4018	334	03323	4021	336
4022	338	03340	4023	340	03403	403	344
405	345	03456	406	350	03502	410	355
412	356	03533	414	361	03573	460	364
461	365	03620	462	373	03656	464	392
465	399	04050	466	404	04117	468	409
469	413	04221	470	415	04241	474	417
480	423	04314					

ENTRY POINTS TO SUBROUTINES REQUESTED FROM LIBRARY,	CHAIN	1 , 3	(RDS)	(RCH)	(TCO)
(TSHM)	EXIT	(IOS)	(RWT)	(STB)	(WLR)
(RTN)	(FPT)	SETH			
(TEF)	(FIL)				
(RLR)	(WRS)				
(TSB)	CHAIN				
ENTRY POINTS TO SUBROUTINES REQUESTED FROM LIBRARY,	CHAIN	2 , 3	(STHM)	(FIL)	(DFMP)
(IOS)	(WER)	(TES)	(RLR)	(STB)	(WLR)
(WRS)	(RWT)	(TSB)			
(RCH)					
DEXP(2					
(DFOP)					
DSQRT					
SQRT					

EXECUTION 16.546

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SUBROUTINE RDLN (NTAPE2, NTAPE3, I )  
1 FORMAT(80H  
1 )  
2 FORMAT(1H1)  
3 FORMAT ( 1H0 )  
  
READ INPUT TAPE NTAPE2, 1  
GOTO (4,5),1  
  
4 WRITE OUTPUT TAPE NTAPE3, 2  
GOTO 6  
  
5 WRITE OUTPUT TAPE NTAPE3, 3  
  
6 WRITE OUTPUT TAPE NTAPE3, 1  
  
RETURN  
END(1,0,0,0,0,0,0,0,0,1,0,0,0,0,0)
```

HM050001
HM050002
HM050003
HM050004
HM050005
HM050006
HM050007
HM050008
HM050009
HM050010
HM050011
HM050012
HM050013
HM050014
HM050015
HM050016
HM050017

STORAGE NOT USED BY PROGRAM

DEC OCT
79 00117

DEC OCT
32561 77461

SYMBOLS AND LOCATIONS FOR SOURCE PROGRAM FORMAT STATEMENTS

8)1	EFN	LOC	1 00115	8)2	EFN	LOC	2 00076	8)3	EFN	LOC	3 00075
-----	-----	-----	---------	-----	-----	-----	---------	-----	-----	-----	---------

LOCATIONS FOR OTHER SYMBOLS NOT APPEARING IN SOURCE PROGRAM

6)	DEC	OCT	54 00066	C)G0	DEC	OCT	78 00116	E)1	DEC	OCT	30 00036
----	-----	-----	----------	------	-----	-----	----------	-----	-----	-----	----------

LOCATIONS OF NAMES IN TRANSFER VECTOR

(FIL)	DEC	OCT	3 00003	(RTN)	DEC	OCT	1 00001	(STH)	DEC	OCT	2 00002	(TSH)	DEC	OCT	0 00000
-------	-----	-----	---------	-------	-----	-----	---------	-------	-----	-----	---------	-------	-----	-----	---------

ENTRY POINTS TO SUBROUTINES NOT OUTPUT FROM LIBRARY

(FIL) (RTN) (STH) (TSH)

EXTERNAL FORMULA NUMBERS WITH CORRESPONDING INTERNAL FORMULA NUMBERS AND SOCIAL LOCATIONS

4	EFN	IFN	LOC	8 00037	5	EFN	IFN	LOC	10 00046	6	EFN	IFN	LOC	11 00054
---	-----	-----	-----	---------	---	-----	-----	-----	----------	---	-----	-----	-----	----------

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SUBROUTINE MMULTO (A,N1,B,N2,C,M,N,K,MA,MB,MC)

OMENSION A(1), 8(1), C(1)

IC=1
IA=MC*K
IB=MA*N
ID=MA
IH=MC
IJ=MC

IF (N1) 4,3,4
3 IF (N2) 7,8,7
4 IB=2*IB

ID=2*ID
IF (N2) 5,6,5
5 IC=2

GOTO 7
6 IH=2*IH

IC=3
7 IA=2*IA

IJ=2*IJ
8 00 18 I=1,M

INC=0
00 11 J=I,IA,IH
C(J)=0.

IN=INC
00 10 L=I,IB,ID
IN=IN+1

C(J)=C(J)+A(L)*B(IN)
10 INC=0
11 INC=INC+M8

GOTO (18,12,15),IC
12 00 14 J=I,IA,IJ
IE=I+MA

IF=J+MC
IN=INC
00 13 L=IE,IB,ID

HM050561
HM050562
HM050563
HM050564
HM050565
HM050566
HM050567
HM050568
HM050569
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HM050571
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HM050614
HM050615
HM050616

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IN=IN+1
IG=IN+MB
C(IF)=C(IF)+A(L)*B(IN)
C(J)=C(J)-A(L)*B(IG)
13 INC=INC+MB *2
14 GOTO 18
15 IE=I+MC
IF=I+MA
00 17 J=IE,IA,IJ
IN=INC
C(J)=0.
DO 16 L=IF,IB,ID
IN=IN+1
C(J)=C(J)+A(L)*B(IN)
16 INC=INC+MB
17 CONTINUE
18 RETURN
END(1,0,0,0,0,0,0,0,1,0,0,0,0,0)

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STORAGE NOT USED BY PROGRAM

DEC	OCT	DEC	OCT
309 00465		32561	77461

STORAGE LOCATIONS FOR VARIABLES NOT APPEARING IN COMMON, DIMENSION, OR EQUIVALENCE STATEMENTS

DEC	OCT	DEC	OCT	DEC	OCT
IA	308 00464	IB	307 00463	IC	306 00462
IE	304 00460	IF	303 00457	IG	302 00456
IJ	300 00454	IN	299 00453	IO	298 00452
J	296 00450				

LOCATIONS FOR OTHER SYMBOLS NOT APPEARING IN SOURCE PROGRAM

DEC	OCT	DEC	OCT	DEC	OCT
2J	278 00426	3J	282 00432	6J	283 00433
C160	290 00442	C162	291 00443	C163	292 00444
C165	294 00446	C1200	295 00447	D1200	174 00256
D140M	268 00414	E11	81 00121		

EXTERNAL FORMULA NUMBERS WITH CORRESPONDING INTERNAL FORMULA NUMBERS AND OCTAL LOCATIONS

EFN	IFN	LOC	EFN	IFN	LOC
3	11	00122	4	12	00126
7	19	00155	8	21	00165
12	32	00246	13	40	00317
16	50	00377	17	51	00405
			5	15	00140
			10	28	00223
			14	41	00327
			18	52	00415
			6	17	00145
			11	29	00231
			15	43	00341

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SUBROUTINE MPRINT (A,M,N,MD,NTAPE)
  DIMENSION A(1), IT(6), C(6)
  EQUIVALENCE (IT,C)

  2 FORMAT (1H , 4X, 6( 6X, 7HCOLUMN 114 ) /// )
  3 FORMAT (1H 114, X, (6E 17.8) )

  N1=N
  N2=6
  N3=6
  N4=1
  4 IF (N3-N1) 6,6,5
  5 N2=N1-N3+6
  N3=N1
  6 K=0
  DO 7 I= N4,N3
    K=K+1
  7 IT(K)=I
  WRITEOUTPUTTAPE NTAPE, 2, (IT(I),I=1,N2)
  DO 9 I=1,M
    K=0
    L=MD*(N4-1)+I
    DO 8 J=N4,N3
      K=K+1
      C(K)=A(L)
      L=L+MD
  8
  9 WRITEOUTPUTTAPE NTAPE, 3, I, (C(K),K=1,N2)
  IF (N3-N1) 10,11,11
  10 N3=N3+6
  N4=N4+6
  GOTO 4
  11 RETURN
  END(1,0,0,0,0,0,0,0,1,0,0,0,0,0)

```

HM050019
 HM050020
 HM050021
 HM050022
 HM050023
 HM050024
 HM050025
 HM050026
 HM050027
 HM050028
 HM050029
 HM050030
 HM050031
 HM050032
 HM050033
 HM050034
 HM050035
 HM050036
 HM050037
 HM050038
 HM050039
 HM050040
 HM050041
 HM050042
 HM050043
 HM050044
 HM050045
 HM050046
 HM050047
 HM050048
 HM050049
 HM050050
 HM050051

STORAGE NOT USED BY PROGRAM

DEC OCT
192 00300
32561 77461

STORAGE LOCATIONS FOR VARIABLES APPEARING IN DIMENSION AND EQUIVALENCE STATEMENTS

DEC OCT DEC OCT DEC OCT
C 191 00277 IT 191 00277

STORAGE LOCATIONS FOR VARIABLES NOT APPEARING IN COMMON, DIMENSION, OR EQUIVALENCE STATEMENTS

DEC OCT DEC OCT DEC OCT
I 185 00271 K 184 00270 L 183 00267 N1 182 00266
N2 181 00265 N3 180 00264 N4 179 00263

SYMBOLS AND LOCATIONS FOR SOURCE PROGRAM FORMAT STATEMENTS

EFN LOC EFN LOC EFN LOC
812 2 00254 813 3 00244

LOCATIONS FOR OTHER SYMBOLS NOT APPEARING IN SOURCE PROGRAM

DEC OCT DEC OCT DEC OCT
1) 174 00256 2) 151 00227 6) 154 00232 9) 173 00255
C160 176 00260 C162 177 00261 C1202 178 00262 E1E 139 00213

LOCATIONS OF NAMES IN TRANSFER VECTOR

DEC OCT DEC OCT DEC OCT
(FIL) 1 00001 (STH) 0 00000

ENTRY POINTS TO SUBROUTINES NOT OUTPUT FROM LIBRARY

(FIL) (STH)

EXTERNAL FORMULA NUMBERS WITH CORRESPONDING INTERNAL FORMULA NUMBERS AND OCTAL LOCATIONS

EFN	IFN	LOC	EFN	IFN	LOC	EFN	IFN	LOC
4	11	00040	5	12	00045	6	14	00053
8	29	00153	9	30	00161	10	37	00214
						11	40	00223

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MREAD

```

C MATRIX READ SUBROUTINE
C CALL MREAD (A,M,N,IFDRM,IRDW,ITRA,IORG,T,MD,NTAPE2,NTAPE3 )
C
C A = MATRIX TO READ IN      ITRA = 0, TRA CARD AFTER MATRIX
C M = NUMBER OF RDWS        =+1, TRA CARD AFTER EACH RDW
C N = NUMBER OF COLUMNS    (OR COLUMN )
C IFDRM = -1, FORMAT(12A6)   IORG = DRIGIN OF FIRST C.B. CARD
C                               T = MDXN TEMPORARY CELLS
C                               = 0, COLUMN BINARY
C                               = +1, FDRMAT(6E12.8) MD= DIMENSIONED NUMBER OF ROWS
C IRDW = 0, MATRIX BY COLUMNS
C                               = +1, MATRIX BY ROWS
C                               NTAPE2 = INPUT TAPE
C                               NTAPE3 = OUTPUT TAPE
C
SUBROUTINE MREAD (A,M,N,IFORM,IRDW,ITRA,IORG,T,MD,NTAPE2,NTAPE3 )
DIMENSION A(1), T(1)
1 FORMAT (6E12.8)
2 FORMAT (12A6)
3 FORMAT ( // 26H THATS ALL YOUR DATA. )
MN=MD*N
DO 5 I=1,MN
T(I)=0.
A(I)=0.
5
IF ( IFORM ) 39,15,6
6 IF ( IROW ) 8,7,8
7 K3=1
K4=N
K5=MD
K6=M-1
K2=1
GOTO 9
8 K2=MN
K3=MD
K4=M
K5=1
K6=0

```

MREAD

```

9 DO 11 I=1,K4
  K1=I*K5-K5+1
  IF (K6) 10,11,10
10 K2=K1+K6
11 READ INPUT TAPE NTAPE2, 1, (A(L),L=K1,K2,K3,
  GO TO 36
15 K1=N
  K2=M
  K3=1
  IF ( IORG-1 ) 16,17,17
16 K3=2
17 IF ( IROW ) 18,19,18
18 K2=N
  K1=M
19 IF ( ITRA ) 22,21,22
21 K1=1
22 K=0
  DO 23 I=1,K1
    K4=K+K3
    K5=1
    CALL BINRO (T(K4), K5, L, NTAPE2, NTAPE3 )
    GO TO (23,38,23,23),L
  23 K=K+K2

24 L=0
  IF ( IROW ) 28,24,28
  IF ( IORG-1) 26,26,25
25 L=IORG-1
26 DO 27 I=1,N
  J=I*MD-MD
  DO 27 K=1,M
    J=J+1
  L=L+1
27 A(J)=T(L)
  GO TO 36

28 L=0
  IF ( IORG-1) 30,30,29

```

HM050091
HM050092
HM050093
HM050094
HM050095
HM050096
HM050097
HM050098
HM050099
HM050100
HM050101
HM050102
HM050103
HM050104
HM050105
HM050106
HM050107
HM050108
HM050109
HM050110
HM050111
HM050112
HM050113
HM050114
HM050115
HM050116
HM050117
HM050118
HM050119
HM050120
HM050121
HM050122
HM050123
HM050124
HM050125
HM050126
HM050127
HM050128

MR. READ

```
29 I=IORG-1
30 00 31 K=1,N
      J=K*MD-MD
      DO 31 I=K,MN,N
```

```

31      J=J+1
      K1=L+I
      A(J)=T(K1)

```

36 RETURN

```
38 WRITE OUTPUT TAPE NTAPE3, 3
CALL EXIT
```

```

39 READ INPUT TAPE NTAPE2, 2, (A(I), I=1,M)
      STOP
      GOTO 36
END(1,0,0,0,0,0,0,0,1,0,0,0,0,0)

```

HM050129
HM050130
HM050131
HM050132
HM050133
HM050134
HM050135
HM050136
HM050137
HM050138
HM050139
HM050140
HM050141
HM050142
HM050143

MREAD

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STORAGE NOT USED BY PROGRAM

DEC	OCT
406 00626	32561 77461

STORAGE LOCATIONS FOR VARIABLES NOT APPEARING IN COMMON, DIMENSION, OR EQUIVALENCE STATEMENTS

DEC	OCT	DEC	OCT	DEC	OCT
I 405 00625	J 404 00624	K1 403 00623	K2 402 00622		
K3 401 00621	K4 400 00620	K5 399 00617	K6 398 00616		
K 397 00615	L 396 00614	MN 395 00613			

SYMBOLS AND LOCATIONS FOR SOURCE PROGRAM FORMAT STATEMENTS

EFN	LOC	EFN	LOC	EFN	LOC
811 1 00601	812 2 00577	813 3 00575			

LOCATIONS FOR OTHER SYMBOLS NOT APPEARING IN SOURCE PROGRAM

DEC	OCT	DEC	OCT	DEC	OCT
2) 363 00553	3) 366 00556	6) 367 00557	9) 386 00602		
C1G1 388 00604	C1G2 389 00605	C1G3 390 00606	C1G4 391 00607		
C1200 392 00610	C1203 393 00611	C1205 394 00612	D1107 128 00200		
D1212 301 00455	D1216 337 00521	E1E 170 00252	E1H 181 00265		
E1J 191 00277	E1N 226 00342				

LOCATIONS OF NAMES IN TRANSFER VECTOR

DEC	OCT	DEC	OCT	DEC	OCT
B1NR0 2 00002	EXIT 5 00005	(FIL) 4 00004	(RTN) 1 00001		
(STH) 3 00003	(TSH) 0 00000				

ENTRY POINTS TO SUBROUTINES NOT OUTPUT FROM LIBRARY

B1NRD	EXIT	(FIL)	(RTN)	(STH)	(TSH)
-------	------	-------	-------	-------	-------

MREAD

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EXTERNAL FORMULA NUMBERS WITH CORRESPONDING INTERNAL FORMULA NUMBERS AND OCTAL LOCATIONS

EFN	IFN	LOC	EFN	IFN	LOC	EFN	IFN	LOC	EFN	IFN	LOC
5	10	00124	6	12	00134	7	13	00140	8	19	00156
9	24	00172	10	27	00213	11	28	00216	15	34	00237
16	38	00253	17	39	00255	18	40	0027	19	42	00266
21	43	00272	22	44	00300	23	51	00335	24	53	00347
25	55	00360	26	56	00365	27	61	00417	28	63	00431
29	65	00442	30	66	00447	31	71	00506	36	72	00522
38	74	00526	39	77	00537						

```

* FIB II FEBUARY 10,1961 ROGER ANDERSON HM050148
* FIB II APRIL 20,1961 REVISED ANDCORRECTED BY R. L. GAUTHIER HM050149
*--*****HM050150
* CALLING SEQUENCE
* TSX BINRD,4 HM050151
* TSX L(ARRAY) HM050152
* TSX L(K1), WHERE K1 IS READ CONTROL. HM050153
* TSX L(K2), WHERE K2 IS ERROR CONTROL. HM050154
* TSX L(OUTPUT TAPE NUMBER) HM050155
* TSX L(OUTPUT TAPE NUMBER) HM050156
* TSX L(OUTPUT TAPE NUMBER) HM050157
* K1 = ZERO, READ ONE CARD. HM050158
* K1 = NONZERO, READ TO TRANSFER CARD. HM050159
* IF K2 = 1, RECORD(S) READ CORRECTLY. HM050160
* K2 = 2, END OF FILE ENCOUNTERED. READ END. HM050161
* K2 = 3, CHECKSUM ERROR OR TAPE CHECK. HM050162
* K2 = 4, (WHEN K1=0) NEXT RECORD IS BCD. HM050163
*--*****HM050164
*--*****HM050165
*--*****HM050166
*--*****HM050167
*--*****HM050168
*--*****HM050169

```

00011

ENTRY BINRD

TRANSFER VECTOR

```

00000 743146623460 (IOS)
00001 745124623460 (RDS)
00002 745123303460 (RCH)
00003 746323463460 (TCO)
00004 746351233460 (TRC)
00005 746325263460 (TEF)
00006 742262513460 (BSR)
00007 256731636060 EXIT
00010 746651623460 (WRS)

```

```

00011 0634 00 4 00156 BINRD SXA BRDXR4,4 HM050170
00012 0634 00 2 00157 SXA BRDXR2,2 HM050171
00013 0634 00 1 00160 SXA BRDXR1,1 HM050172
00014 0500 60 4 00034 CLA* 4,4 HM050173

```

SAVE INDEX REGISTERS.

```

00015 0622 00 0 00252          STD      BRD85D          HM050174
00016 0500 60 4 00005          CLA*      5,4              HM050175
00017 0622 00 0 00251          STD      BRD6D              HM050176
00020 -0500 00 0 00245          CAL      BRD1D              HM050177
00021 0602 60 4 00003          SLW*      3,4              HM050178
00022 -0500 00 0 00245          CAL      BRD1D              HM050179
00023 0602 00 0 00243          SLW      DNE              HM050180
00024 -0500 00 4 00001          CAL      1,4              HM050181
00025 0400 00 0 00244          ADD      BRD1A              HM050182
00026 0621 00 0 00100          STA      BRDARY          HM050183
*****
***** READ INPUT TAPE IN BINARY MDDE.          HM050184
***** IF REDUNDANCY OCCURS, (1) THE RECDRD IS IN BCD MDDE AND ROUTINE          HM050185
* WILL INDICATE K2 = 4, IF SINGLE READ DR K2 = 2 IF MULTI          HM050186
* READ. (2) A LEGITIMATE REDUNDANCY HAS OCCURRED ON A          HM050187
* BINARY RECDRD, A SECDND READ IS ATTEMPTED, IF REDUNDANCY          HM050188
* PERSISTS A MESSAGE IS WRITTEN ON THE OUTPUT TAPE          HM050189
* AND K2 IS SET TO 3.....          HM050190
*****          HM050191
*****          HM050192
*          HM050193
BRDRD CAL BRDRD5D          ESTABLISH BINARY I-D          HM050194
BRDRD1 XEC* $(RDS)          FOR INPUT TAPE.          HM050195
AXC BRDEND,4          READ SELECT INPUT.          HM050196
XEC* $(RCH)          RESET AND LOAD CHANNEL          HM050197
AXC **1,4          DELAY          HM050198
XEC* $(TCD)          HM050199
AXC BRDTPC,4          HM050200
XEC* $(TRC)          HM050201
CAL BRD1D          TEST REDUNDANCY.          HM050202
SLW DNE          HM050203
BRDTEF XEC* BRDEND,4          HM050204
XEC* $(TEF)          HM050205
*****          HM050206
*****          HM050207
***** TEST FOR TRANSFER CARD.          HM050208
* IF TRANSFER CARD IS ENCOUNTERED, TERMINATE READ.          HM050209
*****          HM050210
*****          HM050211

```

```

00044 -0500 00 0 77740 BRDTCT CAL BUFFER IF DECREMENT (WORD COUNT) HM050212
00045 -0320 00 0 00250 ANA BRD037 IS ZERO, THIS IS HM050213
00046 0100 00 0 00152 TZE BROTCO A TRANSFER CARD..... HM050214
***** TEST TO IGNORE CHECKSUM. HM050215
00047 -0500 00 0 77740 CAL BUFFER IF COL 1, ROW G HM050216
00050 -0320 00 0 00254 ANA BROIGN IS PUNCHED, IGNORE HM050217
00051 -0100 00 0 00071 TNZ BROWM THE CHECKSUM. HM050218
* HM050219
***** HM050220
***** ACCUMULATE AND CHECK THE CHECKSUM. ***** HM050221
* IF CHECKSUM ERROR, WRITE MESSAGE, INDICATE HM050222
* ERROR WITH 3 AT K2, AND CONTINUE. HM050223
* (THE CARD WILL BE MOVED INTO THE ARRAY.) HM050224
***** HM050225
* HM050226
00052 -0500 00 0 77740 CAL BUFFER SET UP NO. OF HM050227
00053 -0320 00 0 00250 ANA BRD037 WORDS FOR HM050228
00054 -0734 00 4 00000 PDX 0,4 HM050229
00055 0634 00 4 00063 SXA BINWDS,4 COMPUTING CHECKSUM HM050230
00056 1 00002 4 00057 TXI **1,4,2 HM050231
00057 -0634 00 4 00061 SXO **2,4 HM050232
00060 0774 00 4 77740 AXT BUFFER,4 HM050233
00061 1 00000 4 00065 TXI **1,4,** HM050234
00062 0634 00 4 00065 SXA BINCK,4 HM050235
00063 0774 00 4 00000 BINWDS AXT **4 ACCUMULATE ALL WORDS HM050236
00064 -0500 00 0 77740 CAL BUFFER EXCEPT CHECKSUM WORD. HM050237
00065 0361 00 4 00000 BINCK ACL **4 HM050238
00066 2 00001 4 00065 TIX **1,4,1 HM050239
00067 0322 00 0 77741 ERA BUFFER+1 HM050240
00070 -0100 00 0 00144 TNZ BRD0SE CHECKSUM ERROR HM050241
* HM050242
***** HM050243
***** MOVE THE CARD IMAGE INTO USERS ARRAY. ***** HM050244
* THE ARRAY IS FILLED IN FORTRAN FASHION. (BACKWARDS) HM050245
* THE INDEX NUMBER LOCATES THE FIRST WORD IN ARRAY. HM050246
* THE WORD COUNT IS USED TO TERMINATE THE MOVE. HM050247
* BY DEFN...IF THE INDEX NUMBER = 1, PLACE FIRST WORD AT L(ARRAY)..HM050248
*-----HM050249

```



```

00071 -0500 00 0 77740 * BRDMV CAL BUFFER
00072 0734 00 4 00000 PAX 0,4
00073 -0320 00 0 00250 ANA BRDD37
00074 -0737 00 2 00000 PDC 0,2
00075 -0634 00 2 00103 SXD BRDMV1,2
00076 0774 00 1 00000 AXT 0,1
00077 -0500 00 1 77742 CAL BUFFER+2,1
00100 0602 00 4 00000 BRDARY SLW **,4
00101 1 77777 1 00102 TXI **1,1,-1
00102 1 00001 4 00103 TXI **1,4,1
00103 3 00000 1 00077 BRDMV1 TXH **4,1,**

*****
***** TEST FOR MULTI OR SINGLE READ AND BRANCH ACCORDINGLY.
*****
*
00104 0534 00 4 00156 LXA BRDXR4,4
00105 0520 60 4 00002 ZET* 2,4
00106 0020 00 0 00031 TRA BRDRD1
00107 0020 00 0 00156 TRA BRDXR4

*
*****
*****Q*****
HM050250
HM050251
HM050252
HM050253
HM050254
HM050255
HM050256
HM050257
HM050258
HM050259
HM050260
HM050261
HM050262
HM050263
HM050264
HM050265
HM050266
HM050267
HM050268
HM050269
HM050270
HM050271
HM050272
HM050273
HM050274
HM050275
HM050276
HM050277
HM050278
HM050279
HM050280
HM050281
HM050282
HM050283
HM050284
HM050285
HM050286
HM050287

INDEX ND. TD XR4.
WDRD COUNT TO EXIT
TEST.
**=(ARRAY+1)
**=(WORD COUNT)

POSITION FOR RETRY OR EXIT.
IF A BCD RECORD IS
ENCOUNTERED, THE C(AR)
OF CHANNEL WILL
DENOTE ONLY 14 WORDS
TRANSMITTED. A BINARY
RECORD WILL BE DENOTED
BY A GREATER NUMBER OF
WORDS TRANSMITTED.
(THE SCHX INSTRUCTION IS ABSENT
FROM IOS. MUST BUILD ONE.)

$ (BSR)
BRDCSV,4
26624,2
$(RCH)
**2
**1,2
BRDCSV
BRD5A7
BUFORG
0,4
BRDBTC,4,17

```


BINRD ROUTINE TO READ COL BIN CARDS FROM INPUT TAPE. FIBII

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```

00166 -0625 00 0 00164
00167 -0120 00 0 00007
00170 0534 00 4 00156
00171 0602 60 4 00003
00172 0020 00 0 00157

STL *-2
TMI $EXIT
LXA BRDXR4,4
SLW* 3,4
TRA BRDXR2

***** INDEX REGISTER 1 DENOTES ERROR TYPE.
***** XRI=0, CHECKSUM ERROR
***** XRI=1, REDUNDANCY
*****

00173 0634 00 4 00203
00174 -0500 00 0 00251
00175 0074 00 4 00000
00176 0522 60 0 00010
00177 -3 00000 1 00205
00200 -0774 00 4 00260
00201 0522 60 0 00002
00202 0074 00 2 00221
00203 0774 00 4 00000
00204 0020 00 4 00001

BRDRPT SXA RPTXR4,4
CAL BRD60
CALL (IOS)
XEC* $(WRS)
TXL RPTCSE,1,0
AXC REDCMD,4
XEC* $(RCH)
TSX RPTERR,2
RPTXR4 AXT **,4
TRA 1,4

***** REPORT THE CONTROL WORD OF CARD WITH BAD CHECKSUM.
*****

00205 0774 00 2 00002
00206 0560 00 0 77740
00207 0774 00 1 00006
00210 0767 00 0 00003
00211 -0763 00 0 00003
00212 2 00001 1 00210
00213 0602 00 2 00274
00214 2 00001 2 00207
00215 -0774 00 4 00261
00216 0522 60 0 00002
00217 0074 00 2 00221
00220 0020 00 0 00203

RPTCSE AXT 2,2
LDQ BUFFER
RPTCH AXT 6,1
RPTEDT ALS 3
LGL 3
TIX RPTEDT,1,1
SLW BRDCHK+2,2
TIX RPTCH,2,1
AXC CSECM,4
XEC* $(RCH)
TSX RPTERR,2
TRA RPTXR4

EXIT WHEN EOF IS
ENCOUNTERED AGAIN....
....
....
....

HM050326
HM050327
HM050328
HM050329
HM050330
HM050331
HM050332
HM050333
HM050334
HM050335
HM050336
HM050337
HM050338
HM050339
HM050340
HM050341
HM050342
HM050343
HM050344
HM050345
HM050346
HM050347
HM050348
HM050349
HM050350
HM050351
HM050352
HM050353
HM050354
HM050355
HM050356
HM050357
HM050358
HM050359
HM050360
HM050361
HM050362
HM050363

EDIT THE CONTROL
WORD OF BAD CARD,
PLACE CONTROL WORD
IN MESSAGE.

CHECK WRITE ERROR.

```

BINRD ROUTINE TO READ COL BIN CARDS FROM INPUT TAPE. FIBII

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00221	0634	00	4	00240	* RPTERR	SXA	BRDCOM,4	HM050364
00222	-0774	00	4	00223	AXC	AXC	REMEMBER LAST COMMAND.	HM050365
00223	0522	60	0	00003	XEC*	XEC*	DELAY AND TEST	HM050366
00224	-0774	00	4	00231	AXC	IPTPCH,4	REDUNDANCY ON OUTPUT.	HM050367
00225	0522	60	0	00004	XEC*	XEC*		HM050368
00226	-0500	00	0	00252	CAL	BRDB50		HM050369
00227	0074	00	4	00000	CALL	(IOS)	REESTABLISH IO FOR	HM050370
00230	0020	00	2	00001	TRA	1,2	INPUT TAPE....	HM050371
00231	0522	60	0	00006	IPTPCH	XEC*	BACKSPACE OUTPUT	HM050372
00232	0522	60	0	00010	XEC*	XEC*	TAPE AND ERASE.	HM050373
00233	-0774	00	4	00234	AXC	AXC		HM050374
00234	0522	60	0	00003	XEC*	XEC*		HM050375
00235	-0774	00	4	00237	AXC	AXC	TURN OFF POSSIBLE	HM050376
00236	0522	60	0	00004	XEC*	XEC*	REDUNDANCY.	HM050377
00237	0522	60	0	00010	XEC*	XEC*		HM050378
00240	0774	00	4	00000	BRDCOM	AXT	RETRY WRITE.	HM050379
00241	0522	60	0	00002	XEC*	XEC*		HM050380
00242	0020	00	0	00221	TRA	RPTERR		HM050381
00243	0	00001	0	00000	* ONE	PZE		HM050382
00244	0	00000	0	00001	BRD1A	PZE	0,0,1	HM050383
00245	0	00001	0	00000	BRD1D	PZE	1	HM050384
00246	0	00002	0	00000	BRD2D	PZE	0,0,1	HM050385
00247	0	00003	0	00000	BRD3D	PZE	0,0,2	HM050386
00250	0	00037	0	00000	BRD37	PZE	0,0,3	HM050387
00251	0	00003	0	00000	BRD6D	PZE	0,0,31	HM050388
00252	0	00002	0	00020	BRD85D	PZE	0,0,3	HM050389
00253	+00000007777				BRD5A7	OCT	16,0,2	HM050390
00254	1	00000	0	00000	BRDIGN	PON	7777	HM050391
00255	0	00000	0	77740	BRFORG	PZE	0,0,0	HM050392
00256	0	00000	0	00000	BRDCSV	PZE	0,0,0	HM050393
00257	-3	00033	0	77740	* BRDCMD	IOST	0,0,0	HM050394
00260	0	00005	0	00262	REDCMD	IOCD	0,0,0	HM050395
00261	0	00006	0	00267	CSECD	IOCD	0,0,0	HM050396
					* MESSAGE		0,0,0	HM050397
							0,0,0	HM050398
							0,0,0	HM050399
							0,0,0	HM050400
							0,0,0	HM050401

BINRO ROUTINE TO READ COL BIN CARDS FROM INPUT TAPE. FIBII
POST PROCESSOR ASSEMBLY DATA

275 IS THE FIRST LOCATION NOT USED BY THIS PROGRAM

REFERENCES TO DEFINED SYMBOLS

243	ONE	23, 41, 131, 133, 134
7	EXIT	167
65	BINCK	62
11	BINRD	
244	BRD1A	25
245	BRD1D	20, 22, 40, 126
246	BRD2D	162
247	BRD3D	124, 137, 146
251	BRD6D	17, 174
71	BROMV	51, 151
27	BRDRD	
6	(BSR)	110, 163, 231
0	(IOS)	30, 175, 227
2	(RCH)	33, 113, 201, 216, 241
1	(RDS)	31
3	(TCO)	35, 223, 234
5	(TEF)	43
4	(TRC)	37, 225, 236
10	(WRS)	176, 232, 237
63	BINWDS	55
253	BRD5A7	120
100	BRDARY	26
252	BRD85D	15, 27, 226
131	BRDBTC	123
272	BRDCHK	213
257	BRDCMO	32
240	BRDCDM	221
144	BRDCSE	70
256	BRDCSV	111, 117
250	BRDD37	45, 53, 73
162	BRDEND	42
254	BRDIGN	50
262	BRDMG1	260

BINRD ROUTINE TO READ COL BIN CARDS FROM INPUT TAPE. FIBII
POST PROCESSOR ASSEMBLY DATA

267 BRDMG2	261	
103 BRDMV1	75	
31 BRDRDI	106,	135
173 BRDRPT	142,	150
152 BKDTCD	46	
44 BRDTCT		
42 BRDTFF	143	
110 BRDTPC	36	
160 BRDXR1	13	
157 BRDXR2	12,	130, 172
156 BRDXR4	11,	104, 107, 125, 136, 145, 152, 170
77740 BUFFER	44,	47, 52, 60, 64, 67, 71, 77, 153, 206, 255, 257, 275
255 BUFORG	121	
261 CSECMD	215	
231 IPTPCH	224	
260 REDCMD	200	
205 RPTCSE	177	
207 RPTTECH	214	
210 RPTEDI	212	
221 RPTERR	202,	217, 242
203 RPTXR4	173,	220

NO ERROR IN ABOVE ASSEMBLY.

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HM050408
HM050409
HM050410
HM050411
HM050412
HM050413
HM050414
HM050415
HM050416
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HM050444
HM050445

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SUBROUTINE MNVRX (AR,AI,B,C,KSZ,IGOOD,NOP)
  DIMENSION AR(6,6), AI(6,6), B(6,6), C(6,6)
  IGOOD=0
  IF (NOP) 102,101,102
  101 CALL INVERS (AR,KSZ,IGOOD)
  GO TO 20
  102 CONTINUE
  DO 1 K=1,KSZ
  DO 1 L=1,KSZ
  1 B(K,L)=AR(K,L)
  NO=0
  CALL INVERS(B,KSZ,NO)
  IF (NO) 2,3,2
  C REAL MATRIX NOT SINGULAR
  C MULT B*AI STO. C
  3 DO 4 K=1,KSZ
  DO 4 L=1,KSZ
  C(K,L)=0.0
  DO 4 I=1,KSZ
  4 C(K,L)=C(K,L)+B(K,I)*AI(I,L)
  C MULT AI*C + AR STO. B
  DO 5 K=1,KSZ
  DO 5 L=1,KSZ
  B(K,L)=AR(K,L)
  DO 5 I=1,KSZ
  5 B(K,L)=B(K,L)+AI(K,I)*C(I,L)
  NO=0
  CALL INVERS(B,KSZ,NO)
  IF (NO) 2,7,2
  C SECOND MATRIX NOT SINGULAR
  C MULT -C*B STO. AI ALSO SET AR=B
  7 DO 8 K=1,KSZ
  DO 8 L=1,KSZ
  AI(K,L)=0.0
  AR(K,L)=B(K,L)
  DO 8 I=1,KSZ

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HM050466
HM050467
HM050468
HM050469
HM050470
HM050471
HM050472
HM050473
HM050474
HM050475
HM050476
HM050477
HM050478
HM050479
HM050480

STORAGE NOT USED BY PROGRAM

DEC OCT
516 01004
32561 77461

STORAGE LOCATIONS FOR VARIABLES NOT APPEARING IN COMMON, DIMENSION, OR EQUIVALENCE STATEMENTS

DEC OCT DEC OCT DEC OCT
NC 515 01003

LOCATIONS FOR OTHER SYMBOLS NOT APPEARING IN SOURCE PROGRAM

DEC OCT DEC OCT DEC OCT
2) 501 00765 3) 504 00770 6) 505 00771 9) 511 00777
C1100 513 01001 C1101 514 01002 D120E 230 00346 D140L 286 00436
D1411 364 00554 D141E 465 00721

LOCATIONS OF NAMES IN TRANSFER VECTOR

DEC OCT DEC OCT DEC OCT
INVERS 0 00000

ENTRY POINTS TO SUBROUTINES NOT OUTPUT FROM LIBRARY

INVERS

EXTERNAL FORMULA NUMBERS WITH CORRESPONDING INTERNAL FORMULA NUMBERS AND OCTAL LOCATIONS

EFN	IFN	LOC	EFN	IFN	LOC	EFN	IFN	LOC
101	6	00204	102	9	00211	1	12	00224
4	21	00276	5	26	00354	7	31	00410
2	38	00472	9	40	00505	11	45	00526
13	54	00637	15	59	00673	16	64	00734
20	67	00761				17	66	00757

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PAGE 1

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SUBROUTINE INVERS (A,N,I GOOF O)
DIMENSION A(6,6), L(6), M(6)
IF ACCUMULATOR OVERFLOW 500,500
IF QUOTIENT OVERFLOW 501,501
IF DIVIDE CHECK 502,502
IGOOD=0
SEARCH FOR LARGEST ELEMENT
DO 80 K=1,N
  L(K)=K
  M(K)=K
  BIGA=A(K,K)
  DO 20 I=K,N
    DO 20 J=K,N
      IF(ABSF(BIGA)-ABSF(A(I,J)))10,20,20
10    BIGA=A(I,J)
      L(K)=I
      M(K)=J
20    CONTINUE
  INTERCHANGE ROWS
  JROW=L(K)
  IF(L(K)-K)35,35,25
  DO 30 I=1,N
    HOLD=-A(K,I)
    A(K,I)=A(JROW,I)
    A(JROW,I)=HOLD
  INTERCHANGE COLUMNS
  ICOL=M(K)
  IF(M(K)-K)45,45,37
  DO 40 J=1,N
    HOLD=-A(J,K)
    A(J,K)=A(J,ICOL)
    A(J,ICOL)=HOLD
  DIVIDE COLUMN BY MINUS PIVOT
40  DO 55 IC=1,N
45  IF(IC-K)50,55,50
46  A(IC,K)=A(IC,K)/(-A(K,K))
50  CONTINUE
55  REDUCE MATRIX
C

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HM050483
 HM050484
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 HM050500
 HM050501
 HM050502
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 HM050520

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56      DO 65 I=1,N
57      DO 65 J=1,N
58      IF(I-K)57,65,57
59      IF(J-K)60,65,6C
60      A(I,J)=A(I,K)*A(K,J)+A(I,J)
61      CONTINUE
62      DIVIDE ROW BY PIVOT
63      DO 75 JR=1,N
64      IF(JR-K)70,75,70
65      A(K,JR)=A(K,JR)/A(K,K)
66      CONTINUE
67      CONTINUED PRODUCT OF PIVOTS
68      REPLACE PIVOT BY RECIPROCAL
69      A(K,K)=1./A(K,K)
70      CONTINUE COMPLETE OPERATION
71      CONTINUE
72      80
73      IF DIVIDE CHECK510,503
74      IF ACCUMULATOR OVERFLOW 510,504
75      IF QUOTIENT OVERFLOW 510,505
76      FINAL ROW AND COLUMN INTERCHANGE
77      K=N
78      K=(K-1)
79      IF(K)150,150,103
80      I=L(K)
81      IF(I-K)120,120,105
82      DO 110 J=1,N
83      HOLD=A(J,K)
84      A(J,K)=-A(J,I)
85      A(J,I)=HOLD
86      J=M(K)
87      IF(J-K)100,100,125
88      DO 130 I=1,N
89      HOLD=A(K,I)
90      A(K,I)=-A(J,I)
91      A(J,I)=HOLD
92      GO TO 100
93      RETURN
94      510 I GOOF D=1

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HM050521
HM050522
HM050523
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HM050558

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STORAGE NOT USED BY PROGRAM

DEC OCT
516 01004
32561 77461

STORAGE LOCATIONS FOR VARIABLES APPEARING IN DIMENSION AND EQUIVALENCE STATEMENTS

DEC OCT
L 515 01003
M 509 00775

STORAGE LOCATIONS FOR VARIABLES NOT APPEARING IN COMMON, DIMENSION, OR EQUIVALENCE STATEMENTS

DEC OCT
BICA 503 00767
I 499 00763
K 495 00757
HOLD 502 00766
JROW 498 00762
ICOL 501 00765
JR 497 00761
IC 500 00764
J 496 00760

LOCATIONS FOR OTHER SYMBOLS NOT APPEARING IN SOURCE PROGRAM

DEC OCT
1) 475 00733
6) 468 00724
C1101 481 00741
C1107 485 00745
C1200 489 00751
C1206 493 00755
D110J 246 00366
D111B 426 00652
D120U 322 00502
D131D 443 00673
D1719 413 00635
E117 381 00575
2) 464 00720
9) 474 00732
C1102 482 00742
C1108 486 00746
C1202 490 00752
C1207 494 00756
D110P 282 00432
D111D 444 00674
D130J 245 00365
D140G 224 00340
D171D 442 00672
3) 467 00723
C1100 479 00737
C1104 483 00743
C1109 487 00747
C1203 491 00753
D110C 194 00302
D110R 292 00444
D111F 456 00710
D130U 321 00501
D140I 237 00355
E1A 185 00271
4) 32767 77777
C1100 480 00740
C1106 484 00744
C110A 488 00750
C1205 492 00754
D110E 207 00317
D1119 415 00637
D120N 273 00421
D1319 414 00636
D170J 244 00364
E1T 311 00467

EXTERNAL FORMULA NUMBERS WITH CORRESPONDING INTERNAL FORMULA NUMBERS AND OCTAL LOCATIONS

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EFN	IFN	LOC	EFN	IFN	LOC	EFN	IFN	LOC	EFN	IFN	LOC
500	6	00106	501	8	00111	502	9	00113	10	17	00250
20	20	00256	25	23	00301	30	26	00307	35	27	00320
37	29	00327	40	32	00345	45	33	00356	46	34	00367
50	35	00372	55	36	00377	56	39	00427	57	40	00433
60	41	00436	65	42	00445	68	44	00503	70	45	00506
75	46	00511	80	48	00521	80	50	00554	504	52	00557
505	54	00562	100	55	00565	103	57	00576	105	59	00605
110	62	00644	120	63	00653	125	65	00662	130	68	00701
150	70	00711	510	72	00715						

```

*****HM051203
* CALLING SEQUENCE*****HM051204
* TSX BINPU,4*****HM051205
* TSX LOC (ARRAY TO BE PUNCHED)*****HM051206
* TSX LOC (NO. WORDS TO PUNCH)*****HM051207
* TSX LOC (CARD ORIGIN FOR 1ST CARD)*****HM051208
* *TSX LOC (SEQ NO. OF 1ST CARD)*****HM051209
* *TSX LOC (BCD ID FOR THIS DECK, 1ST AND 2ND CHARACTER BLANKS)*****HM051210
* TSX LOC (OUTPUT TAPE NUMBER)*****HM051211
* *****CONTRARY TO BELOW, NO ITEMS MAY BE OMITTED IN THIS MODIFICATION.*****HM051212
* *****HM051213
* *****HM051214
* *****HM051215
* *****HM051216
* *****HM051217
* *****HM051218
* *****HM051219
* *****HM051220
* *****HM051221
* *****HM051222
* *****HM051223

```

00006

TRANSFER VECTOR

```

00000 743146623460 (IDS)
00001 746651623460 (WRS)
00002 745123303460 (RCH)
00003 746663233460 (WTC)
00004 746625213460 (WER)
00005 746325623460 (TES)

```

```

00006 0634 00 1 00142 BINPU SXA
00007 0634 00 2 00143 SXA
00010 -0500 60 4 00006 CAL*
00011 0622 00 0 00331 STD
00012 0500 00 4 00001 CLA
00013 0621 00 0 00062 STA
00014 -0500 60 4 00002 CAL*
00015 0602 00 0 77776 SLW

```

```

SXA X1,1
SXA X2,2
CAL* 6,4
STD 14D
CLA 1,4
STA ARRAY
CAL* 2,4
SLW END

```

LOC OF ARRAY

```

WORD COUNT
END=0 IF TRANSFER CARD

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HM051224
HM051225
HM051226
HM051227
HM051228
HM051229
HM051230
HM051231

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BINPU ROUTINE TO WRITE COL BIN CARDS ON TAPE. FIBII

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00016	C402	00	0	00325	SUB	D1	HM051232
00017	0622	00	0	00066	STD	LOCN	HM051233
00020	0634	00	0	00061	SXA	COUNT,0	HM051234
00021	0500	60	4	00003	CLA*	3,4	HM051235
00022	0771	00	0	00022	ARS	18	HM051236
00023	-0120	00	0	00025	TMI	**2	HM051237
00024	-0501	00	0	00266	ORA	REL	HM051238
00025	-0501	00	0	00334	ORA	IMAGE	HM051239
00026	0602	00	0	07740	SLW	CIMAGE	HM051240

* TEST FOR FOURTH AND-OR FIFTH ARGUMENTS.							
* DETERMINE WHETHER ARGUMENT REFERS TO ID OR SEQ NUMBER							
* AND SET CELLS FROM CALLING SEQUENCE.							

00027	0774	00	2	00002	AXT	2,2	HM051241
00030	-0625	00	0	00302	STL	BLSEQ	HM051242
00031	-0500	00	4	00004	CAL	4,4	HM051247
00032	-0320	00	0	00265	ANA	MSKPD	HM051248
00033	0322	00	0	00307	ERA	MSKTSX	HM051249
00034	-0100	00	0	00054	TNZ	G2	HM051250
00035	0500	60	4	00004	CLA*	4,4	HM051251
00036	-0340	00	0	00262	LAS	BCI8	HM051252
00037	0020	00	0	00051	TRA	G3	HM051253
00040	0600	00	0	00302	STZ	BLSEQ	HM051254
00041	-0100	00	0	00043	TNZ	**2	HM051255
00042	-0754	00	0	00000	PXD		HM051256
00043	-0130	00	0	00000	XCL		HM051257
00044	0634	00	4	00046	SXA	**2,4	HM051258
00045	0074	00	4	00172	TSX	COSEQ,4	HM051259
00046	0774	00	4	00000	AXT	**4	HM051260
00047	0602	00	0	00267	SLW	SEQNO	HM051261
00050	1 7777	4	00053	TXI		G5,4,-1	HM051262
00051	0601	00	0	00305	STO	BCDID	HM051263
00052	1 7777	4	00053	TXI		G5,4,-1	HM051264
00053	2 0001	2	00031	TIX		G4,2,1	HM051265
00054	0634	00	4	00144	SXA	X4,4	HM051266
00055	-0520	00	0	07776	NZT	END	HM051267
00056	0020	00	0	00152	TRA	TRCD	HM051268

SET BLSEQ TO ITS NORMAL STATE							
TEST FOR 4TH, 5TH ARGS							
NO MORE TSXES							
BIG, THIS IS ID							
EQUAL, FLAG BLANK SEQ. NO.							
IS SEQ NO NON-ZERO.							
NO							
SMALL, THIS IS SEQ NO.							
CONVERT SEQ NO TO BCO							
SAVE							
MOVE TO NEXT ARGUMENT							
AT MOST 2 EXTRA ARGS.							
IS WORD COUNT ZERO							
MUST BE A TRANSFER CARD							

1

BINPU ROUTINE TO WRITE COL BIN CARDS ON TAPE. FIBII

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00113	0602 00 2 77734	SLW	LAST+4,2	COL BIN AT LAST TO LAST+3	HM051308
00114	1 77777 2 00115	TXI	**1,2,-1		HM051309
00115	2 00001 4 00105	TIX	ABC,4,1		HM051310
00116	0560 00 0 00326	LDQ	IDLCD	FINISH W/SAVED C(MQ).	HM051311
00117	3 00000 2 00104	TXH	ABC-1,2,0		HM051312
00120	0774 00 1 00000	SVI	**1		HM051313
		AXT			HM051314
					HM051315
					HM051316
					HM051317
					HM051318
					HM051319
					HM051320
00121	0761 00 0 00000	WRITE NOP			HM051321
00122	-0500 00 0 00331	WRITE1 CAL	14D	ESTABLISH I/O FOR TAPE 14.	HM051322
00123	0074 00 4 00000	CALL	(IOS)		HM051323
00124	0522 60 0 00001	XEC*	\$(WRS)		HM051324
00125	-0774 00 4 00213	AXC	PUNCMD,4		HM051325
00126	0522 60 0 00002	XEC*	\$(RCH)		HM051326
00127	0754 00 4 00000	PXA	0,4	SET (WER) FOR RETRY.	HM051327
00130	0621 60 0 00003	STA*	\$(WTC)		HM051328
00131	0074 00 4 00004	TSX	\$(WER),4		HM051329
00132	-0500 00 0 00267	CAL	SEQNO	INCREMENT CARD COUNT.	HM051330
00133	0400 00 0 00327	ADD	L(1)		HM051331
00134	0114 06 0 00215	CVR	TBI,6		HM051332
00135	0602 00 0 00267	SLW	SEQNO		HM051333
00136	0520 00 0 77776	ZET	END	TEST IF LAST CARD.	HM051334
00137	0020 00 0 00146	TRA	SWICH	NOT THE LAST CARD....	HM051335
00140	-0500 00 0 00131	CAL	BPTES		HM051336
00141	0602 60 0 00005	SLW*	\$(TES)		HM051337
00142	0774 00 1 00000	X1	**1	ALL DONE. EXIT	HM051338
00143	0774 00 2 00000	X2	**2		HM051339
00144	0774 00 4 00000	X4	**4		HM051340
00145	0020 00 4 00005	TRA	5,4		HM051341
					HM051342
00146	-0500 00 0 77740	SWICH CAL	CIMAGE	UPDATE THE CARD ORIGIN.	HM051343
00147	0361 00 0 00333	ACL	A22		HM051344
00150	0602 00 0 77740	SLW	CIMAGE		HM051345

BINPU ROUTINE TO WRITE COL BIN CARDS ON TAPE. FIBII

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00151	0020	00	0	00057	TRA	NEXT	HM051346
*****							HM051347
00152	0774	00	2	00027	TRCD	AXT	HM051348
00153	0600	00	2	77770	STZ	STZ	HM051349
00154	2	00001	2	00153	TIX	CIMAGE+24,2	HM051350
00155	0500	00	0	00322	CLA	*-1,2,1	HM051351
00156	0622	00	0	77740	STD	ZWC	HM051352
00157	0020	00	0	00073	TRA	CIMAGE	HM051353
*****							HM051354
*****							HM051355
*****							HM051356
00160	0600	00	0	77776	OUT	END	HM051357
00161	-2	00001	2	00070	TNX	IN,2,1	HM051358
00162	0602	00	0	77777	SLW	COMMON	HM051359
00163	-0754	00	2	00000	PXD	0,2	HM051360
00164	0402	00	0	77740	SUB	CIMAGE	HM051361
00165	0622	00	0	77740	STD	CIMAGE	HM051362
00166	-0500	00	0	77777	CAL	COMMON	HM051363
00167	-3	00000	2	00070	TXL	IN,2,0	HM051364
00170	0600	00	2	77770	STZ	CIMAGE+24,2	HM051365
00171	1	77777	2	00167	TXI	*-2,2,-1	HM051366
*****							HM051367
*****							HM051368
*****							HM051369
*****							HM051370
*****							HM051371
*****							HM051372
00172	-0754	00	0	00000	COSEQ	PXD	HM051373
00173	-0520	00	0	00302	NZT	BLSEQ	HM051374
00174	0020	00	0	00211	TRA	COSEQX	HM051375
00175	0765	00	0	00022	LRS	18	HM051376
00176	0221	00	0	00332	DVP	TEN	HM051377
00177	0601	00	0	77777	STD	COMMON	HM051378
00200	-0754	00	0	00000	PXD	PXD	HM051379
00201	0221	00	0	00332	DVP	TEN	HM051380
00202	0767	00	0	00006	ALS	6	HM051381
00203	-0602	00	0	77777	ORS	COMMON	HM051382
00204	-0754	00	0	00000	PXD	PXD	HM051383

TAB	HTR	TB	0 WITH CARRY
* TABLES FOR BCD-COL. BIN. CONVERSION			
* HOLES ARE FILLED IN WITH CONSTANTS			
TAB	OCT	1000,400,200,100,40,20,10,4,2,1	

MSK2CH OCT 77777770000,102,42

HM051412

BINPU ROUTINE TO WRITE COL BIN CARDS ON TAPE. FIBII

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00243 +000000000102					
00244 +000000000042					
00245 +000000000000	10123 OCT	0,0,0			HM051413
00246 +000000000000					
00247 +000000000000					
00250 +000000004000	OCT	4000,4400,4200,4100,4040,4020,4010,4004,4002,4001			HM051414
00251 +000000004400					
00252 +000000004200					
00253 +000000004100					
00254 +000000004040					
00255 +000000004020					
00256 +000000004010					
00257 +000000004004					
00260 +000000004002					
00261 +000000004001					
00262 106060606060	BC18 BCI	1,8			HM051415
00263 +000000004102	OCT	4102,4042			HM051416
00264 +000000004042					
00265 -3 7777 7 00000	MSKPD1 TXL	0,7,-1			HM051417
00266 0400 00 0 00000	REL ADD				HM051418
00267 606060606060	SEQNO BCI	1,			HM051419
00270 +000000002000	OCT	2000,2400,2200,2100,2040,2020,2010,2004,2002,2001			HM051420
00271 +000000002400					
00272 +000000002200					
00273 +000000002100					
00274 +000000002040					
00275 +000000002020					
00276 +000000002010					
00277 +000000002004					
00300 +000000002002					
00301 +000000002001					
00302 0 0000 0 00000	BLSEQ				HM051421
00303 +000000002102	OCT	2102,2042			HM051422
00304 +000000002042					
00305 606060606060	BC10 BCI	1,			HM051423
00306 606060606060	BLANK BCI	1,			HM051424
00307 0074 00 0 00000	MSKTSX TSX	0,			HM051425
00310 +000000000000	OCT	0,1400,1200,1100,1040,1020,1010,1004,1002,1001			HM051426

BINPU ROUTINE TO WRITE COL BIN CARDS ON TAPE. FIBII

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PAGE 8

00311 +000000001400
 00312 +000000001200
 00313 +000000001100
 00314 +000000001040
 00315 +000000001020
 00316 +000000001010
 00317 +000000001004
 00320 +000000001002
 00321 +000000001001
 00322 +000500000000
 00323 +000000001102
 00324 +000000001042
 00325 0 00001 0 00000
 00326 0 00000 0 00000
 00327 0 00000 0 00001
 00330 0 00000 0 00005
 00331 0 00014 0 00020
 00332 +000000000012
 00333 0000 00 0 00026
 00334 +000526000000

77777
 77740
 77730
 77776

ZWC OCT OCT
 OCT OCT

000500000000
 1102,1042

0,0,1

DI
 IDLCD
 L(1)
 5A
 TEN
 A22
 IMAGE
 COMMON
 CIMAGE
 LAST
 END

PZE
 PZE
 PZE
 PZE
 DEC
 HTR
 OCT

CONTROL WORD SKELETON

000526000000
 -1
 -32
 -40
 COMMON-1

HM051427
 HM051428

HM051429
 HM051430
 HM051431
 HM051432
 HM051433
 HM051434
 HM051435
 HM051436
 HM051437
 HM051438
 HM051439
 HM051440
 HM051441

BINPU ROUTINE TO WRITE COL BIN CARDS ON TAPE. FIBII
POST PROCESSOR ASSEMBLY DATA

335 IS THE FIRST LOCATION NOT USED BY THIS PROGRAM

REFERENCES TO DEFINED SYMBOLS

330	5A				
325	D1	16			
54	G2	34			
51	G3	37			
31	G4	53			
53	G5	50,	52		
70	IN	161,	167		
216	T8	227			
142	X1	6			
143	X2	7			
144	X4	54			
331	14D	11,	122		
333	A22	147			
105	ABC	115,	117		
77776	END	15,	55,	136,	160,
160	OUT	66			
266	REL	24			
120	SV1	102			
230	TAB	107			
215	T81	134,	215,	216,	217,
332	TEN	176,	201,	205	
322	ZWC	155			
262	BCI8	36			
73	EDIT	157			
77730	LAST	113,	214,	335	
66	LOCN	17			
327	L(1)	74,	133		
57	NEXT	151			
152	TRCD	56			
62	ARRAY	13,	67		
305	BCDID	51,	76		
6	BINPU				
306	BLANK	211			

BINPU ROUTINE TO WRITE COL BIN CARDS ON TAPE. FIBII
POST PROCESSOR ASSEMBLY DATA

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302	BLSEQ	30,	40,	173
131	BPTES	140		
172	COSEQ	45		
61	COUNT	20,	70	
245	ID123			
326	IDLCD	100,	116	
334	IMAGE	25		
267	SEQNO	47,	73,	132, 135
146	SWTCH	137		
121	WRITE			
0	(IOS)	123		
2	(RCH)	126		
5	(TES)	141		
4	(WER)	131		
1	(WRS)	124		
3	(WTC)	130		
77740	CIMAGE	26,	63,	71, 72, 146, 150, 153, 156, 164, 165, 170, 213, 335
77777	COMMON	162,	166,	177, 203, 207, 335
211	COSEQX	174		
242	MSK2CH			
265	MSKPTI	32		
307	MSKTSX	33		
213	PUNCMD	125		
122	WRITE1			

NO ERROR IN ABOVE ASSEMBLY.

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```

MPUNCH
C  MATRIX COLUMN BINARY PUNCH SUBROUTINE
C  CALLING SEQUENCE.....
C  CALL MPUNCH (A,M,N,IDUT,ITRA,IORG,BCDZ,MAXM,NTAPE )
C
C      A = MATRIX TO BE PUNCHED      IORG = ORIGIN OF FIRST CARD
C      M = NUMBER OF ROWS             BCDZ = BCD IDENTIFICATION WRD
C      N = NUMBER OF COLUMNS         MAXM = DIMENSIONED NUMBER OF
C      IDUT = 0, PUNCH BY COLUMNS    ROWS
C      = 1, PUNCH BY ROWS            NTAPE= OUTPUT PUNCH TAPE
C      ITRA = 0, TRA CARD AFTER WHOLE MATRIX
C      = 1, TRA CARD AFTER EACH ROW OR COLUMN
C
SUBROUTINE MPUNCH (A,M,N,IDUT,ITRA,IORG,BCDZ,MAXM,NTAPE,NCARDS )
DIMENSION  A(1), T(22)
IS=NCARDS
MN = MAXM*N
IF ( IDUT ) 8,2,8

C PUNCH BY COLUMNS
2 J = 1
J1= IORG
DO 5 I=1,N
CALL BINPU (A(J), M, J1, BCDZ, IS, NTAPE )
J = J+MAXM
IS= IS+1+M/22
IF ( ITRA ) 3,4,3

C PUNCH TRA CARD AFTER EACH COLUMN
3 CALL BINPU (A, 0, 0, BCDZ, IS, NTAPE )
IS=IS+1
GOTO 5

```

HM051443
 HM051444
 HM051445
 HM051446
 HM051447
 HM051448
 HM051449
 HM051450
 HM051451
 HM051452
 HM051453
 HM051454
 HM051455
 HM051456
 HM051457
 HM051458
 HM051459
 HM051460
 HM051461
 HM051462
 HM051463
 HM051464
 HM051465
 HM051466
 HM051467
 HM051468
 HM051469
 HM051470
 HM051471
 HM051472
 HM051473
 HM051474
 HM051475
 HM051476
 HM051477
 HM051478
 HM051479

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MPUNCH

4 J1=J1+M
5 CONTINUE

IF (ITRA) 16,6,16

C PUNCH TRA CARD AFTER WHOLE MATRIX

6 CALL BINPU (A, 0, 0, BCDZ, IS, NTAPE)
GOTO 16

C PUNCH MATRIX BY ROWS.

8 J=0
J1=IORG
DO 14 I=1,M
DO 10 K=1,MN,MAXM
J=J+1
T(J)=A(K)
IF (J-22) 10,9,9

9 CALL BINPU (T,22,J1,BCDZ,IS,NTAPE)
J1=J1+22
IS=IS+1
J=0
10 CONTINUE

IF (J) 12,12,11
11 CALL BINPU (T,J,J1,BCDZ,IS,NTAPE)
J1=J1+J
IS=IS+1
12 IF (ITRA) 13,14,13

C PUNCH TRA CARD AFTER EVERY ROW
13 CALL BINPU (T,0,0,BCDZ,IS,NTAPE)
IS=IS+1
J1=IORG
14 J=0

C OR AFTER ENTIRE MATRIX

HM051480
HM051481
HM051482
HM051483
HM051484
HM051485
HM051486
HM051487
HM051488
HM051489
HM051490
HM051491
HM051492
HM051493
HM051494
HM051495
HM051496
HM051497
HM051498
HM051499
HM051500
HM051501
HM051502
HM051503
HM051504
HM051505
HM051506
HM051507
HM051508
HM051509
HM051510
HM051511
HM051512
HM051513
HM051514
HM051515
HM051516
HM051517

MPUNCH

```
IF ( ITRA ) 16,15,16
15 CALL BINPU (A,O,O,BCDZ,IS,NTAPE)
IS=IS+1
16 NCARDS=IS-1
RETURN
END(1,0,0,0,0,0,0,0,0,1,0,0,0,0,0)
```

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PAGE 3

HM051518
HM051519
HM051520
HM051521
HM051522
HM051523

MPUNCH

3/12/63

PAGE 4

STORAGE NOT USED BY PROGRAM

DEC OCT
288 00440
32561 77461

STORAGE LOCATIONS FOR VARIABLES APPEARING IN DIMENSION AND EQUIVALENCE STATEMENTS

DEC OCT DEC OCT DEC OCT
T 287 00437

STORAGE LOCATIONS FOR VARIABLES NOT APPEARING IN COMMON, DIMENSION, OR EQUIVALENCE STATEMENTS

DEC OCT DEC OCT DEC OCT
I 265 00411 IS 264 00410 J1 263 00407 J 262 00406
MN 261 00405

LOCATIONS FOR OTHER SYMBOLS NOT APPEARING IN SOURCE PROGRAM

DEC OCT DEC OCT DEC OCT
1) 258 00402 2) 247 00367 6) 250 00372 9) 256 00400
C) 260 00404 E) 117 00165 E) 123 00173 E) 218 00332

LOCATIONS OF NAMES IN TRANSFER VECTOR

DEC OCT DEC OCT DEC OCT
BINPU 0 00030

ENTRY POINTS TO SUBROUTINES NOT OUTPUT FROM LIBRARY

BINPU

EXTERNAL FORMULA NUMBERS WITH CORRESPONDING INTERNAL FORMULA NUMBERS AND OCTAL LOCATIONS

EFN	IFN	LOC	EFN	IFN	LOC	EFN	IFN	LOC
2	7	00110	3	15	00152	4	19	00166
6	22	00200	8	25	00210	9	32	00244
11	39	00274	12	43	00313	13	44	00315
15	50	00346	16	53	00360	14	48	00333
						5	20	00171
						10	37	00266

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MATRIX ITERATION SUBROUTINE, REAL OR COMPLEX.

```

C CALLING SEQUENCE.....
C A = MATRIX, DIMENSIONED (MAXR X 2*N) - REAL
C NTAPE=UTILITY TAPE, IF + MATRIX IS IN CORE AT A, IF - ITS ON NTAPE
C N = ORDER OF MATRIX
C GUESS=1ST. GUESS VECTOR, DIMENSIONED (MAXR X 1) - REAL
C NGUESS=0, ROUTINE SUPPLIES GUESS VECTOR
C =+1, GUESS CONTAINS GUESS VECTOR
C NMODE=NUMBER OF EIGEN SOLUTIONS REQUESTED.
C VECTOR=EIGENVECTORS, DIMENSIONED (MAXR X 2*NMODE) - COMPLEX
C EIGVAL=EIGENVALUES (NMODE X 1) - REAL
C (NMODE*2 X 1) - COMPLEX
C NITER=NUMBER OF ITERATIONS PER MODE
C NITRSP, NITROP = MAXIMUM NUMBER OF SINGLE AND DOUBLE PREC. ITERATION
C EPSP, EPOP=CONVERGENCE CRITERIA FOR SINGLE AND DOUBLE ROOTS.
C IR = ERROR INDICATOR
C US=CHECK EIGENVECTORS, DIMENSIONED (MAXR X NMODE) - REAL
C H=WORKING AREA OF CORE DIMENSIONED (MAXR X 2*NMODE) - COMPLEX
C (MAXR X (NMODE+4)) - REAL
C (MAXR X 2*(NMODE+4)) - COMPLEX
C WILL CONTAIN CHECK EIGENVALUES, IF REQUESTED.
C NTAPE1=TAPE NUMBER OF OUTPUT PRINT TAPE
C IF = 0, NO RESULTS WILL BE PRINTED
C MAXR = DIMENSIONED NUMBER OF ROWS
C NC = 1, PROBLEM REAL
C = 2, PROBLEM COMPLEX
C AITKEN = AITKEN CONVERGENCE CRITERIA
C NAKSR,NAKDR = NUMBER OF TIMES AITKEN APPLIED IN EACH MODE.

SUBROUTINE MITERS (A, NTAPE, N, GUESS, NGUESS, NMODE, VECTOR,
1 EIGVAL, NITER, NITRSP, NITROP, EPSP, EPOP, MTRSO370,
MTRSO371, MTRSO372, MTRSO373, MTRSO374, MTRSO375, MTRSO376,
MTRSO340, MTRSO341, MTRSO342, MTRSO343, MTRSO344, MTRSO345, MTRSO346,
MTRSO347, MTRSO348, MTRSO349, MTRSO350, MTRSO351, MTRSO352,
MTRSO353, MTRSO354, MTRSO355, MTRSO356, MTRSO357, MTRSO358,
MTRSO359, MTRSO360, MTRSO361, MTRSO362, MTRSO363, MTRSO364,
MTRSO365, MTRSO366, MTRSO367, MTRSO368, MTRSO369, MTRSO370,
MTRSO371, MTRSO372, MTRSO373, MTRSO374, MTRSO375, MTRSO376)

```

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MATRIX ITERATION SUBROUTINE, REAL OR COMPLEX.

```

2      IR, US, H, MAXR, NC, AITKEN, NAKSR, NAKOR,
3      NTAPE1 )

      DIMENSION A(1), GUESS(1), VECTOR(1), EIGVAL(1), NITER(1), US(1),
1      H(1), NAKSR(1), NAKOR(1), BOOLT(6)
      BOOLT(1)=606046652551
      BOOLT(2)=264346666060
      BOOLT(3)=243165312425
      BOOLT(4)=602330252342
      BOOLT(5)=314560234346
      BOOLT(6)=622560626422

      IF ACCUMULATOR OVERFLOW      3,3
      IF DIVIDE CHECK      4,4

3      FIND MATRIX AND STORE ON TAPE IF NECESSARY
4      I=NTAPE
      J2=MAXR*NC*N
      IF ( I )      5,8,6
5      I=-I
      REWIND I
      DO 1 J=1,N
1      READ TAPE I,      ( A(K),K=J,J2,MAXR )
      NTAPE = I
      GOTO 7

6      REWIND NTAPE
      DO 2 J=1,N
2      WRITE TAPE NTAPE,      ( A(K),K=J,J2,MAXR )
7      REWIND NTAPE

C      DEFINE PROGRAM CONSTANTS AND ZEROS.
8      MODE=0
      IR=0
      AT=AITKEN**2
      IF ( EPSP )      10,9,10
9      EPSP = .1E-08
10     IF ( EPDP )      12,11,12

```

MATRIX ITERATION SUBROUTINE, REAL OR COMPLEX.

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```

11 EPOP = .1E-08
12 IF ( NGUESS ) 15,13,15
13 J1=MAXR*(NC-1)
   00 14 I=1,N
      K=J1+I
      GUESS(K)=0.
14   GUESS(I)=1.
15 MODE = MODE+1
   NAKSR(MODE)=0
   NAKDR(MODE)=0
   IGO=1
   NITER(MODE)=0
   K1=NC*MAXR*(MODE-1)
   K2=K1+1
   K3=NC*(MODE-1)+1
   K4= NC*MAXR
   K5=K4*NMODE
   K6=K5+K4

C MOVE FIRST GUESS INTO POSITION
DO 16 J=1,NC
  J1=MAXR*(J-1)
  00 16 I=1,N
    K=K1+J1+I
    L=J1+I
    H(K)=GUESS(L)
16
17 NAK=0
18 NITER(MODE)=NITER(MODE)+1
   NAK=NAK+1
   INDEX=0
   CALL DPMLTX (A,NC, H(K2),NC, VECTOR(K2), N,N,1, MAXR,MAXR,MAXR,1)
   CALL NPNRMX (VECTOR(K2), H(K2), N, EIGVAL(K3), INDEX, MAXR, NC,1)

C TEST FOR SINGLE ROOT CONVERGENCE

```

MTRS0415
MTRS0416
MTRS0417
MTRS0418
MTRS0419
MTRS0420
MTRS0421
MTRS0422
MTRS0423
MTRS0424
MTRS0425
MTRS0426
MTRS0427
MTRS0428
MTRS0429
MTRS0430
MTRS0431
MTRS0432
MTRS0433
MTRS0434
MTRS0435
MTRS0436
MTRS0437
MTRS0438
MTRS0439
MTRS0440
MTRS0441
MTRS0442
MTRS0443
MTRS0444
MTRS0445
MTRS0446
MTRS0447
MTRS0448
MTRS0449
MTRS0450
MTRS0451
MTRS0452

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MATRIX ITERATION SUBROUTINE, REAL OR COMPLEX.

```

DO 23 J=1,NC
  J1=(J-1)*MAXR
  K=K1+J1
  GOTO (24,19,21),NAK
19  L=K5+J1
   00 20 I=1,N
   L=L+1
   K=K+1
   IF ( ABSF(H(L)-H(K)) - EPSP )
20   CONTINUE
   GOTO 100
21  DO 22 I=1,N
   K=K+1
   IF ( ABSF(US(K)-H(K)) - EPSP )
22   CONTINUE
23  CONTINUE
100 IF ACCUMULATOR OVERFLOW      108,102
102 IF DIVIDE CHECK                104,56
104 NP=3
   GOTO 109

C NO CONVERGENCE, SO TEST MAXIMUM NUMBER OF ITERATIONS.
24 IF ( NITER(MODE)-NITRSP )
25   GOTO (40,44,31),NAK

C NDT YET EXCEEDED, SO TRY FOR AITKENS TIME.

C TEST FOR AITKENS CONVERGENCE.
31 GOTO (26,36),NC

26 00 28 I=1,N
   J=K5+I
   K=K1+I
   IF ( US(K)-H(J) ) 27,261,27
261 IF ( H(K)-US(K) ) 32,28,32
27  IF ( ABSF( H(K)-US(K) ) / ( US(K)-H(J) ) ) - AITKEN ) 28,28,32
28  CONTINUE

```

MTRS0453
MTRS0454
MTRS0455
MTRS0456
MTRS0457
MTRS0458
MTRS0459
MTRS0460
MTRS0461
MTRS0462
MTRS0463
MTRS0464
MTRS0465
MTRS0466
MTRS0467
MTRS0468
MTRS0469
MTRS0470
MTRS0471
MTRS0472
MTRS0473
MTRS0474
MTRS0475
MTRS0476
MTRS0477
MTRS0478
MTRS0479
MTRS0480
MTRS0481
MTRS0482
MTRS0483
MTRS0484
MTRS0485
MTRS0486
MTRS0487
MTRS0488
MTRS0489
MTRS0490

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MATRIX ITERATION SUBROUTINE, REAL OR COMPLEX.

```

C ALL VECTOR ELEMENTS OK, SO APPLY AITKENS SPEEDER-UPPER.
00 30 I=1,N
    J=K5+I
    K=K1+I
    Q=(H(K)-2.*US(K)+H(J))
    IF ( Q ) 29,30,29
29 H(K)=H(J) - ( (US(K)-H(J))**2 / Q )
30 CONTINUE
    NAKSR(MODE)=NAKSR(MODE) + 1
    GOTO 17

C CONVERGENCE TEST NOT MET. RESTORE AND TRY AGAIN.
32 00 33 L=1,NC
    J1=(L-1)*MAXR
    00 33 I=1,N
    J=K1+J1+I
    K=K5+J1+I
    H(K)=US(J)
33 US(J)=H(J)
    NAK=2
    GOTO 18

C IF PROBLEM COMPLEX, REPEAT ALL ABOVE FOR COMPLEX ARITHMETIC.
36 00 38 I=1,N
    J=K5+I
    K=K1+I
    JJ=J+MAXR
    KK=K+MAXR
    Q = (US(K)-H(J))**2 + (US(KK)-H(JJ))**2
    IF ( Q ) 37,36,37
361 IF ( (H(K)-US(K))**2 + (H(KK) - US(KK))**2 ) 32,38,32
37 IF ( (H(K)-US(K))**2 + (H(KK) - US(KK))**2 ) / Q-AT) 38,38,32
38 CONTINUE
    00 39 I=1,N
    J=K5+I

```

MTRS0491
MTRS0492
MTRS0493
MTRS0494
MTRS0495
MTRS0496
MTRS0497
MTRS0498
MTRS0499
MTRS0500
MTRS0501
MTRS0502
MTRS0503
MTRS0504
MTRS0505
MTRS0506
MTRS0507
MTRS0508
MTRS0509
MTRS0510
MTRS0511
MTRS0512
MTRS0513
MTRS0514
MTRS0515
MTRS0516
MTRS0517
MTRS0518
MTRS0519
MTRS0520
MTRS0521
MTRS0522
MTRS0523
MTRS0524
MTRS0525
MTRS0526
MTRS0527
MTRS0528

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MATRIX ITERATION SJ8ROUTINE, REAL OR COMPLEX.

```

JJ=J+MAXR
K=K1+1
KK=K+MAXR
Q = (H(K)-2.*US(K)+H(J))**2 + (H(KK)-2.*US(KK)+H(JJ))**2
IF ( Q ) 35,39,35
35 X=H(K)

H(K)= H(J) - ( ((US(K)-H(J))**2-(US(KK)-H(JJ))**2)*(H(K)-2.*
US(K)+H(J))+2.*US(K)-H(J))*(US(KK)-H(JJ))*
(H(KK)-2.*US(KK)+H(JJ)) / Q )
H(KK)=H(JJ)-(((2.*US(K)-H(J))*(US(KK)-H(JJ))*(X-2.*
US(K)+H(J))-((US(K)-H(J))**2-(US(KK)-H(JJ))**2)
*(H(KK)-2.*US(KK)+H(JJ)) / Q )

39 CONTINUE
NAKSR(MODE) = NAKSR(MODE) + 1
GOTO 17

40 DO 41 J=1,NC
J1=MAXR*(J-1)
DO 41 I=1,N
K=K1+J1+I
L=K5+J1+I
H(L)=H(K)
41 GOTO (18,56),IGO

44 DO 45 J=1,NC
J1=MAXR*(J-1)
DO 45 I=1,N
K=K1+J1+I
US(K)=H(K)
45 GOTO 18

C MODE DID NOT CONVERGE IN NORMAL ITERATION, SO SETUP AND TRY CLOSE
C ROOTS PROCEDURE
46 K=K6+2*K4+1
L=K+2*K4
J=K3+NC

```

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MATRIX ITERATION SUBROUTINE, REAL OR COMPLEX.

```

      IF ACCUMULATOR OVERFLOW      108,113
113 IF OVIDE CHECK      104,49
49 CALL CLOSE (A,H(K2),H(K),H(K6+1),EIGVAL(K3),EIGVAL(J), INOEX1,
1 INOEX,N,NC,NITRDP,NTRY,NAKDR(MOOE),EPOP,EPSP,AITKEN,
2 AT,MAXR,US(K2),H(L),IR)
      IF ( IR ) 114,52,50
114 NP=5
      GOTO 109
50 IR=0
      NITER(MOOE) = NITRSP+NTRY
      GOTO 56
52 00 54 J=1,NC
      JJ=MAXR*(J-1)+INDEX1
      00 54 I=1,N
      K=K1+JJ+I
      KK=K1+K4+JJ+I
      LL=K6+2*K4+JJ+I
      L=K6+JJ+I
      H(K)=H(LL)
      H(KK)=H(L)
      US(K)=A(JI)
54 JI=J1+K4

      CALL SWEEPX (VECTOR,A,H,US,EIGVAL,MOOE,N,MAXR,NC,INDEX1,EPSP)
      IF ACCUMULATOR OVERFLOW      108,116
123 NITS=1
      GOTO 107
116 IF OVIDE CHECK      123,117
117 J1=(NC-1)*MAXR+INOEX1
      GUESS(J1)=0.
      GUESS(INDEX1)=0.
      NITER(MOOE)=0
      MODE=MOOE+1
      IF ( NMODE-MODE ) 53,55,55
53 NITER(MODE)=NITRSP+NTRY
      K1=NC*MAXR*(MODE-1)
      00 531 I=1,NC

```

MTRS0567
MTRS0568
MTRS0569
MTRS0570
MTRS0571
MTRS0572
MTRS0573
MTRS0574
MTRS0575
MTRS0576
MTRS0577
MTRS0578
MTRS0579
MTRS0580
MTRS0581
MTRS0582
MTRS0583
MTRS0584
MTRS0585
MTRS0586
MTRS0587
MTRS0588
MTRS0589
MTRS0590
MTRS0591
MTRS0592
MTRS0593
MTRS0594
MTRS0595
MTRS0596
MTRS0597
MTRS0598
MTRS0599
MTRS0600
MTRS0601
MTRS0602
MTRS0603
MTRS0604

MATRIX ITERATION SUBROUTINE, REAL OR COMPLEX.

```

      K= K1+MAXR*(I-1)
      DO 531 J=1,N
      L=K+J
      VECTOR(L)=0.
531  GOTD 59
      55 NAKDR(MDDE)=NAKDR(MDDE-1)
      NAKDR(MDDE-1)=0
      NITER(MDDE) = NITERSP + NTRY
      K1=NC*MAXR*(MDDE-1)
56 DO 58 J=1,NC
      J1=MAXR*(J-1)+INDEX
      DO 58 I=1,N
      K=K1+MAXR*(J-1)+I
      US(K)=A(J1)
      J1=J1+K4
58  CALL SWEEPX (VECTOR,A,H,US,EIGVAL,MDDE,N,MAXR,NC,INDEX,EPSP)
      IF ACCUMULATOR OVERFLOW 108,106
106 IF DIVIDE CHECK 124,59
124 NUTS=2
107 IF (NTAPE1) 119,118,119
118 IR=MODE
      GOTD 120
119 WRITE OUTPUT TAPE NTAPE1, 131, BOOIT(3),BOOIT(4)
      WRITE OUTPUT TAPE NTAPE1, 134, MDDE
120 DO 125 I=1,NC
      K=K1+MAXR*(I-1)
      DO 125 J=1,N
      K=K+1
125  VECTOR(K)=0.
      GOTD (117,59),NUTS
59  J1=(NC-1)*MAXR+INDEX
      GUESS(J1)=0.
      GUESS(INDEX)=0.
62 IF (NMODE-MDDE) 70,70,15

```

MTRS0605
MTRS0606
MTRS0607
MTRS0608
MTRS0609
MTRS0610
MTRS0611
MTRS0612
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MTRS0630
MTRS0631
MTRS0632
MTRS0633
MTRS0634
MTRS0635
MTRS0636
MTRS0637
MTRS0638
MTRS0639
MTRS0640
MTRS0641
MTRS0642

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MATRIX ITERATION SUBROUTINE, REAL OR COMPLEX.

```

108 NP=1
109 IR=MODE
    IF ( NTAPE1 ) 121, 70, 121
121 WRITE OUTPUT TAPE NTAPE1, 131, BOOILT(NP), BOOILT(NP+1)
    MODE=MODE-1
    WRITE OUTPUT TAPE NTAPE1, 132, MODE
    NMODE = MODE
    GOTO 70

70 IF ( NTAPE ) 71, 75, 71
71 DO 73 J=1, N
73 READ TAPE NTAPE, (A(I), I=J, J2, MAXR)
    CALL DPMLTX (A, NC, VECTOR, NC, US, N, N, MODE, MAXR, MAXR, MAXR, 1)
    J=1
    K=1
    DO 72 I=1, MODE
    INDEX=0
        CALL NPNRMX (US(J), US(J), N, H(K), INDEX, MAXR, NC, 1)
        J=J+K4
72 K=K+NC
75 IF ( NTAPE1 ) 92, 92, 80

80 WRITE OUTPUT TAPE NTAPE1, 95
    DO 86 I=1, MODE
        LITR=NITER(I)-NITRSP
        IF ( LITR ) 81, 82, 82
81 LITR=0
        GOTO 85
82 NITER(I)=NITRSP
        IF (LITR-NITRDP) 85, 85, 87
87 WRITE OUTPUT TAPE NTAPE1, 94, I
85 GOTO (83, 84), NC
83 WRITE OUTPUT TAPE NTAPE1, 97, ( I, EIGVAL(I), NITER(I), LITR,
    1 NAKSR(I), NAKDR(I))
        GOTO 86
84 L=2*I-1
    WRITE OUTPUT TAPE NTAPE1, 96, ( I, EIGVAL(L), EIGVAL(L+1), NITER(I), NITER(I+1))
MTRS0643
MTRS0644
MTRS0645
MTRS0646
MTRS0647
MTRS0648
MTRS0649
MTRS0650
MTRS0651
MTRS0652
MTRS0653
MTRS0654
MTRS0655
MTRS0656
MTRS0657
MTRS0658
MTRS0659
MTRS0660
MTRS0661
MTRS0662
MTRS0663
MTRS0664
MTRS0665
MTRS0666
MTRS0667
MTRS0668
MTRS0669
MTRS0670
MTRS0671
MTRS0672
MTRS0673
MTRS0674
MTRS0675
MTRS0676
MTRS0677
MTRS0678
MTRS0679
MTRS0680

```

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MATRIX ITERATION SUBROUTINE, REAL OR COMPLEX.

```

1 CONTINUE
86 IF ( MODE ) 92,92,88
88 WRITE OUTPUT TAPE NTAP1, 98
L=MODE*NC
CALL MPRINT (VECTOR,N,L,MAXR,NTAP1)
IF ( NTAP1 ) 92,92,90
90 WRITE OUTPUT TAPE NTAP1, 99
WRITE OUTPUT TAPE NTAP1, 93, (H(I),I=1,L)
CALL MPRINT (US,N,L,MAXR,NTAP1)
92 RETURN
93 FORMAT ( 1H 6E16.8 )
94 FORMAT (5H MODE 114, 40H HAS NOT CONVERGED IN MAXIMUM ITERATIONS)
95 FORMAT (1H 5X, 6H MODE 13X, 11H EIGENVALUE 19X,
2 24H ITERATIONS S.P. D.P. 6X, 13H AITKENS S.P. 2X,
3 5H D.P. ///)
96 FORMAT (1H 1111, 2E19.8, 1122, 117, 1119, 117)
97 FORMAT (1H 1111, 9X, 1E20.8, 9X, 1122, 117, 1119, 117)
98 FORMAT (1H0 /// 1H0 46X, 14H EIGENVECTORS /// )
99 FORMAT (1H1 /// 1H0 36H CHECK EIGENVALUES AND EIGENVECTORS )
131 FORMAT (35H ERROR IN ITERATION SUBROUTINE... ( 2A6, 1H) )
132 FORMAT (25H+ CALCULATION TERMINATED. 116,19H MODES ARE CORRECT.)
134 FORMAT (14H+ IN TRUE MODE 116, 27H CALCULATION. MODIFIED MODE
1 12H IS CORRECT. )
END(1,0,0,0,0,0,0,0,0,1,0,0,0,0,0)

```

,LITR,NAKSR(I),NAKDR(I))

MTRS0681
MTRS0682
MTRS0683
MTRS0684
MTRS0685
MTRS0686
MTRS0687
MTRS0688
MTRS0689
MTRS0690
MTRS0691
MTRS0692
MTRS0693
MTRS0694
MTRS0695
MTRS0696
MTRS0697
MTRS0698
MTRS0699
MTRS0700
MTRS0701
MTRS0702
MTRS0703
MTRS0704

STORAGE NOT USED BY PROGRAM

DEC	OCT	DEC	OCT
2305	04401	32561	77461

STORAGE LOCATIONS FOR VARIABLES APPEARING IN DIMENSION AND EQUIVALENCE STATEMENTS

DEC	OCT	DEC	OCT
800LT	2304 04400		

STORAGE LOCATIONS FOR VARIABLES NOT APPEARING IN COMMON, DIMENSION, OR EQUIVALENCE STATEMENTS

DEC	OCT	DEC	OCT	DEC	OCT
AT	2298 04372	IGO	2297 04371	INDEX1	2296 04370
I	2294 04366	J1	2293 04365	J2	2292 04364
J	2290 04362	K1	2289 04361	K2	2288 04360
K4	2286 04356	K5	2285 04355	K6	2284 04354
K	2282 04352	L1TR	2281 04351	LL	2280 04350
MOOE	2278 04346	NAK	2277 04345	NP	2276 04344
NUTS	2274 04342	Q	2273 04341	X	2272 04340
				INDEX	2295 04367
				JJ	2291 04363
				K3	2287 04357
				KK	2283 04353
				L	2279 04347
				NTRY	2275 04343

SYMBOLS AND LOCATIONS FOR SOURCE PROGRAM FORMAT STATEMENTS

EFN	LOC	EFN	LOC	EFN	LOC
812T	93 04256	812U	94 04253	812V	95 04241
8131	97 04200	8132	98 04167	8133	99 04157
8144	132 04133	8146	134 04121		

LOCATIONS FOR OTHER SYMBOLS NOT APPEARING IN SOURCE PROGRAM

DEC	OCT	DEC	OCT
1)	2226 04262	2)	2094 04056
6)	2109 04075	3)	2099 04063
C1G3	2238 04276	C1G1	2236 04274
C1G7	2242 04302	C1G4	2239 04277
C1G8	2246 04306	C1G8	2243 04303
C1GF	2250 04312	C1GC	2247 04307
		C1GG	2251 04313
		C1GJ	2254 04316
		C1G2	2237 77777
		C1G6	2241 04301
		C1GA	2245 04305
		C1GE	2249 04311
		C1GI	2253 04315
		C1G5	2252 04317
		C1G9	2244 04304
		C1G0	2248 04310
		C1GH	2252 04314
		C1204	2256 04320
		C1203	2255 04317
		C1205	2257 04321
		C1G2	2237 77777
		C1G6	2241 04301
		C1GA	2245 04305
		C1GE	2249 04311
		C1GI	2253 04315
		C1G5	2252 04317
		C1G9	2244 04304
		C1G0	2248 04310
		C1GH	2252 04314
		C1204	2256 04320
		C1203	2255 04317
		C1205	2257 04321

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MATRIX ITERATION SUBROUTINE, REAL DR COMPLEX.

C1206	2258	04322	C1207	2259	04323	C1208	2260	04324	C120A	2261	04325
C1208	2262	04326	C120C	2263	04327	C120E	2264	04330	C120G	2265	04331
C120H	2266	04332	C120I	2267	04333	C120K	2268	04334	C120M	2269	04335
C1200	2270	04336	C120Q	2271	04337	D110S	678	01246	D110U	765	01375
D111A	842	01512	D1110	858	01532	D111F	867	01543	D111Q	960	01700
D1136	1630	03136	D113P	1840	03460	D120R	673	01241	D1211	795	01433
D121M	918	01626	D122A	1247	02337	D1231	1583	03057	D123V	1895	03547
D1241	1940	03624	D1243	1956	03644	D124G	2089	04051	D131A	841	01511
D132A	1246	02336	D1331	1582	03056	D1336	1629	03135	D133V	1894	03546
D1341	1939	03623	D1343	1955	03643	D134G	2088	04050	D140M	576	01100
D140N	636	01174	D140T	748	01354	D1416	825	01471	D141C	850	01522
D1426	1139	02163	D1427	1229	02315	D142F	1295	02417	D142P	1431	02627
D142V	1539	03003	D1430	1708	03254	D143H	1744	03320	D143L	1775	03357
D143M	1796	03404	D143N	1803	03413	D151D	857	01531	D1526	1138	02162
D1527	1228	02314	D152F	1294	02416	D152V	1538	03002	D153N	1802	03412
D153P	1839	03457	D160N	635	01173	D161C	849	01521	D162P	1430	02626
D163D	1707	03253	D163L	1774	03356	D163M	1795	03403	D171A	840	01510
D171D	856	01530	D173P	1838	03456	E1F	533	01025	E1I	543	01037
E1J	548	01044	E111	792	01430	E113	803	01443	E116	823	01467
E118	834	01502	E111	888	01570	E11L	910	01616	E110	948	01664
E12T	1529	02771	E13A	1677	03215	E13M	1794	03402	E145	1966	03656
E148	1986	03702	E148	2039	03767	E141Q	957	01675			

LOCATIONS OF NAMES IN TRANSFER VECTOR

	DEC	OCT	OPMLTX (FIL) (STH)	DEC	OCT	MPRINT (RLR) (TSB)	DEC	OCT	NPNRMX (RWT) (WLR)	DEC	OCT
CLOSE	7	00007		5	00005		11	00013	6	00006	
SWEEX	8	00010		10	00012		2	00002	0	00000	
(ST8)	3	00003		9	00011		1	00001	4	00004	
ENTRY POINTS TO SUBROUTINES NOT OUTPUT FROM LIBRARY											
CLOSE			OPMLTX (STH)			MPRINT (TSB)			NPNRMX (WLR)		
(ST8)						SWEEX			(RLR)		(RWT)

EXTERNAL FORMULA NUMBERS WITH CORRESPONDING INTERNAL FORMULA NUMBERS AND OCTAL LOCATIONS

EFN	IFN	LOC	EFN	IFN	LOC	EFN	IFN	LOC	EFN	IFN	LOC
3	22	00672	4	23	00674	5	26	00710	1	29	00723
6	36	00751	2	38	00762	7	43	01005	8	44	01011
9	48	01026	10	49	01030	11	50	01034	12	51	01040
13	52	01045	14	56	01072	15	57	01101	16	73	01226
17	74	01242	18	75	01247	19	86	01401	20	91	01441
21	93	01451	22	96	01500	23	97	01503	100	98	01513
102	100	01516	104	101	01523	24	103	01533	25	104	01537
31	105	01544	26	106	01546	261	110	01573	27	111	01600
28	112	01613	29	118	01651	30	119	01665	32	122	01701
33	128	01742	36	131	01761	361	138	02035	37	139	02054
38	140	02075	35	148	02164	39	151	02316	40	154	02330
41	159	02372	44	161	02410	45	165	02444	46	167	02455
113	172	02504	49	173	02506	114	176	02604	50	178	02611
52	181	02620	54	192	02725	123	197	02772	116	199	02777
117	200	03004	53	206	03034	531	212	03104	55	214	03115
56	218	03137	58	223	03201	106	228	03243	124	229	03246
107	230	03255	118	231	03261	119	233	03264	120	237	03312
125	241	03343	59	243	03360	62	246	03375	108	247	03405
109	248	03414	121	250	03420	70	257	03461	71	258	03463
73	259	03470	72	273	03611	75	274	03625	80	275	03631
81	279	03653	82	281	03657	87	283	03666	85	285	03700
83	286	03703	84	290	03730	86	294	03764	88	296	03774
90	301	04021	92	309	04052						

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SINGLE OR DOUBLE PRECISION, REAL OR COMPLEX VECTOR NORMALIZATION.

```

C      CALL NPNRMX(A, B, N, FL, INDEX, MD, NX, NP )
C      A=VECTOR TO BE NORMALIZED      H=NORMALIZED VECTOR(MAY=A)
C      N=SIZE                          FL=NORMALIZING NUMBER
C      INDEX=+ ON ENTRY, NORMALIZE ON NUMBER WHOSE INDEX IS INDEX
C      =0 ON ENTRY, NORMALIZE ON LARGEST S.H. AND SET
C      INDEX=TO ITS INDEX.
C      == ON ENTRY, NORMALIZE ON FL.
C      MD=SINGLE PRECISION DIMENSIONED NUMBER OF ROWS OF A AND B
C      NX=1, VECTOR REAL
C      =2, VECTOR COMPLEX
C      NP=1, SINGLE PRECISION
C      =2, DOUBLE PRECISION
C
C      SUBROUTINE NPNRMX (A, B, N, FL, INDEX, MD, NX, NP )
D      DIMENSION A(1), B(1), FL(1), D(1), C(1)
C
C      N1=1+NP
C      N2=N*NP
C      N4=MD*NP
C      IF ( INDEX ) 32, 7, 38
C      7 GOTO (11, 8), NX
C      8 FL= (A(1)*2+A(N4+1))*2)
C      INDEX=1
C      DO 10 K=N1, N2, NP
C      I=K+N4
C      D= (A(I)*2+A(I)*2)
C      IF ( FL-D ) 9, 9, 10
C      9 FL=0
C      INDEX=K
C      10 CONTINUE
C      6 FL=A(INDEX)
C      GOTO (18, 25), NP
C
C      11 FL=ABSF(A(1))
C      INDEX=1
C      DO 13 K=N1, N2, NP

```

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SINGLE OR DOUBLE PRECISION, REAL OR COMPLEX VECTOR NORMALIZATION.

```

D=ABSF(A(K))
IF (FL=0) 12,12,13
12 FL=D
INDEX=K
13 CONTINUE
14 FL=A(INDEX)
GOTO (16,21),NP
16 DO 17 I=1,N
17 B(I)=A(I)/FL
GOTO 30
18 I=INDEX+MD
FL(2)=A(I)
19 D=FL(1)**2+FL(2)**2
DO 20 I=1,N
K=I+MD
C=A(I)*FL(2)-A(K)*FL(1)
B(I)=(A(I)*FL(1)+A(K)*FL(2))/D
20 B(K)=-C/D
GOTO 30
21 FL(2)=A(INDEX+1)
23 DO 24 I=1,N2,NP
24 B(I)=A(I)/FL
D
GOTO 26
25 FL(2)=A(INDEX+1)
I=INDEX+N4
FL(3)=A(I)
D
26 D=FL(1)**2+FL(3)**2
DO 27 I=1,N2,NP
K=I+N4
C=A(I)*FL(3)-A(K)*FL(1)
D
B(I)=(A(I)*FL(1)+A(K)*FL(3))/D
27 B(K)=-C/D
D
28 INDEX=INDEX/2+1

```

MTRS0172
MTRS0173
MTRS0174
MTRS0175
MTRS0176
MTRS0177
MTRS0178
MTRS0179
MTRS0180
MTRS0181
MTRS0182
MTRS0183
MTRS0184
MTRS0185
MTRS0186
MTRS0187
MTRS0188
MTRS0189
MTRS0190
MTRS0191
MTRS0192
MTRS0193
MTRS0194
MTRS0195
MTRS0196
MTRS0197
MTRS0198
MTRS0199
MTRS0200
MTRS0201
MTRS0202
MTRS0203
MTRS0204
MTRS0205
MTRS0206
MTRS0207
MTRS0208
MTRS0209

SINGLE OR DOUBLE PRECISION, REAL OR COMPLEX VECTOR NORMALIZATION.

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30 RETURN

32 GOTO (34,36), NX

34 GOTO (16,23), NP

36 GOTO (19,26), NP

38 GOTO (40,39), NP

39 INDEX=2*INDEX-1

40 GOTO (14,6), NX

END(1,0,0,0,0,0,0,0,0,1,0,0,0,0,0)

MTRS0210

MTRS0211

MTRS0212

MTRS0213

MTRS0214

MTRS0215

MTRS0216

MTRS0217

MTRS0218

MTRS0219

STORAGE NOT USED BY PROGRAM

DEC	OCT	DEC	OCT
590	01116	32561	77461

STORAGE LOCATIONS FOR VARIABLES APPEARING IN DIMENSION AND EQUIVALENCE STATEMENTS

DEC	OCT	DEC	OCT	DEC	OCT
C	587 01113	D	589 01115		

STORAGE LOCATIONS FOR VARIABLES NOT APPEARING IN COMMON, DIMENSION, OR EQUIVALENCE STATEMENTS

DEC	OCT	DEC	OCT	DEC	OCT
I	585 01111	K	584 01110	N1	583 01107
N4	581 01105			N2	582 01106

LOCATIONS FOR OTHER SYMBOLS NOT APPEARING IN SOURCE PROGRAM

DEC	OCT	DEC	OCT	DEC	OCT
1)	571 01073	2)	560 01060	3)	562 01062
9)	568 01070	C)G0	575 01077	C)G3	577 01101
C)G4	578 01102	C)G5	579 01103	C)G6	580 01104
D)107	251 00373	D)10D	287 00437	D)10E	295 00447
D)10M	357 00545	D)10U	531 01023	O)1214	556 01054
D)607	248 00370	E)2	200 00310	E)5	243 00363
E)V	536 01030	E)12	545 01041	E)10E	292 00444

LOCATIONS OF NAMES IN TRANSFER VECTOR

DEC	OCT	DEC	OCT	DEC	OCT
OEXP(2	1 00001	(DFAD)	2 00002	(DFDP)	0 00000
(DP+SB)	4 00004			(DFMP)	3 00003

ENTRY POINTS TO SUBROUTINES NOT OUTPUT FROM LIBRARY

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SINGLE OR DOUBLE PRECISION, REAL OR COMPLEX VECTOR NORMALIZATION.

DEXP(2) (DFA0) (OFDP) (OFMP) (DFS8)

EXTERNAL FORMULA NUMBERS WITH CORRESPONDING INTERNAL FORMULA NUMBERS AND OCTAL LOCATIONS

EFN	IFN	LOC	EFN	IFN	LOC	EFN	IFN	LOC	EFN	IFN	LOC
7	8	00305	8	9	00311	9	15	00354	10	17	00364
6	18	00374	11	20	00401	12	25	00425	13	27	00433
14	28	00440	16	30	00450	17	31	00453	18	33	00461
19	35	00471	20	40	00531	21	42	00542	23	43	00546
24	44	00553	25	46	00574	26	49	00612	27	54	00760
28	55	01011	30	56	01024	32	58	01031	34	59	01033
36	60	01036	38	61	01042	39	62	01045	40	63	01055

```

C COMPUTES TRUE MODE AND SWEEPS IT FROM THE MATRIX. (REAL OR COMPLEX)
C
C HTRUE = TRUE MODAL COLUMNS, AS COMPUTED. U = DYNAMIC MATRIX.
C H = SERIES OF MODIFIED MODAL COLUMNS. FL= COLUMN OF EIGENVALUES.
C US = SERIES OF MODIFIED MODAL ROWS OF U.
C MODE = MODE NOW BEING COMPUTED. N = SIZE
C MD = DIMENSIONED NUMBER OF ROWS OF U,US,H,HTRUE
C NX = 1 IF PROBLEM IS REAL.
C      = 2 IF PROBLEM IS COMPLEX.
C
SUBROUTINE SWEEPX (HTRUE, U,H, US,FL, MODE, N, MD, NC, INDEX, EP)
DIMENSION H(1), US(1), U(1), HTRUE(1), FL(1), G(4)
M=MODE-1
K1=M*NC*MD
DO 6 J=1,NC
  K=K1+(J-1)*MD
  DO 6 L=1,N
    K=K+1
    HTRUE(K)=H(K)
6
  IF ( M ) 31,31,8
8 DO 25 I=1,M
  L1=NC*MD*(MODE-1) -NC*MD
  GOTO ( 9,11),NC
9 G=0.
DO 10 J=1,N
  L=L1+J
  K=K1+J
10 G=G+US(L)*HTRUE(K)
  GOTO 13
11 G(1)=0.
  G(2)=0.
DO 12 J1=1,N

```

MTRS0222
MTRS0223
MTRS0224
MTRS0225
MTRS0226
MTRS0227
MTRS0228
MTRS0229
MTRS0230
MTRS0231
MTRS0232
MTRS0233
MTRS0234
MTRS0235
MTRS0236
MTRS0237
MTRS0238
MTRS0239
MTRS0240
MTRS0241
MTRS0242
MTRS0243
MTRS0244
MTRS0245
MTRS0246
MTRS0247
MTRS0248
MTRS0249
MTRS0250
MTRS0251
MTRS0252
MTRS0253
MTRS0254
MTRS0255
MTRS0256
MTRS0257
MTRS0258

S W E E P X S U B R O U T I N E

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PAGE 2

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MTRS0259
MTRS0260
MTRS0261
MTRS0262
MTRS0263
MTRS0264
MTRS0265
MTRS0266
MTRS0267
MTRS0268
MTRS0269
MTRS0270
MTRS0271
MTRS0272
MTRS0273
MTRS0274
MTRS0275
MTRS0276
MTRS0277
MTRS0278
MTRS0279
MTRS0280
MTRS0281
MTRS0282
MTRS0283
MTRS0284
MTRS0285
MTRS0286
MTRS0287
MTRS0288
MTRS0289
MTRS0290
MTRS0291
MTRS0292
MTRS0293
MTRS0294
MTRS0295
MTRS0296

L=L1+J1
K=K1+J1
L2=L+MD
K2=K+MD
G(1)=G(1)+US(L)*HTRUE(K)-US(L2)*HTRUE(K2)
G(2)=G(2)+US(L)*HTRUE(K2)+US(L2)*HTRUE(K)
12 G(2)=G(2)+US(L)*HTRUE(K)-US(L2)*HTRUE(K2)
13 K=MODE-I
14 IF (ABSF(FL(K)/FL(MODE)-1.) - EP) 15,15,16
15 G=1.
16 GOTO 17
17 G=(FL(K)-FL(MODE)) / G
18 DO 18 J=1,N
19 K=K1+J
20 L=L1+J
21 HTRUE(K)=H(L)-G(1)*HTRUE(K)
22 K=2*K
23 J=2*MODE
24 IF (ABSF((FL(K-1)*FL(J-1)+FL(K)*FL(J))/(FL(J-1)**2+FL(J)**2)-1.)
25 -EP) 20,20,22
26 IF (ABSF((FL(K)*FL(J-1)-FL(K-1)*FL(J)) / (FL(J-1)**2+FL(J)**2))
27 -EP) 21,21,22
28 G(1)=1.
29 G(2)=0.
30 GOTO 23
31 G(3)=G(1)**2+G(2)**2
32 G(4)=(FL(K)-FL(J))*G(1)-(FL(K-1)-FL(J-1))*G(2)
33 G(1)=(FL(K-1)-FL(J-1))*G(1)+(FL(K)-FL(J))*G(2) / G(3)
34 G(2)=G(4) / G(3)
35 DO 24 J=1,N
36 K=K1+J1
37 K2=K+MD
38 L=L1+J1
39 L2=L+MD
40 G(2)=HTRUE(K)
41 HTRUE(K)=H(L)+ G(2)*HTRUE(K2)-G(1)*HTRUE(K)

```

```

      HTRUE(K2)= H(L2)- G(1)*HTRUE(K2)-G(2)*G(3)
24      CONTINUE
25      CONTINUE

      I=0
      CALL NPXRMX (HTRUE(K1+1),HTRUE(K1+1),N,G,I,MD,NC,1)

31      GOTO (26,32),NC
26      DO 29 J=1,N
          L1=(J-1)*MD
          L2=K1+J
          DO 29 I=1,N
              L=L1+I
              IF (I-INDEX) 28,27,28
27          U(L)=0.
              GOTO 29
28          K=K1+I
              U(L)=U(L)-H(K)*US(L2)
29          CONTINUE

30      RETURN

32      DO 35 I=1,N
          L1=MD*NC*(I-1)
          L2=K1+I
          J=L2+MD
          DO 35 J1=1,N
              L=L1+J1
              K3=L+MD
              IF (J1-INDEX) 34,33,34
33          U(L)=0.
              U(K3)=0.
              GOTO 35
34          K=K1+J1
              K2=K+MD
              U(L)=U(L)-H(K)*US(L2)+H(K2)*US(J)
              U(K3)=U(K3)-H(K2)*US(L2)-H(K)*US(J)
35          CONTINUE

```

MTRS0297
MTRS0298
MTRS0299
MTRS0300
MTRS0301
MTRS0302
MTRS0303
MTRS0304
MTRS0305
MTRS0306
MTRS0307
MTRS0308
MTRS0309
MTRS0310
MTRS0311
MTRS0312
MTRS0313
MTRS0314
MTRS0315
MTRS0316
MTRS0317
MTRS0318
MTRS0319
MTRS0320
MTRS0321
MTRS0322
MTRS0323
MTRS0324
MTRS0325
MTRS0326
MTRS0327
MTRS0328
MTRS0329
MTRS0330
MTRS0331
MTRS0332
MTRS0333
MTRS0334

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MTRS0335
MTRS0336
MTRS0337

S W E E P X S U B R O U T I N E

GOTO 30

END(1,0,0,0,0,0,0,0,0,1,0,0,0,0,0)

EMENTS

EMENTS

ENTRY POINTS TO SUBROUTINES NOT OUTPUT FROM LIBRARY

NPNRMX

EXTERNAL FORMULA NUMBERS WITH CORRESPONDING INTERNAL FORMULA NUMBERS AND OCTAL LOCATIONS

EFN	IFN	LOC	EFN	IFN	LOC	EFN	IFN	LOC	EFN	IFN	LOC
6	10	00276	8	12	00311	9	15	00341	10	19	00363
11	21	00373	12	29	00442	13	30	00462	14	32	00472
15	33	00503	16	35	00506	17	36	00512	18	39	00532
19	41	00543	20	44	00503	21	45	00626	22	48	00633
23	52	00674	24	60	00753	25	61	00761	31	65	01016
26	66	01020	27	72	01062	28	74	01066	29	76	01102
30	77	01116	32	79	01122	33	87	01200	34	90	01206
35	94	01251									

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--CLOSE-- DOUBLE PRECISION, REAL OR COMPLEX ARITHMETIC.

```

C SUBROUTINE FINDS 2 CLOSE EIGENVALUES AND THEIR ASSOCIATED EIGENVECTORS
C OF A MATRIX ON WHICH ITERATIONS IN PROGRESS.
C
C U = DYNAMIC MATRIX
C GUESS=1ST. GUESS VECTOR
C (RESULT OF K PRIOR ITERATIONS)
C H1=1ST. VECTOR SOLUTION
C H2=2ND. VECTOR SOLUTION
C INDEX = NORMALIZING NUMBER, 1ST. EIGENVALUE
C INDEX1 = NORMALIZING NUMBER, 2ND. EIGENVALUE
C MAXR= SINGLE PRECISION DIMENSIONED NO. OF ROWS OF U, H1, H2, GUESS.
C NC = 1, MATRIX REAL
C = 2, MATRIX COMPLEX
C H= BLOCK OF WORKING CORE (2*MAXR-REAL, 4*MAXR-COMPLEX)*2
C FL=20 CORE STORAGES FOR TEMPORARY RESULTS
C TEST = TEST NUMBER FOR DETERMINING CONVERGENCE OF P AND Q
C AITKEN = TEST NUMBER FOR DETERMINING CONVERGENCE OF SPEEDER-UPPER
C
C SUBROUTINE CLOSE (U, GUESS, H1, H2, EIG1, EIG2, INDEX, INDEX1, N,
C NC, MAXITR, ITERS, NAKOR, TEST, TESTSR, AITKEN,
C AT, MAXR, H, FL, IR)
C
C DIMENSION U(1), GUESS(1), H1(1), H2(1), H(1), FL(1), TEST(1), AITKEN(1)
C , EIG1(1), EIG2(1)
C DIMENSION P(1), PI(1), Q(1), QI(1), P1(1), P1I(1), Q1(1), Q1I(1)
C
C IF ACCUMULATOR OVERFLOW 5,5
C IF DIVIDE CHECK 6,6
C IR=0
C NAK=4
C ITERS=0
C MX=NC*N
C NX=2*N
C CALL OPFORM (O,U,U,0,N,MX,MAXR,MAXR)
C CALL OPFORM (O,GUESS,GUESS,0,N,NC,MAXR,MAXR)
C IGOTO=1

```

--CLOSE-- DOUBLE PRECISION, REAL OR COMPLEX ARITHMETIC.

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```

      K1=2*NC*MAXR
      B ITERK = ITERK+1
      INDEX=0
      I1=4*(NAK-1)+1
      CALL OPMLTX (U,NC,GUESS,NC,H,N,N,1,MAXR,MAXR,MAXR,2)
      CALL NPNRMX (H,GUESS,N,FL(I1),INDEX,MAXR,NC,2)

      IF ACCUMULATOR OVERFLOW 84,9
      9 IF OVIOE CHECK 84,10

      10 GOTO (11,45,49,51),NAK
      11 GOTO (12,35),NC
      12 I1=2*INDEX-1
      I2=I1+K1
      P1=FL(5)*((FL(9)*H2(I1)*H1(I1)-FL(1)*GUESS(I1)*H(I2))/(FL(5)*
      1 H2(I1)*H(I2)-FL(9)*H1(I1)**2) )
      Q1=FL(5)*FL(9)*((FL(1)*GUESS(I1)*H1(I1)-FL(5)*H2(I1)**2) /
      1 (FL(5)*H2(I1)*H(I2)-FL(9)*H1(I1)**2) )
      P1I=0.
      Q1I=0.
      FL(19)=(FL(1)-FL(5))**2
      GOTO (56,13),IGOTO

      13 FL(17) = ( SQRTF( (P1-P)**2+(P1I-PI)**2)/( SQRTF(Q1**2+Q1I**2)))
      FL(18) = ABSF (1.-SQRTF( (Q1**2+Q1I**2) / (Q**2+QI**2) ) )

      IF OVIOE CHECK 22,22
      22 IF ( FL(17)-TEST ) 14,14,15
      14 IF ( FL(18)-TEST ) 60,60,15
      15 00 16 J=1,NC
      J1=2*MAXR*(J-1)
      00 16 I=1,NX,2
      K=J1+I
      IF ( ABSF(H2(K)-GUESS(K)) - TESTSR ) 16,16,17
      16 GOTO 82

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MTRS0744
 MTRS0745
 MTRS0746
 MTRS0747
 MTRS0748
 MTRS0749
 MTRS0750
 MTRS0751
 MTRS0752
 MTRS0753
 MTRS0754
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 MTRS0759
 MTRS0760
 MTRS0761
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 MTRS0768
 MTRS0769
 MTRS0770
 MTRS0771
 MTRS0772
 MTRS0773
 MTRS0774
 MTRS0775
 MTRS0776
 MTRS0777
 MTRS0778
 MTRS0779
 MTRS0780
 MTRS0781

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--CLOSE-- DOUBLE PRECISION, REAL OR COMPLEX ARITHMETIC.

```

17 GOTO (56,18),IGOTO
18 GOTO (19,40),NC

19 IF (MAXITR-ITERS-5) 54,54,190
190 DO 21 I=1,NX,2
    IF (H2(I)-H1(I)) 20,191,20
191 IF ( ( GUESS(I)-H2(I) ) ) 54,21,54
20 IF (ABS( ( GUESS(I)-H2(I))/(H2(I)-H1(I)) ) - AITKEN) 21,21,54
21 CONTINUE

DO 28 I=1,NX,2
    Q= ( GUESS(I)-2.*H2(I)+H1(I) )
    IF ( Q ) 25,24,25
0 24 H(I)=H1(I)
    GOTO 28
0 25 H(I)= H1(I)- ( (H2(I)-H1(I))*2) / Q
28 CONTINUE
29 NAK=5
    NAKDR=NAKDR+1
    IGOTO=1

C EXCHANGE TRIAL VECTOR AND TEST FOR MAXIMUM NUMBER OF ITERATIONS.
30 DO 31 J=1,NC
    J1=MAXR*(J-1)
    DO 31 I=1,NX
        K=J1+I
        GUESS(K)=H(K)
32 NAK=NAK-1
33 IF ( ITERS-MAXITR ) 8,85,85

C COMPLEX P1 AND Q1 COMPUTATIONS
35 I1=2*INDEX-1
    I2=I1+K1
    I3=I1+2*MAXR
    I4=I2+2*MAXR
    X=FL(I1)*GUESS(I1)-FL(I3)*GUESS(I3)
    X1=FL(I2)*H1(I1)-FL(I1)*H1(I3)
    X2=FL(I3)*GUESS(I1)+FL(I1)*GUESS(I3)
0
0
0

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MTRS0782
MTRS0783
MTRS0784
MTRS0785
MTRS0786
MTRS0787
MTRS0788
MTRS0789
MTRS0790
MTRS0791
MTRS0792
MTRS0793
MTRS0794
MTRS0795
MTRS0796
MTRS0797
MTRS0798
MTRS0799
MTRS0800
MTRS0801
MTRS0802
MTRS0803
MTRS0804
MTRS0805
MTRS0806
MTRS0807
MTRS0808
MTRS0809
MTRS0810
MTRS0811
MTRS0812
MTRS0813
MTRS0814
MTRS0815
MTRS0816
MTRS0817
MTRS0818
MTRS0819

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--CLOSE-- DOUBLE PRECISION, REAL OR COMPLEX ARITHMETIC.

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D      X3=FL(I1)*H1(I1)+FL(9)*H1(I3)
D      /X4=FL(5)*H2(I1)-FL(7)*H2(I3)
D      X5=FL(7)*H2(I1)+FL(5)*H2(I3)
D      X6=H(I2)*X4-H(I4)*X5-H(I1)*X1+H(I3)*X3
D      X7=H(I2)*X5+H(I4)*X4-H(I1)*X3-H(I3)*X1
D
D      X8= FL(5)*(H2(I1)*X1-H2(I3)*X3-H(I2)*X+H(I4)*X2) - FL(7)*(H2(I3)
1      *X1+H2(I1)*X3-H(I4)*X+H(I2)*X2)
D      X9= FL(5)*(H2(I3)*X1+H2(I1)*X3-H(I4)*X+H(I2)*X2) + FL(7)*(H2(I1)
1      *X1-H2(I3)*X3-H(I2)*X+H(I4)*X2)
D      P1= (X8*X6+X9*X7) / (X6**2+X7**2)
D      P11= (X9*X6-X8*X7) / (X6**2+X7**2)
D
D      X8= (FL(5)*FL(9)-FL(7)*FL(I1))*(H1(I1)*X-H1(I3)*X2-H2(I1)*X4+
1      H2(I3)*X5) - (FL(7)*FL(9)+FL(5)*FL(I1))*(H1(I3)*X+
2      H1(I1)*X2-H2(I3)*X4-H2(I1)*X5)
D      X9=(FL(5)*FL(9)-FL(7)*FL(I1))*(H1(I3)*X+H1(I1)*X2-H2(I3)*X4-
1      H2(I1)*X5) + (FL(7)*FL(9)+FL(5)*FL(I1))*(H1(I1)*X-
2      H1(I3)*X2-H2(I1)*X4+H2(I3)*X5)
D      Q1= (X8*X6+X9*X7) / (X6**2+X7**2)
D      Q11=(X9*X6-X8*X7) / (X6**2+X7**2)
D      FL(I9)=(FL(I1)-FL(5))*2+(FL(I3)-FL(7))*2
D      GOTO (56,13),IGOTO
40 IF (MAX1TR-ITERS-5) 54,54,400
400 DO 41 I=1,NX,2
      J1=2*MAXR+1
      X=(H2(I1)-H1(I1))*2 + (H2(J1)-H1(J1))*2
      IF ( X ) 37,401,37
401 IF ( ( GUESS(I1)-H2(I1))*2 + ( GUESS(J1)-H2(J1))*2 ) 54,41,54
37 IF ( ( GUESS(I1)-H2(I1))*2 + ( GUESS(J1)-H2(J1))*2 ) /X -AT )
1 41,41,54
41 CONTINUE
DO 44 I=1,NX,2
  J=2*MAXR+1
  X=GUESS(I)-2.*H2(I)+H1(I)
  IF ( X ) 43,42,43
D 42 H(I)=H1(I)
D H(J)=H1(J)
MTRS0820
MTRS0821
MTRS0822
MTRS0823
MTRS0824
MTRS0825
MTRS0826
MTRS0827
MTRS0828
MTRS0829
MTRS0830
MTRS0831
MTRS0832
MTRS0833
MTRS0834
MTRS0835
MTRS0836
MTRS0837
MTRS0838
MTRS0839
MTRS0840
MTRS0841
MTRS0842
MTRS0843
MTRS0844
MTRS0845
MTRS0846
MTRS0847
MTRS0848
MTRS0849
MTRS0850
MTRS0851
MTRS0852
MTRS0853
MTRS0854
MTRS0855
MTRS0856
MTRS0857

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--CLOSE-- DOUBLE PRECISION, REAL OR COMPLEX ARITHMETIC.

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D 43      GOTO 44
D          X1=GUESS(J)-2.*H2(J)+H1(J)
D          X2=H2(I)-H1(I)
D          X3=H2(J)-H1(J)
D          H(I) = H1(I) - ((X2**2-X3**2)*X+(2.*X2*X3*X1))/(X**2+X1**2)
D          H(J) = H1(J) - ((2.*X2*X3*X)-(X2**2-X3**2)*X1) / (X**2+X1**2)
D          44 CONTINUE
D          GOTO 29

45 00 46 J=1,NC
      J1=MAXR*(J-1)*2
      DO 46 I=1,NX
      K=J1+I
      H2(K)=GUESS(K)
      GOTO 32

46 00 50 J=1,NC
      J1=MAXR*(J-1)*2
      DO 50 I=1,NX
      K=J1+I
      H1(K)=GUESS(K)
      GOTO 32

51 00 52 J=1,NC
      J1=2*MAXR*(J-1)
      DO 52 I=1,NX
      K=J1+I
      L=K1+I +J1
      H(L)=GUESS(K)
      GOTO 32

54 IF ( ITERS-MAXITR ) 56,85,85

56 00 58 J=1,NC
      J1=MAXR*(J-1)*2
      DO 58 I=1,NX
      K=J1+I
      L=K1+J1+I
      H(L)=H1(K)

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MTRS0858
MTRS0859
MTRS0860
MTRS0861
MTRS0862
MTRS0863
MTRS0864
MTRS0865
MTRS0866
MTRS0867
MTRS0868
MTRS0869
MTRS0870
MTRS0871
MTRS0872
MTRS0873
MTRS0874
MTRS0875
MTRS0876
MTRS0877
MTRS0878
MTRS0879
MTRS0880
MTRS0881
MTRS0882
MTRS0883
MTRS0884
MTRS0885
MTRS0886
MTRS0887
MTRS0888
MTRS0889
MTRS0890
MTRS0891
MTRS0892
MTRS0893
MTRS0894
MTRS0895

--CLOSE-- DOUBLE PRECISION, REAL OR COMPLEX ARITHMETIC.

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      H1(K)=H2(K)
      H2(K)=GUESS(K)
58      IGOTO=2
      J=17
      DO 59 I=1,12
        J=J-1
        FL(J)=FL(J-4)
59      P=P1
      Q=Q1
      P1=P11
      Q1=Q11
      GOTO 33
60      GOTO (64,70),NC

      D 64 P= SQRTF( P1**2-4.*Q1)
      D 64 FL(9) = (-P1+P) / 2.
      D 64 FL(5) = (-P1-P) / 2.
      D 64 IF ( ABSF(FL(9)) - ABSF(FL(5)) )
      D 64 65 P=FL(5)
      D 64 FL(5)=FL(9)
      D 64 FL(9)=P
      D 64 DO 68 I=1,NX,2
      D 64 H1(I)=FL(5)*H2(I)-FL(1)*GUESS(I)
      D 64 H2(I)=FL(1)*GUESS(I)-FL(9)*H2(I)
      D 64 GOTO 76

      D 70 X=(P1**2-4.*Q1-P11**2)
      D 70 X1=(2.*P1*P11-4.*Q11)
      D 70 IF ( X )
      D 70 73 P1= SQRTF( (-X + SQRTF( X**2 + X1**2 )) / 2.)
      D 70 P = X1 / (2.*P1)
      D 70 GOTO 77
      D 75 P = SQRTF( ( X + SQRTF( X**2 + X1**2 )) / 2.)
      D 75 P1= X1 / (2.*P)
      D 77 FL(9) = (-P1+P) / 2.
      D 77 FL(11)= (-P11+P1) / 2.
      D 77 FL(5) = (-P1-P) / 2.
      D 77 FL(7) = (-P11-P1) / 2.

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--CLDSE-- DOUBLE PRECISION, REAL DR COMPLEX ARITHMETIC.

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```

D IF ( (FL(9)**2+FL(11)**2) - (FL(5)**2+FL(7)**2) ) 71,72,72
D 71 P=FL(5)
D FL(5)=FL(9)
D FL(9)=P
D P=FL(7)
D FL(7)=FL(11)
D FL(11)=P
D 72 DD 74 I=1,NX,2
J=2*MAXR+1
D H1(I) = FL(5)*H2(I)-FL(7)*H2(J)-FL(1)*GUESS(I)+FL(3)*GUESS(J)
D H1(J) = FL(5)*H2(J)+FL(7)*H2(I)-FL(1)*GUESS(J)-FL(3)*GUESS(I)
D P=FL(11)*H2(I)
D H2(I) = FL(1)*GUESS(I)-FL(3)*GUESS(J)-FL(9)*H2(I)+FL(11)*H2(J)
D 74 H2(J) = FL(3)*GUESS(I)+FL(1)*GUESS(J)-FL(9)*H2(J)-P

76 INDEX=0
CALL NPNRMX (H1,H1,N,FL(13),INDEX,MAXR,NC,2)
CALL NPNRMX (H2,H2,N,FL(13),INDEX,MAXR,NC,2)

DD 78 J=1,NC
J1=2*MAXR*(J-1)
DD 78 I=1,NX,2
K=J1+I
H2(K)=H1(K)-H2(K)
78 INDEX1=0
CALL DPEDRM (O,H2,H2,1,N,NC,MAXR,MAXR)
CALL NPNRMX (H2,H2,N,FL(13),INDEX1,MAXR,NC,1)

EIG1=FL(9)
EIG2=FL(5)
GOTD (80,79),NC
79 EIG1(2)=FL(11)
EIG2(2)=FL(7)

80 CALL DPEDRM (O,U,U,1,N,MX,MAXR,MAXR)
CALL DPEDRM (O,H1,H1,1,N,NC,MAXR,MAXR)
IF ACCUMULATOR OVERFLOW 92,90
90 IF QUOTIENT OVERFLOW 92,91

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MTRS0934
MTRS0935
MTRS0936
MTRS0937
MTRS0938
MTRS0939
MTRS0940
MTRS0941
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MTRS0943
MTRS0944
MTRS0945
MTRS0946
MTRS0947
MTRS0948
MTRS0949
MTRS0950
MTRS0951
MTRS0952
MTRS0953
MTRS0954
MTRS0955
MTRS0956
MTRS0957
MTRS0958
MTRS0959
MTRS0960
MTRS0961
MTRS0962
MTRS0963
MTRS0964
MTRS0965
MTRS0966
MTRS0967
MTRS0968
MTRS0969
MTRS0970
MTRS0971

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--CLOSE-- DOUBLE PRECISION, REAL OR COMPLEX ARITHMETIC.

```

91 IF DIVIDE CHECK          92,81
92   IR=-1

81 RETURN
82 CALL DPFORM (0,GUESS,GUESS,1,N,NC,MAXR,MAXR)
   CALL DPFORM (0,U,U,1,N,MX,MAXR,MAXR)
   INDEX1=INDEX
   EIG1=FL(1)
   IR=1
   GOTO (81,83),NC
83 EIG1(2)=FL(3)
   GOTO 81
84 IR=-1
   GOTO 80

85 X=FL(17) / TEST
   X1=FL(18) / TEST
   GOTO (94,97),NC
94 X2 = (GUESS(1)-H2(1))*2
   DO 96 I=3,NX,2
     IF (X2-(GUESS(I)-H2(I))*2) 95,96,96
95 X2= (GUESS(I)-H2(I))*2
96 CONTINUE
   GOTO 100
97 X2=(GUESS(1)-H2(1))*2 + (GUESS(2*MAXR+1)-H2(2*MAXR+1))*2
   DO 99 I=3,NX,2
     J=MAXR+1
     IF (X2 - (GUESS(I)-H2(I))*2+(GUESS(J)-H2(J))*2 ) 98,99,99
98 X2= (GUESS(I)-H2(I))*2 + (GUESS(J)-H2(J))*2
99 CONTINUE
100 X2=SQRTF(X2) / TESTSR
    ITES=ITES+1
    IF (X-X1) 86,87,87
86 X=X1
87 IF (X-X2) 60,60,82

      ENO(1,0,0,0,0,0,0,0,1,0,0,0,0,0)

```

MTRS0972
MTRS0973
MTRS0974
MTRS0975
MTRS0976
MTRS0977
MTRS0978
MTRS0979
MTRS0980
MTRS0981
MTRS0982
MTRS0983
MTRS0984
MTRS0985
MTRS0985
MTRS0987
MTRS0988
MTRS0989
MTRS0990
MTRS0991
MTRS0992
MTRS0993
MTRS0994
MTRS0995
MTRS0996
MTRS0997
MTRS0998
MTRS0999
MTRS1000
MTRS1001
MTRS1002
MTRS1003
MTRS1004
MTRS1005
MTRS1006
MTRS1007

--CLOSE-- DOUBLE PRECISION, REAL OR COMPLEX ARITHMETIC.

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STORAGE NOT USED BY PROGRAM

DEC	OCT
4956	11534
	32561 77461

STORAGE LOCATIONS FOR VARIABLES APPEARING IN DIMENSION AND EQUIVALENCE STATEMENTS

	DEC	OCT		DEC	OCT		DEC	OCT
PL1	4945	11521	PI	4947	11523	PI	4953	11531
Q11	4941	11515	Q1	4943	11517	Q1	4949	11525
X1	4937	11511	X2	4935	11507	X3	4933	11505
X5	4929	11501	X6	4927	11477	X7	4925	11475
X9	4921	11471	X	4939	11513			

STORAGE LOCATIONS FOR VARIABLES NOT APPEARING IN COMMON, DIMENSION, OR EQUIVALENCE STATEMENTS

	DEC	OCT		DEC	OCT		DEC	OCT
I1	4919	11467	I2	4918	11466	I3	4917	11465
IGOTO	4915	11463	I	4914	11462	J1	4913	11461
K1	4911	11457	K	4910	11456	L	4909	11455
NAK	4907	11453	NX	4906	11452			

LOCATIONS FOR OTHER SYMBOLS NOT APPEARING IN SOURCE PROGRAM

	DEC	OCT		DEC	OCT		DEC	OCT
1)	4859	11373	2)	4812	11314	3)	4819	11323
6)	4832	11340	9)	4838	11346	AGD	4803	11303
C1G1	4888	11430	C1G2	4889	11431	C1G4	4890	11432
C1G6	4892	11434	C1G7	4893	11435	C1G8	4894	11436
C1GA	4896	11440	C1G8	4897	11441	C1GC	4898	11442
C1200	4900	11444	C1201	4901	11445	C1205	4902	11446
C1208	4904	11450	C120E	4905	11451	O1118	1783	03367
O113L	4784	11260	O1208	1499	02733	O1201	1612	03114
O1219	1790	03376	O1222	3487	06537	O1226	3527	06707
O1228	3541	06725	O1237	4644	11044	O123A	4689	11121
O1322	3486	06636	O1326	3526	06706	O1327	3532	06714
O133A	4688	11120	O1403	1199	02257	O140C	1570	03042
O142E	3637	07065	O1421	4526	10656	O1432	4599	10767
						4)	32767	77777
						C1G0	4887	11427
						C1G5	4891	11433
						C1G9	4895	11437
						C1GD	4899	11443
						C1209	4903	11447
						O111T	3448	06570
						O120J	1616	03120
						O1227	3533	06715
						O1308	1498	02732
						O1328	3540	06724
						O1410	3411	06523
						O1436	4639	11037

0143G	4726	11166	01510	3410	06522	0162E	3636	07064	D162T	4525	10655
01636	4638	11036	0172E	3635	07063	E1E	1598	03076	E1G	1606	03106
E10	1638	03146	E1P	1642	03152	E1R	1658	03172	E133	4628	11024
E131	4765	11235	E13K	4782	11256						

LOCATIONS OF NAMES IN TRANSFER VECTOR

	DEC	OCT		DEC	OCT		DEC	OCT		DEC	OCT
	DEXP(2)	3 0003	DIFORM	0 0000	DPMLTX	1 0001	DSQRT	9 00011			
	NNPRMX	2 0002	SQRT	7 0007	(OFAD)	8 0010	(DFOP)	6 00006			
	(DFMP)	4 0004	(OFSB)	5 0005							

ENTRY POINTS TO SUBROUTINES NOT OUTPUT FROM LIBRARY

	DEXP(2 (DFMP)	DPFORM (OF5B)	DPLTX	DSQRT	NPNRMX	SQRT	(OFA0)	(DFOP)
--	------------------	------------------	-------	-------	--------	------	--------	--------

EXTERNAL FORMULA NUMBERS WITH CORRESPONDING INTERNAL FORMULA NUMBERS AND OCTAL LOCATIONS

EFN	IFN	LOC	EFN	IFN	LOC	EFN	IFN	LOC	EFN	IFN	LOC
5	7	02167	6	8	02171	8	19	02260	9	28	02350
10	29	02352	11	30	02356	12	31	02360	13	39	02734
22	42	03022	14	43	03026	15	44	03033	16	49	03103
17	51	03115	18	52	03121	19	53	03123	190	54	03133
191	56	03141	20	57	03156	21	58	03176	24	62	03237
25	64	03250	28	65	03312	29	66	03314	30	69	03327
31	73	03357	32	74	03370	33	75	03377	35	76	03404
40	98	05661	400	99	05667	401	103	05725	37	104	05744
41	105	05765	42	110	06041	43	113	06062	44	118	06510
45	120	06514	46	124	06552	49	126	06563	50	130	06617
51	132	06630	52	137	06673	54	139	06710	56	140	06716
58	147	06766	59	152	07016	60	158	07066	64	159	07070
65	163	07234	66	166	07264	68	168	07323	70	170	07362
73	173	07503	75	176	07626	77	178	07750	71	183	10153
72	189	10233	74	195	10527	76	196	10606	78	205	10704
79	214	10763	80	216	10770	90	222	11025	91	224	11030
92	225	11033	81	226	11040	82	228	11045	83	236	11112

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--CLOSE-- DOUBLE PRECISION, REAL OR COMPLEX ARITHMETIC.

95 246 11155
99 254 11253

94 243 11133
98 253 11236
87 259 11276

85 240 11122
97 249 11167
86 258 11274

84 238 11115
96 247 11163
100 255 11261

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DPFORM

```

C SUBROUTINE PICKSUP MATRIX IN SINGLE PRECISION FORMAT AND STORES IT
C IN DOUBLE PRECISION FORMAT WITH L.S.H. = ZERO.
C OR.....
C PICKS UP DOUBLE PRECISION MATRIX AND STORES IT IN SINGLE
C PRECISION FORMAT WITH L.S.H. LOST.
C
C NUTAPE = ZERO, MATRIX IS IN CORE
C = NON-ZERO, MATRIX IS ON TAPE UNIT NUTAPE, BY
C COLUMNS WITH EACH COLUMN = 1 RECORD
C
C A = SINGLE PRECISION MATRIX
C B = DOUBLE PRECISION MATRIX (MAY = A)
C ICHUZ = 0, CONVERT SINGLE PR. TO DOUBLE PR.
C = NON-ZERO, CONVERT DOUBLE PR. TO SINGLE PR.
C
C M = NUMBER ROWS
C N = NUMBER COLUMNS
C MA = DIMENSIONED NUMBER ROWS OF A
C MB = DIMENSIONED NUMBER ROWS OF B
C MA,MB ARE SINGLE PRECISION DIMENSIONS, HOWEVER B MUST BE
C LARGE ENUF TO ACCOMMODATE THE DOUBLE MATRIX.
C IF MATRIX IS COMPLEX, N=TWICE THE ACTUAL NUMBER OF
C COLUMNS. IN THIS CASE THE RESERVEO NUMBER OF CORES FOR
C A MUST BE AT LEAST 2*M*N(ACTUAL) AND FOR B, AT LEAST
C 4*M*N(ACTUAL).
C
C SUBROUTINE OPFORM (NUTAPE,A,B,ICHUZ,M,N,MA,MB)
C
C DIMENSION A(1),B(1)
C
C MB1 = 2*MB
C IF ( ICHUZ ) 20,10,20
C 10 IF ( NUTAPE ) 11,13,11
C 11 REWIND NUTAPE
C
MTRS1010
MTRS1011
MTRS1012
MTRS1013
MTRS1014
MTRS1015
MTRS1016
MTRS1017
MTRS1018
MTRS1019
MTRS1020
MTRS1021
MTRS1022
MTRS1023
MTRS1024
MTRS1025
MTRS1026
MTRS1027
MTRS1028
MTRS1029
MTRS1030
MTRS1031
MTRS1032
MTRS1033
MTRS1034
MTRS1035
MTRS1036
MTRS1037
MTRS1038
MTRS1039
MTRS1040
MTRS1041
MTRS1042
MTRS1043
MTRS1044
MTRS1045
MTRS1046

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DPFORM

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```

K = 1
DO 12 I=1,N
  KK = K+M
  READ TAPE NUTAPE, (A(J),J=K,KK)
12 K = K+M
  MB1 = 2*M
  GOTO 14

13 IX = 1
  IF (M-MB) 15,14,15
14 IX=1

15 DO 16 I=1,N
  K = (N-1)*M+M+1
  KK = (N-1)*MB1+IX+2*M+1
  DO 16 J=1,M
    IGET = K-J
    IPUT = KK-2*J
    B( IPUT ) = A( IGET )
    B( IPUT+1 ) = 0.
16 GOTO 26

20 IF (NUTAPE) 21,23,21
21 REWIND NUTAPE
  K = 1
  DO 22 I=1,N
    KK = K+2*M
    READ TAPE NUTAPE, (B(J),J=K,KK)
22 K = K+2*M
  MB1 = 2*M

23 K = 0
  KK = 0
  DO 25 I=1,N
    DO 24 J=1,M
      IGET = KK+2*J-1
      IPUT = K+J
      A( IPUT ) = B( IGET )
24 A( IPUT ) = B( IGET )

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MTRS1047
 MTRS1048
 MTRS1049
 MTRS1050
 MTRS1051
 MTRS1052
 MTRS1053
 MTRS1054
 MTRS1055
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 MTRS1062
 MTRS1063
 MTRS1064
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 MTRS1066
 MTRS1067
 MTRS1068
 MTRS1069
 MTRS1070
 MTRS1071
 MTRS1072
 MTRS1073
 MTRS1074
 MTRS1075
 MTRS1076
 MTRS1077
 MTRS1078
 MTRS1079
 MTRS1080
 MTRS1081
 MTRS1082
 MTRS1083
 MTRS1084

DPFORM

K = K+MA

25 KK = KK+MB1

26 RETURN

END(1,0,0,0,0,0,0,0,0,1,0,0,0,0,0)

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MTRS1085

MTRS1086

MTRS1087

MTRS1088

OPFORM

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STORAGE NOT USED BY PROGRAM

DEC OCT
280 00430
32561 77461

STORAGE LOCATIONS FOR VARIABLES NOT APPEARING IN COMMON, DIMENSION, OR EQUIVALENCE STATEMENTS

DEC	OCT	DEC	OCT	DEC	OCT	DEC	OCT
IGET	279 00427	INPUT	278 00426	I	277 00425	IX	276 00424
J	275 00423	KK	274 00422	K	273 00421	MB1	272 00420

LOCATIONS FOR OTHER SYMBOLS NOT APPEARING IN SOURCE PROGRAM

DEC	OCT	DEC	OCT	DEC	OCT	DEC	OCT
1)	265 00411	2)	253 00375	3)	256 00400	6)	257 00401
9)	263 00407	C)G1	268 00414	C)G2	269 00415	C)202	270 00416
C)205	271 00417	D)20A	115 00163	D)20L	215 00327	O)30A	114 00162
D)30L	214 00326						

LOCATIONS OF NAMES IN TRANSFER VECTOR

DEC	OCT	DEC	OCT	DEC	OCT
(RLR)	2 00002	(RWT)	0 00000	(TSB)	1 00001

ENTRY POINTS TO SUBROUTINES NOT OUTPUT FROM LIBRARY

(RLR) (RWT) (TSB)

EXTERNAL FORMULA NUMBERS WITH CORRESPONDING INTERNAL FORMULA NUMBERS AND OCTAL LOCATIONS

EFN	IFN	LOC	EFN	IFN	LOC	EFN	IFN	LOC
10	6	00103	11	7	00105	12	16	00131
14	21	00152	15	22	00154	16	29	00240
21	32	00255	22	41	00303	23	43	00316
25	51	00362	26	52	00371	24	49	00352

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SINGLE OR DOUBLE PRECISION, REAL OR COMPLEX MATRIX MULTIPLICATION

```

C
C      SUBROUTINE
C
C      CALLING SEQUENCE.....
C      CALL DPMLTX (A,NA,B,NB,C,M,N,K,MA,MB,MC,NP)
C
C      A = PRMULTIPLIER      MA = DIMENSIONED NUMBER OF ROWS A
C      B = POSTMULTIPLIER    MB = DIMENSIONED NUMBER OF ROWS B
C      C = PRODUCT           MC = DIMENSIONED NUMBER OF ROWS C
C      M = NO. ROWS IN A     = NO. ROWS IN C
C      N = NO. COLUMNS IN A = NO. ROWS IN B
C      K = NO. COLUMNS IN B = NO. COLUMNS IN C
C      NA AND NB = 1 IF A OR B, RESPECTIVELY, ARE REAL.
C                  = 2 IF A OR B, RESPECTIVELY, ARE COMPLEX.
C      NP= PRECISION, (1 OR 2)
C      MA,MB,MC ARE SINGLE PRECISION DIMENSIONS,(1/2 OF ACTUAL CORE
C      RESERVED FOR REAL DOUBLE PRECISION MATRICES, AND 1/4 OF
C      ACTUAL CORE RESERVED FOR COMPLEX DOUBLE PRECISION MATRICES)
C
C      SUBROUTINE DPMLTX (A,NA,B,NB,C,M,N,K,MA,MB,MC,NP)
C
C      DIMENSION A(1), B(1), C(1)
C
C      IA=MC*K*NP
C      IB=MA*4*NP*NA
C      IC=1
C      ID=MA*NP*NA
C      IH=MC*NP
C      IJ=MC*NP
C      IK=N*NP
C      IL=1*NP
C      IM=1*NP
C      GOTO (5,7),NA
C      6 GOTO (11,10),NB
C      7 GOTO (9,8),NB
C      8      IC=2
C          GOTO 10
C      9 IH=2*IH

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MTRS0035
MTRS0036
MTRS0037
MTRS0038
MTRS0039
MTRS0040
MTRS0041
MTRS0042
MTRS0043
MTRS0044
MTRS0045
MTRS0046
MTRS0047
MTRS0048
MTRS0049
MTRS0050
MTRS0051
MTRS0052
MTRS0053
MTRS0054
MTRS0055
MTRS0056
MTRS0057
MTRS0058
MTRS0059
MTRS0060
MTRS0061
MTRS0062
MTRS0063
MTRS0064
MTRS0065
MTRS0066
MTRS0067
MTRS0068
MTRS0069
MTRS0070
MTRS0071

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MTR0072
MTR0073
MTR0074
MTR0075
MTR0076
MTR0077
MTR0078
MTR0079
MTR0080
MTR0081
MTR0082
MTR0083
MTR0084
MTR0085
MTR0086
MTR0087
MTR0088
MTR0089
MTR0090
MTR0091
MTR0092
MTR0093
MTR0094
MTR0095
MTR0096
MTR0097
MTR0098
MTR0099
MTR0100
MTR0101
MTR0102
MTR0103
MTR0104
MTR0105
MTR0106
MTR0107
MTR0108
MTR0109

      IC=3
      IA=2*IA
      IJ=2*IJ
      I=1, IK, IL
      INC=IM
      DO 16 J=1, IA, IH
      IN=INC
      GOTO (12,14), NP

12  C(J)=0.
      DO 13 L=1, IB, ID
      IN=IN+NP
      C(J)=C(J)+A(L)*B(IN)
      GOTO 16

13  C(J)=0.
      DO 15 L=1, IB, ID
      IN=IN+NP
      C(J)=C(J)+A(L)*B(IN)

D 14  C(J)=0.
      DO 15 L=1, IB, ID
      IN=IN+NP
      C(J)=C(J)+A(L)*B(IN)

D 15  INC=INC+NP*MB
      GOTO (30,18,24), IC
      IE=1+NP*MA
      INC=IM
      DO 23 J=1, IA, IJ
      IN=INC
      IF=J+NP*MC
      GOTO (19,21), NP
      DO 20 L=1, IB, ID
      IN=IN+1
      IG=IN+MB
      C(IF)=C(IF)+A(L)*B(IN)
      C(J)=C(J)-A(L)*B(IG)
      GOTO 23

20  DO 22 L=1, IB, ID
      IN=IN+NP
      IG=IN+NP*MB
      C(IF)=C(IF)+A(L)*B(IN)

D 21  C(IF)=C(IF)+A(L)*B(IN)

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SINGLE OR DOUBLE PRECISION, REAL OR COMPLEX MATRIX MULTIPLICATION

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```

D 22  C(J)=C(J)-A(L)*B(IG)
      INC=INC+2*NP*MB
      GOTO 30
24  IE=I+NP*MC
      IF=I+NP*MA
      INC=IM
      DO 29  J=IE,IA,IJ
      IN=INC
      GOTO (25,27),NP
25  C(J)=0.
      DO 26  L=IF,IR,ID
      IN=IN+1
      C(J)=C(J)+A(L)*B(IN)
      GOTO 29
D 27  C(J)=0.
      DO 28  L=IF,IB,ID
      IN=IN+NP
      C(J)=C(J)+A(L)*B(IN)
      INC=INC+NP*MB
D 28  29
30  CONTINUE
      RETURN
      END(1,0,0,0,0,0,0,0,0,1,0,0,0,0,0)

```

MTRS0110
MTRS0111
MTRS0112
MTRS0113
MTRS0114
MTRS0115
MTRS0116
MTRS0117
MTRS0118
MTRS0119
MTRS0120
MTRS0121
MTRS0122
MTRS0123
MTRS0124
MTRS0125
MTRS0126
MTRS0127
MTRS0128
MTRS0129
MTRS0130
MTRS0131
MTRS0132

STORAGE NOT USED BY PROGRAM

DEC OCT
606 01136
32561 77461

STORAGE LOCATIONS FOR VARIABLES NOT APPEARING IN COMMON, DIMENSION, OR EQUIVALENCE STATEMENTS

DEC	OCT	DEC	OCT	DEC	OCT	DEC	OCT
IA	605 01135	IB	604 01134	IC	603 01133	ID	602 01132
IE	601 01131	IF	600 01130	IG	599 01127	IH	598 01126
IJ	597 01125	IK	596 01124	IL	595 01123	IM	594 01122
INC	593 01121	IN	592 01120	I	591 01117	J	590 01116

LOCATIONS FOR OTHER SYMBOLS NOT APPEARING IN SOURCE PROGRAM

DEC	OCT	DEC	OCT	DEC	OCT	DEC	OCT
2)	566 01066	3)	569 01071	4)	32767 77777	6)	572 01074
9)	578 01102	C)G0	580 01104	C)G1	581 01105	C)G2	582 01106
C)G3	583 01107	C)G4	584 01110	C)G5	585 01111	C)G6	586 01112
C)G7	537 01113	C)G8	588 01114	C)200	589 01115	D)107	242 00362
D)115	556 01054	D)401	343 00527	E)1	206 00316		

LOCATIONS OF NAMES IN TRANSFER VECTOR

DEC	OCT	DEC	OCT	DEC	OCT
(DFAD)	1 00001	(DFMP)	0 00000		

ENTRY POINTS TO SUBROUTINES NOT OUTPUT FROM LIBRARY

(DFAD) (DFMP)

EXTERNAL FORMULA NUMBERS WITH CORRESPONDING INTERNAL FORMULA NUMBERS AND OCTAL LOCATIONS

EFN	IFN	LOC	EFN	IFN	LOC	EFN	IFN	LOC
6	14	00317	7	15	00322	8	16	00325
10	20	00342	11	22	00352	12	27	00401
14	32	00425	15	35	00447	16	36	00473
						18	38	00510

<p>Aerospace Corporation, El Segundo, California. FLUTTER AND VIBRATION ANALYSIS BY A MODAL METHOD: ANALYTICAL DEVELOP- MENT AND COMPUTATIONAL PROCEDURE, prepared by W. P. Rodden, E. F. Farkas, and H. A. Malcom. 31 July 1963. [221]p. incl. illus. (Report TDR-169(3230-11)TN-15;SSD-TDR-63-158) (Contract AF 04(695)-169) Unclassified report</p> <p>A modal solution of the flutter and vibration prob- lems for a multiple component system is presented. The formulation requires the aerodynamic charac- teristics in the form of aerodynamic influence coefficients, and the structural and inertial characteristics in the form of free vibration modes and frequencies and a mass matrix for each com- ponent. The use of a rigid-body modal matrix permits a general analysis for a system free in space with up to six rigid-body degrees of freedom. The solution also utilizes a newly developed (over)</p>	<p>UNCLASSIFIED</p>
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<p>Aerospace Corporation, El Segundo, California. FLUTTER AND VIBRATION ANALYSIS BY A MODAL METHOD: ANALYTICAL DEVELOP- MENT AND COMPUTATIONAL PROCEDURE, prepared by W. P. Rodden, E. F. Farkas, and H. A. Malcom. 31 July 1963. [221]p. incl. illus. (Report TDR-169(3230-11)TN-15;SSD-TDR-63-158) (Contract AF 04(695)-169) Unclassified report</p> <p>A modal solution of the flutter and vibration prob- lems for a multiple component system is presented. The formulation requires the aerodynamic charac- teristics in the form of aerodynamic influence coefficients, and the structural and inertial characteristics in the form of free vibration modes and frequencies and a mass matrix for each com- ponent. The use of a rigid-body modal matrix permits a general analysis for a system free in space with up to six rigid-body degrees of freedom. The solution also utilizes a newly developed (over)</p>	<p>UNCLASSIFIED</p>
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